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The Ranger 5 Flight Path and Its Determination From Tracking Data

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ABSTRACT

24/02

This Report describes the current best estimate of the Ranger 5 spacecraft flight path and the way in which it was determined. The spacecraft was tracked in the two-way doppler mode until 8 hr after launch, when the batteries were depleted. The transmitter in the roughlanding capsule was tracked for 11 days after launch, except during occultation by the Moon. It is concluded that a small maneuver took place along the Sun-line near the time that power depletion terminated contact with the spacecraft.

A new orbit determination program which treats the effects of station location and physical constant errors was used to estimate the flight path. The results reported on GM_{\oplus} by Ranger 4 (Ref. 1) are reevaluated more accurately to confirm the previous results.

I. INTRODUCTION

This Report describes the current best estimate of the Ranger 5 spacecraft flight path and the way it was determined. It very much parallels the Ranger 4 Report (Ref. 1), except that a more elaborate orbit determination tool was used to obtain the results.

Although a spacecraft failure prevented transponder tracking after launch plus 8 hr, the subsequent tracking of the rough-landing capsule's beacon indicated that Ranger 5 came within 750 km of the Moon's surface, and then continued on into a heliocentric orbit. Power to operate the two-way doppler system was depleted after 8 hr, and as a result, a complete midcourse maneuver was not executed. The capsule beacon data revealed a small

perturbation in the orbit, originally determined by the spacecraft transponder data, which could be explained by the attitude control gas released after the spacecraft failed or by partial execution of the midcourse maneuver.

Section II describes the Deep Space Instrumentation Facility (DSIF) transponder orbit and the orbit determined by occultation of the capsule beacon in terms of its trajectory parameters near the Earth, in translunar flight, and near the Moon. Explanations of symbols used and definitions of key trajectory quantities are given.

Section III summarizes the key events in the tracking of Ranger 5 and gives a general description of the DSIF stations and tracking modes.

Section IV describes the DSIF transponder orbit determination and displays the effects of physical constants in the solution.

Section V discusses the "midcourse maneuver" situation and the study to determine the cause of the perturbation in occultation time. While the spacecraft batteries were depleted at 8 hr after launch, the beacon carried within the capsule continued to operate on its own power supply and was tracked by the DSIF throughout the mission. Valuable data were taken at Goldstone, DSIF 2, and at Johannesburg, South Africa, DSIF 5, in the several

hours prior to and after lunar encounter. Results of these data are presented in Section V.

Section VI gives a functional description of the in-flight determination of the flight path together with a description of the techniques used in editing and weighting the tracking data.

Acknowledgments on the development of the computing programs used in the analysis are also given. The Appendices show the complete printout from the trajectory and the orbit determination programs used to compile this Report.

II. TRAJECTORY DESCRIPTION

A. Launch Phase

The Ranger 5 spacecraft was launched at 16:59:07.84 Greenwich Mean Time (GMT) on October 18, 1962, from the Atlantic Missile Range (AMR) using the Atlas D-Agena B boost vehicle. After liftoff, the booster rolled to a launch azimuth of 95.6 deg (east of north) and performed a programmed pitch maneuver until booster cutoff. During the Atlas sustainer and vernier stages, adjustments in vehicle attitude and engine cutoff times were commanded as required by the ground guidance computer to adjust the altitude and velocity at Atlas vernier engine cutoff. After Atlas-Agena separation, there was a short coast period prior to the first Agena ignition. At a preset value of sensed velocity increase, the Agena engine was cut off. At this time both the Agena and spacecraft were coasting in a nearly circular parking orbit at an altitude of 188 km and a speed of 7.8 km/sec (space-fixed). After a total coast time of 25.9 min in the parking orbit, the second Agena ignition occurred. This parking-orbit coast time was determined after liftoff by the ground guidance computer and transmitted to the Agena during the Atlas vernier stage. The launch phase was nearly nominal with the exception of the parking-orbit altitude, which was slightly above nominal (approximately 21 km high at second Agena ignition; see Fig. 1). A typical sequence of events is shown in Fig. 2.

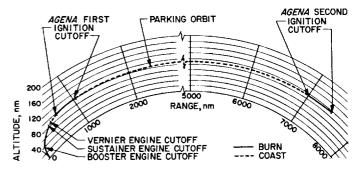


Fig. 1. Ascent trajectory profile

B. Cruise Phase

Injection (Agena final cutoff) occurred at 17:34:46 GMT, at which time the Agena and spacecraft were traveling at a speed of 10.962 km/sec (space-fixed). The geocentric latitude and longitude of injection were -21.4 and 36.6 deg, respectively, with injection taking place over the eastern coast of South Africa. The Agena and spacecraft separated 2½ min after injection occurred. The Agena then performed a programmed 180-deg yaw maneuver and ignited its retrorocket. The retrorocket impulse was designed to eliminate interference with the spacecraft operation and reduce the chance of lunar impact by the Agena. The geocentric characteristics of the

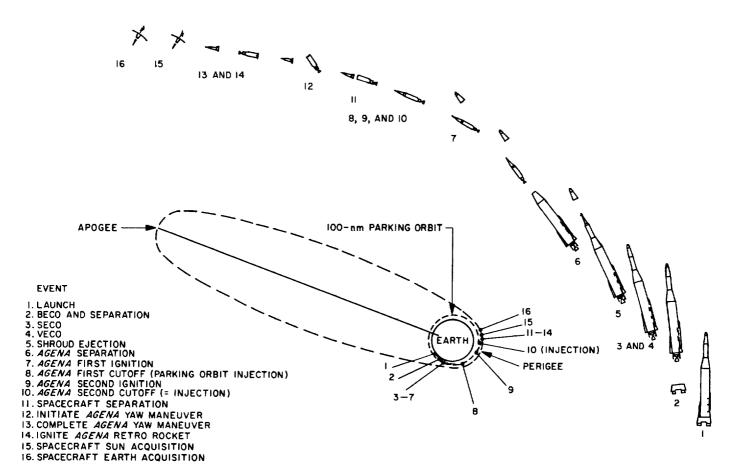


Fig. 2. Sequence of events

Ranger 5 premidcourse orbit are presented in Table 1. (Definitions of the symbols used in Tables 1, 2, and 3 are given in Table 4.)

Within 1 hr after injection, the spacecraft was receding from the Earth in an almost radial direction with decreasing speed. This reduced the geocentric angular rate of the spacecraft (in inertial coordinates) until, at 1.4 hr after injection, the angular rate of the Earth exceeded that of the spacecraft. This caused the Earth track of the spacecraft to reverse its direction from increasing to decreasing Earth longitude (Fig. 3). Note in Fig. 3 where the spacecraft entered the Earth's shadow during the parking orbit and emerged shortly after injection.

The Ranger 5 premidcourse orbit, as determined using the limited tracking data available, indicated that the spacecraft would miss the Moon by 631 km on the trailing edge, 7.84 deg below the lunar equator. Only 40% of the spacecraft midcourse maneuver capability would have been required to obtain a lunar impact in the target

area if the spacecraft had performed properly. However, 40 min past injection a malfunction in the electrical power distribution system made it necessary for the spacecraft to operate from battery power. Because of the relatively short battery life, a nonstandard midcourse maneuver was attempted. The batteries operated within design limits, but were depleted at 8 hr, 15 min past injection during the execution of the midcourse maneuver sequence before any significant maneuver was performed. The midcourse motor burn would have been completed at 8 hr, 21 min past injection.

After the midcourse maneuver attempt, some perturbations due to unbalanced attitude control torques or sources connected with the midcourse rocket system are presumed to have slightly altered the trajectory. Subsequent tracking of the capsule beacon through lunar encounter provided data to determine the postmidcourse orbit. This orbit indicates that the spacecraft missed the Moon's surface by 735 km on the trailing edge, 7.79 deg below the lunar equator. At the time the midcourse maneuver was attempted, the spacecraft was at a distance of

Table 1. Ranger 5 trajectory characteristics, geocentric

			Epoch			Central body		Epoch	Epoch of pericenter passage	e Boss	Period
Event	Orbit	×	>	z	·×	٠,	ż	B	i	3	Apogee
			+9-	8	>	٨	٥	•	T.	À	Perigee
Injection®	Premidoourse	10-18-62	17:37:23.000	3.000		Earth		10-18-62	17:34	17:34:14.322	25022.369
		5807.2046	-1871.2696	-2891.7849	3.6254794	9.7646336	-2.8587352	283386.38	28.267556	227.09996	6595.0505
		6751.8656	-25.359354	51.068163	10.373187	9.0762677	102.48129	0.97672770	100.32216	17.636897	560177.71
Midcourse	Postmidcourse	10-19-62	01:56:08.000	8.000		Earth		10-18-62	17:3	17:34:17.998	24359.324
		- 50885.925	91512.070	18173.407	-1.6380212	1.6998046	0.70499460	278358.62	28.324781	227.12511	6585.3187
		106273.74	9.8462939	62.976418	7.4244739	18.603799	272.47972	0.97634232	100.29226	154.00054	550131.92
At closest	Postmidoourse	10-21-62	15:53:43.304	13.304		Earth					
to Moon"		-266515.31	267754.70	120687.60	-1.9612389	1.0016943	0.85827348				
		396596.12	17.716569	226.82526	26.960412	4.7987901	270.38492				
After	Postmidcourse	11-8-62	00:00:00:00	0000		Earth					
with Moon		-2285917.5	- 594766.98	120984.52	-1.2436209	-0.63197976	-0.05552585				
		2365122.4	2.9321635	147.88414	171.94858	0.45252991	269.95829				
Prior to	Postmidcourse	8-5-63	00:00:00:00	0.000		Earth					
with Earth		6441339.0	-202736.00	-1182141.5	-1.5404861	0.23252583	0.17680949				
		6552054.1	-10,394401	45.372136	469.76109	-1.8948481	269.98716				
At closest	Postmidcourse	10-11-63	07:39:44.200	4.200		Earth		10-11-63	07:3	07:39:44.191	NA°
to Earth		-45776.424	-1428093.1	-627674.04	-9.4190401	-0.07787804	0.24588253	900029.1	151.67288	302.04701	NA°
		1560615.3	-23.715566	134.0523	105,13094	-0.48371003E-6	270,14636	2.7339608	213.58342	0,25613208E-5	1560615.3
After	Postmidcourse	12-3-63	00:00:00	0.000		Earth					
with Earth		-4866905.0	-1692771.0	230269.50	-1.1175573	-0.42921603	-0.05969867				[
		5158029.1	2.5587015	128.07617	375.71805	0,18188111	269.98275				
* See Fig. 4.	b See Fig. 19.		c Not applicable								

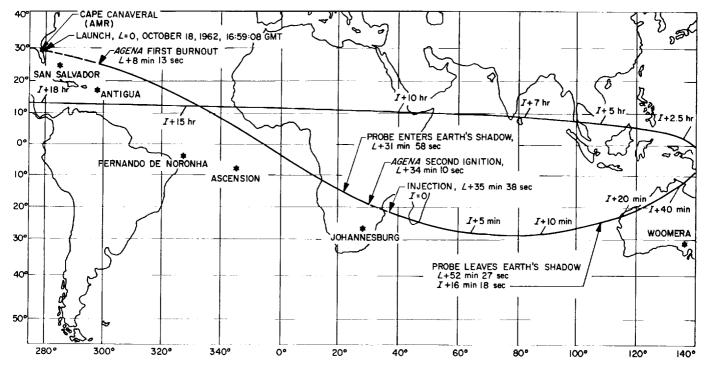


Fig. 3. Earth track of Ranger 5

106,000 km and traveling at a speed of 2.46 km/sec relative to the Earth. The spacecraft continued moving primarily under the gravitational influence of the Earth in a highly elliptical geocentric orbit on its transit to lunar encounter. Figure 4 illustrates the geometrical relations of the trajectory through lunar encounter. Figures 5 through 9 show geocentric radii (distance to probe), geocentric inertial speed, Earth-probe-Sun angle, Earthprobe-Moon angle, and Sun-probe-Moon angle versus flight time from injection to lunar encounter. At 63 hr past injection and 370,000 km from Earth, the speed of the spacecraft reached a minimum of 0.92 km/sec with respect to Earth (Fig. 6). The spacecraft accelerated thereafter due to the gravitational influence of the Moon. The geocentric postinjection orbit characteristics before and after lunar encounter are also given in Table 1.

C. Encounter Phase

On October 21, 1962, the spacecraft encountered the Moon, approaching in a hyperbolic selenocentric orbit with closest approach near the Moon's trailing edge. Figure 10 shows the selenocentric geometry of the flight past the Moon and illustrates the position of the spacecraft as it passed through the Moon's shadow and when it was occulted from Earth by the Moon. Closest approach occurred at 15:53:43 GMT, some 70.3 hr past injection,

at a distance of 2,473 km from the center of the Moon or 735 km from its surface. Figures 11 through 17 show the trajectory characteristics during lunar encounter, including geocentric radii (distance to probe), geocentric inertial speed, Earth-probe-Sun angle, Earth-probe-Moon angle, Sun-probe-Moon angle, selenocentric altitude, and selenocentric inertial speed versus GMT. Figure 18 shows the time derivative of the range rate of the spacecraft as seen at the Goldstone Tracking Station, Because the Goldstone Tracking Station was recording the doppler signal from the capsule beacon during lunar encounter, the range rate derivative could be derived in spite of slow frequency drift of the capsule transmitter. The beginning and end of spacecraft occultation by the Moon were also observed. The capsule beacon was tracked until its signal reached DSIF threshold 11 days after launch. These data were most useful in determining the postmidcourse orbit. The selenocentric characteristics of the trajectory are shown in Table 2.

D. Postencounter Phase

The Moon's gravitational influence altered the space-craft's initial geocentric orbit during lunar encounter. This resulted in an increase in energy of the spacecraft relative to Earth, so that after lunar encounter the spacecraft's orbit became hyperbolic with respect to Earth.

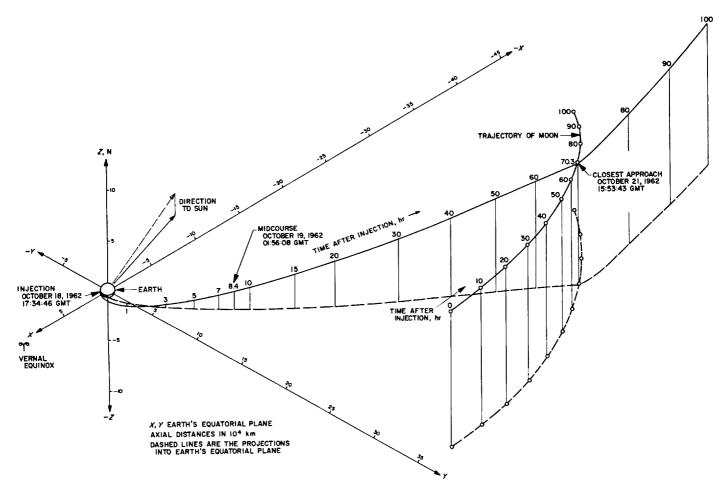


Fig. 4. Ranger 5 Earth-Moon transit geometry

Note in Fig. 12 that an increase in geocentric inertial speed was effected by encounter. After lunar encounter the spacecraft was able to escape from the gravitational influence of the Earth-Moon system and become a satellite of the Sun in an elliptical orbit similar to the Earth's, but with greater eccentricity. The heliocentric elliptical orbit characteristics after lunar encounter are given in Table 3. The spacecraft reached a perihelion distance of 1,420,000 km on January 5, 1963. Figure 19 shows the relative positions of the spacecraft and Earth through November 25, 1963, in their respective orbits about the Sun. In March 1963 the spacecraft was at its maximum distance of 20 million km from the Earth. Its geocentric distance decreased until October 11, 1963, at which time the spacecraft encountered the Earth with a closest approach distance of 1.5 million km. The geocentric characteristics of the trajectory at encounter with Earth are listed in Table 1. Just as the geocentric orbit was altered by lunar encounter, the heliocentric orbit was altered by the October 1963 geocentric encounter. The heliocentric orbit characteristics before and after encounter with Earth are given in Table 3.

A study of the Ranger 5 trajectory can be made by examination of the detailed trajectory printout presented in Appendices B and C. Appendix B contains the trajectory listing corresponding to the premideourse orbit from injection to the mideourse epoch. Appendix C contains the trajectory listing corresponding to the postmideourse orbit from mideourse through lunar encounter into the heliocentric orbit, and on through the first return encounter with Earth. Table D-1 (Appendix D) is a key to the trajectory printout. Table D-2 contains the definitions of the printed quantities. Constants and conversion factors used in all Ranger 5 trajectory computations are listed in Table D-3. The miss parameter B, used to measure the miss distance for the lunar trajectory, is defined in Appendix A.

Table 2. Ranger 5 trajectory characteristics, selenocentric

			Epoch			Central body		Epoch	Epoch of pericenter passage	passage	Period
Event	Orbit	×	٨	Z	·×	٠,	ż	8		3	Apogee
		•	ф	8	^	٨	ь	•	а		Perigee
At closest approach	_	10-21-62	15:53:43.304	13.304		Moon		10-21-62	15.4	15:53:43.305	NA
to Moon*	Postmidcourse	2052.2633	1349.3010	291.63704	-1.2056512 1.6114740	1.6114740	1.0284995	-4278.6451 28.039960 14.526641	28.039960	14.526641	A A
		2473.3475	-7.7901483	87.497005	-7.7901483 87.497005 2.2536800	-0.17023913E-4 82.230666 1.5780680 20.441789 -0.34150945E-4 2473.3479	82.230666	1.5780680	20.441789	-0.34150945E-4	2473.3479
* See Fig. 4.	b Not applicable.	ble.									

Table 3. Ranger 5 trajectory characteristics, heliocentric

			Epoch			Central body		Epoch	Epoch of pericenter passage	*sage	Period
Event	Orbit	×	٨	Z	·×	٠٨	·z	в	-	3	Apogee
		•	\$	•	^	٨	۵	•	C	,	Perigee
After		11-8-62	00:00:00:00	0.000		Sen		1-4-63	13:43:26.688	6.688	368.97689
with Moon*	Postmidcourse	0.10232351E9	0.10450206E9	347960.50	-22.833729	20.303836	0.20121688	0.15061302E9	0.40770429	81.242877	0.15905266E9
		0.14625629E9	0.13631521	45.603487	30.555939	-2.7519301	89.615704	0.05603532	26.073698	-61.712644	-61.712644 0.14217336E9
Prior to encounter		8-5-63	00:00:00	0.000		Sun		1-5-64	04:32:13.250	3.250	365.41625
with Earth*	Postmidcourse	0.10774806E9	-0.11365815E9	-0.100378	20.143050	20.053632	0.07081693	0.14964250E9	0.39079587	83.709337	0.15751821E9
		0.15661681E9	-0.36722170	313.47094	28.423505	-1.4024846	89.866190	0.05263014	23.450232	-153.68906	-153.68906 0.14176680E9
Closest		10-11-63	07:39:44.200	4.200		Sun					
to Earth*	Postmidoourse	0.14256450E9	42783543.	-7765.1250	-10.264175	28.371885	0.25552714				
		0.14884578E9	-0.00298906	16.704476	30.172545	-3.1841761	89.514180		 	!	
After encounter		12-3-63	00:00:00:00	000.		Sun		12-8-63	03:07:41.637		377.27291
with Earth*	Postmidcourse	45346745.	0.13719529E9	884777.49	-29.599953	9.6200026	0.11477125	0.15286221E9	0.41033545	64.205699	0.16126079E9
		0.14449795E9	0.35083096	71.709883	31.124184	-0.28462081	89.786970	0.05494215	12.976893	-5.4720606	-5.4720606 0.14446364E9
* See Fig. 19.											

Table 4. Legend for tables 1, 2, and 3

		Definition	
Parameters	Earth as central body	Moon as central body	Sun as central body
X,Y,Z	Vernal equinox cartesian coordinates in a geocentric equatorial system. The origin is the center of the central body. The principal direction X is the vernal equinox direction of date, and the principal plane X, Y is the Earth equatorial plane of date.	Vernal equinox cartesian coordinates in a geocentric equatorial system. The origin is the center of the central body. The principal direction X is the vernal equinox direction of date, and the principal plane X, Y is the Earth equatorial plane of date. Z is along the direction of the Earth's spin axis of date.	Vernal equinox cartesian coordinates in a heliocentric equatorial system. The origin is the center of the Sun. The principal direction \vec{X} is the vernal equinox direction of date, and the principal plane X , Y is the ecliptic plane of date, \vec{Z} is normal to the ecliptic plane of date, km.
x, y, z	First time derivatives of X, Y, and Z, respectively; i.e., cartesian components of the probe space-fixed velocity vector, km/sec.	First time derivatives of X, Y, and Z, respectively; i.e., cartesian components of the probe space-fixed velocity vector, km/sec.	First time derivatives of X, Y, and Z, respectively; i.e., cartesian components of the probe space-fixed velocity vector, km/sec.
ι,	Probe radius distance, km	Probe radius distance, km	Probe radius distance, km
6	Probe geocentric latitude, deg	Probe selenocentric latitude, deg	Probe celestial latitude, deg
8	Probe east longitude, deg	Probe selenocentric east longitude, deg	Probe celestial longitude, deg
>	Probe Earth-fixed velocity, km/sec	Probe selenocentric-fixed velocity, km/sec	Probe heliocentric inertial velocity vector, km/sec
۸	Path angle of the probe Earth-fixed velocity vector with respect to the local horizontal, deg	Path angle of the probe selenocentric-fixed velocity vector with respect to the local horizontal, deg	Path angle of the probe heliocentric inertial velocity vector with respect to the local horizontal, deg
ь	Azimuth angle of the probe Earth-fixed velocity vector measured east of true north, deg	Azimuth angle of the probe selenocentric-fixed velocity vector measured east of the Moon's mean spin axis, deg	Azimuth angle of the probe heliocentric inertial velocity vector measured east of the celestial north direction, deg
0	Semimajor axis, km (negative for hyperbolic orbit)	Semimajor axis, km (negative for hyperbolic orbit)	Semimajor axis, km
0	Eccentricity	Eccentricity	Eccentricity
-	Inclination of the orbit plane to the Earth equatorial plane, deg	Inclination of the orbit plane to the Earth equatorial plane, deg	Inclination of the orbit plane to the ecliptic, deg
a	Longitude of the ascending node, deg	Longitude of the ascending node, deg	Longitude of the ascending node, deg
3	Argument of pericenter, deg	Argument of pericenter, deg	Argument of pericenter, deg
•	True anomaly, deg	True anomaly, deg	True anomaly, deg
Period	Measured in sec	Measured in sec	Measured in days
Apogee	Measured in km	Measured in km	Measured in km
Perigee	Measured in km	Measured in km	Measured in km

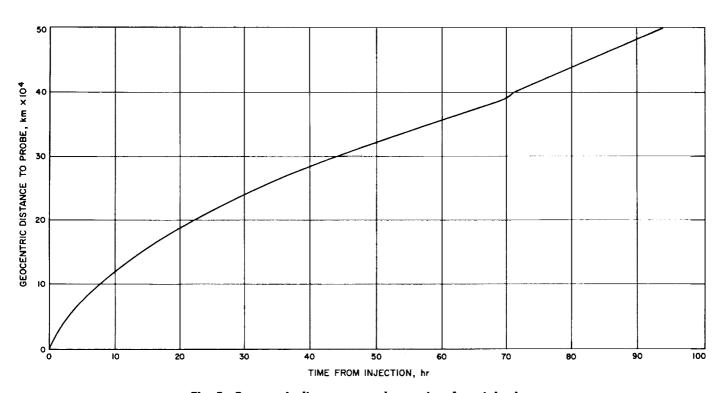


Fig. 5. Geocentric distance to probe vs. time from injection

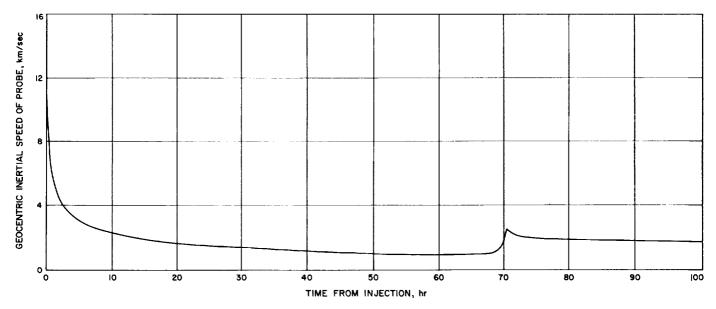


Fig. 6. Geocentric inertial speed of probe vs. time from injection

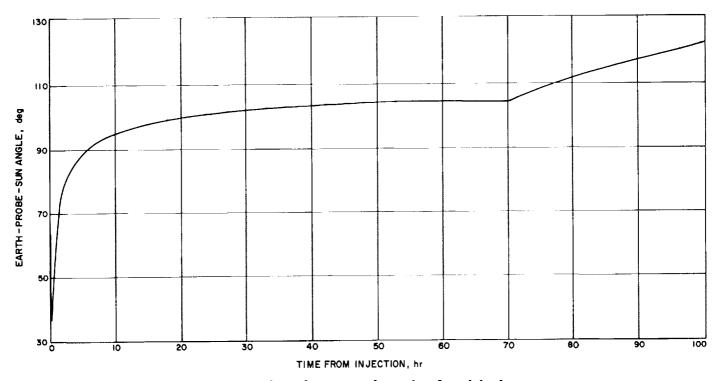


Fig. 7. Earth-probe-Sun angle vs. time from injection

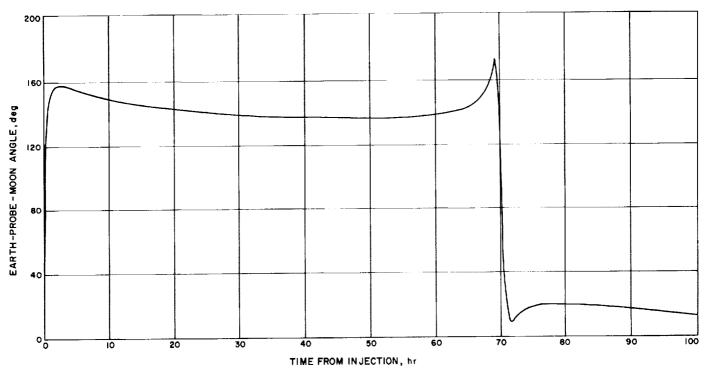


Fig. 8. Earth-probe-Moon angle vs. time from injection

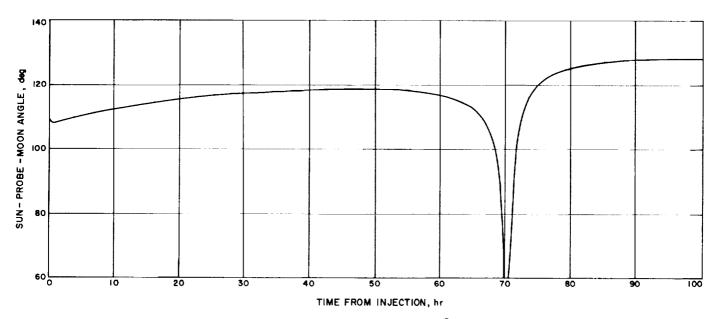


Fig. 9. Sun-probe-Moon angle vs. time from injection

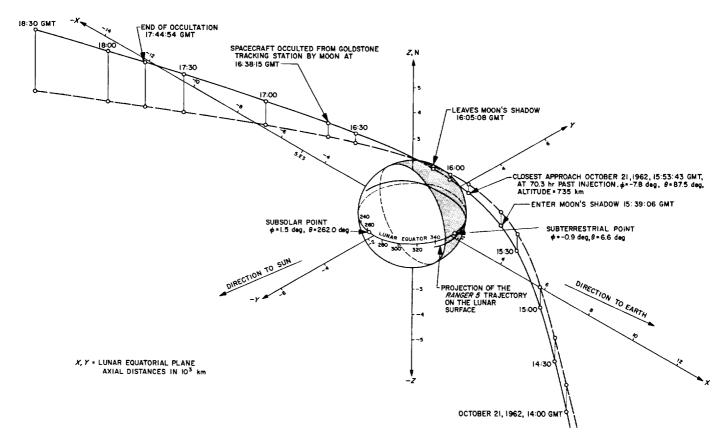


Fig. 10. Ranger 5 lunar encounter

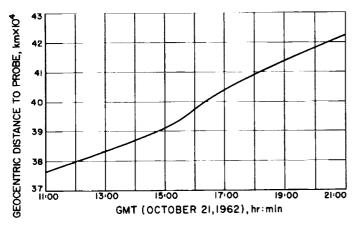


Fig. 11. Geocentric distance to probe vs. GMT during lunar encounter

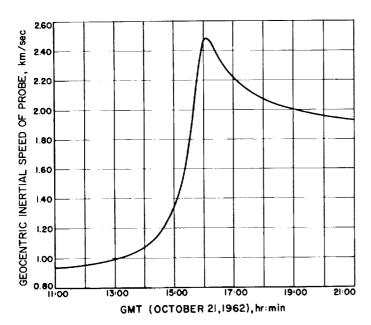


Fig. 12. Geocentric inertial speed of probe vs.

GMT during lunar encounter

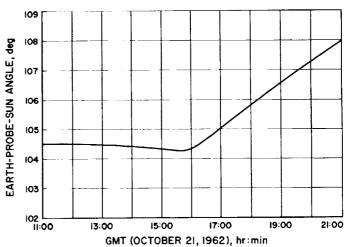


Fig. 13. Earth—probe—Sun angle vs. GMT during lunar encounter

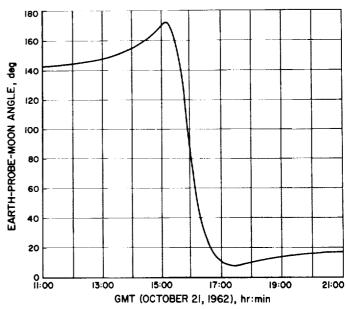


Fig. 14. Earth-probe-Moon angle vs. GMT during lunar encounter

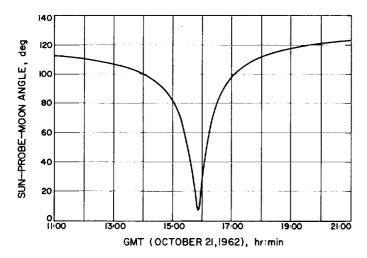


Fig. 15. Sun—probe—Moon angle vs. GMT during lunar encounter.

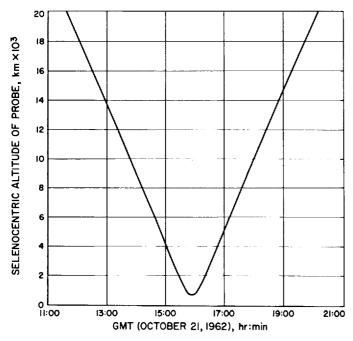


Fig. 16. Selenocentric altitude of probe vs. GMT during lunar encounter

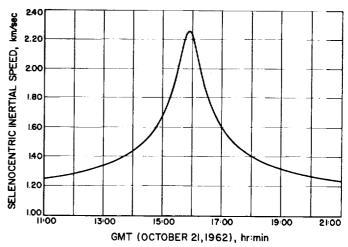


Fig. 17. Selenocentric inertial speed of probe vs. GMT during lunar encounter

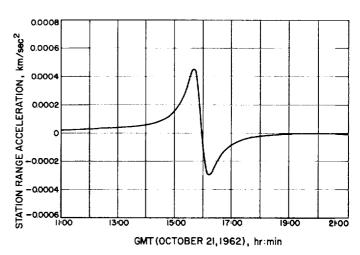


Fig. 18. Range acceleration at Goldstone for the probe vs. GMT during lunar encounter

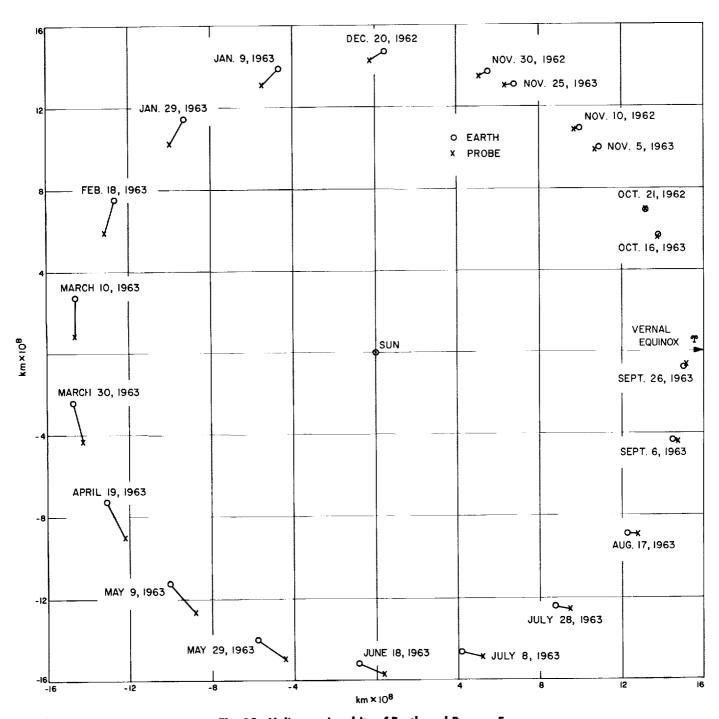


Fig. 19. Heliocentric orbits of Earth and Ranger 5

III. THE TRACKING SEQUENCE OF EVENTS

A. Introduction

This section summarizes the key events in the tracking of the Ranger 5 and the Agena stage. Part B describes the DSIF postinjection tracking of the Ranger 5 transponder and the payload "rough landing" capsule beacon. Part C summarizes the AMR post parking-orbit tracking of the Agena C-band transponder by the Twin

Table 5. Review of key events

	Events	Date	GMT ^s	Remarks
1	Atlas lift-off (L)	Oct. 18 1962	16:59:07.84	L + 0
2	Agena stage parking- orbit injection (I ₁)		17:07:20.9	L + 8 ^m 13.1°
3	Agena stage trans- lunar orbit injection (I ₂)		17:34:46.0	L + 35™38"
4	First reference epoch for orbit determi- nation (E ₁)		17:34:49.0	L + 35 ^m 41*
5	Second reference epoch for orbit determination (E ₂)		17:37:23.0	L + 38 ^m 15*
6	Mechanical separation of Agena and spacecraft		17:37:23	L + 38 ^m 15 ^s
7	Ignite Agena retro- motor		17:43:37	L + 44 ^m 45 ^s
8	Burnout of Agena retromotor		17:43:57	L + 45 ^m 05 ^s
9	Loss of transponder due to battery depletion	Oct. 19	01:46	L + 8 ^h 11 ^m
10	Loss of capsule beacon signal due to oc- cultation by Moon at DSIF 2	Oct. 21	16:38:14.78 ^b	L + 2 ^d 23 ^h 3 ^m
11	Reacquisition of cap- sule beacon signal at DSIF 2	Oct. 21	17:44:57.06°	L + 3 ^d 0 ^h 10 ^m
12	Lost contact with beacon	Oct. 29		L + 11 ^d

[&]quot; Universal time at occurrence of event.

Falls Victory (TFV) ship, the Ascension Island, FPS-16, and the Pretoria Tracking Station.

To help interpret the results of the analysis of the tracking data given in Section IV of this Report, Table 5 summarizes the key events of the launch to lunar impact sequence. When comparing the Agena orbit with the spacecraft orbit it is important to note that Agena C-band transponder tracking occurs under the following conditions:

- 1. Before I_2 , when the Agena rocket motor is thrusting,
- Between I₂ and event 6, when the spacecraft and the Agena are still mechanically attached (the path of the combination differs from the final spacecraft orbit due to the imparting of about 0.3 m/sec relative velocity at mechanical separation),
- 3. Between event 6 and event 7, when the *Agena* orbit is slightly changed by the mechanical separation velocity,
- 4. Between events 7 and 8, when the Agena orbit is being changed by the retrorocket thrust,
- 5. After event 8, when the Agena orbit has undergone significant change from its orbit prior to event 7.

In using the Agena C-band transponder data it is quite important to employ only the data corresponding to the desired Agena orbit.

B. DSIF Tracking of Ranger 5 Transponder and Payload Beacon

I. General Information

The names and locations of the DSIF stations employed in the Ranger 5 missions are given in Table 6. The detailed characteristics of the stations may be obtained from Ref. 2.

Table 7 shows the nominal visibility periods of the spacecraft to the DSIF stations during the course of the mission. Rise time refers to the time that the spacecraft first appears at a 5 deg geometrical elevation angle. All times are given in GMT. Since these are nominal periods, it is possible that signals may be received prior to "rise" and after "set" times.

 $^{^{\}mathrm{b}}$ Corrected by -1.32 sec to account for signal travel time to DSIF 2.

 $^{^{\}rm c}$ Corrected by -1.34 sec to account for signal travel time to DSIF 2.

Table 6	DSIF	station	locations
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DSIF station	Location ^a	Geodetic latitude, deg	Astronomic longitude, deg	Paraboloidal antenna type	Antenna size, ft	Transmitter power
1	Johannesburg, S. Africa	25.9\$	27.7E	AZ-EL	10	25
2	Goldstone, Calif., USA	35.4N	243.2E	HA-DEC	85	NAb
3	Goldstone, Calif., USA	35.3N	243.2E	HA-DEC	85	200
4	Woomera, Australia	31.45	136.9E	HA-DEC	85	50
5	Johannesburg, S. Africa	25.98	27.7E	HA-DEC	85	200

Table 7. Nominal view periods at DSIF stations^a

Date of rise	DSIF station	Rise GMT	Set GMT	View period
Oct. 18, 1962	1,5	Before injection	17:35:18	0 ^h 0 ^m 29 ^s
	4	17:46:31	01:55:40 ^b	8 ^h 09 ^m 09*
	1,5	22:56:47	09:41:31 ^b	10 ^h 44 ^m 44 ^s
Oct. 19	2,3	08:23:17	20:26:42	12 ^h 03 ^m 25 ^s
	4	16:41:38	02:32:16 ^b	9 ^h 50 ^m 38 ^s
	1,5	23:56:59	10:02:28 ^b	10 ^h 05 ^m 29 ^s
Oct. 20	2,3	08:45:14	20:42:02	11 ^h 56 ^m 48*
	4	17:01:19	02:39:47 ^b	9 ^h 38 ^m 29 ^s
Oct. 21	1,5	00:10:46	10:07:46	9 ^h 57 ^m 00*
	2,3	08:52:01	16:38:16°	7 ^h 46 ^m 15 ^s

Based on 5-deg elevation angle and postflight transponder determined orbit. GMT at spacecraft.

The modes of operation of the DSIF are identified as ground modes (GM) and are defined as follows:

GM-1. Ground receiver tracks the transponder signal in the 2-way mode, obtaining angles, telemetry and 2-way doppler. This type of doppler is obtained when the station receives a signal from the transponder which is being interrogated by a ground transmitter radiating through the same antenna utilized by the receiver (diplexer operation). This mode is possible at DSIF 1, 4, and 5.

GM-2. Ground receiver listens^a for the transponder signal in the 2-way mode, obtaining telemetry and 2-way doppler. This mode is possible at DSIF 3, 4, and 5.

GM-3. Ground receiver tracks the transponder signal in the 1-way mode, obtaining angles, telemetry and

1-way or 3-way noncoherent doppler. The 1-way doppler is obtained when the station receives a signal from the transponder which is not being interrogated by a ground transmitter. Accuracy of 1-way doppler is limited due to the unknown drift in the spacecraft crystal frequency. The 3-way noncoherent doppler is obtained when the station receives a signal from the transponder which is being interrogated by a ground transmitter remotely located with respect to the receiver and with no reference frequency between the two stations. Accuracy of this type of doppler is limited by variations in the reference frequency of the transmitting station. This mode is possible at DSIF 1, 4, 5.

GM-4. Ground receiver listens^a for the transponder in the I-way mode, obtaining telemetry and 1- or 3-way noncoherent doppler. This mode is possible at DSIF 2, 3, 4, and 5.

GM-5. Ground receiver listens^a for the transponder in the 1-way mode, obtaining telemetry and 3-way coherent doppler. The 3-way coherent doppler is obtained when the station receives a signal from the transponder which is being interrogated by a ground transmitter located away from the receiver but with a reference frequency between the two stations. This mode is possible only with the combination DSIF 3 transmitting and DSIF 2 receiving.

GM-6. Ground transmitter is transmitting only to the transponder. No signal is received, and no doppler obtained. Used to send spacecraft commands. This mode is possible at DSIF 3 and 5.

^b Set occurs on the day after rise.

Loss of capsule beacon signal due to occultation by Moon.

[&]quot;Tracking and listening capability is strictly a function of existing hardware at a station. DSIF 1 did not have listening capability.

GM-7. Ground receiver listens^a for the capsule beacon signal in the 1-way mode, obtaining telemetry and 1-way doppler. In addition to the limitations previously cited for 1-way doppler, data obtained in this mode are degraded because of the lower, and varying, signal level of the capsule beacon. This mode is possible at DSIF 2, 3, 4, and 5.

To determine the Ranger 5 spacecraft orbit, all 2-way doppler data from DSIF 1, 4, and 5 plus the angular data from DSIF 5 were used. Angular data from DSIF 1 were rejected because carefully calibrated, more accurate, data were available from DSIF 5. Angular data from DSIF 4 were rejected due to extremely large variations caused by problems in the servo tracking speed. (See residual plots, Fig. 20–30.)

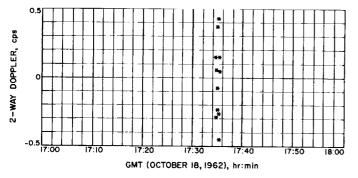


Fig. 20. Station 1 residuals (17:00 GMT)

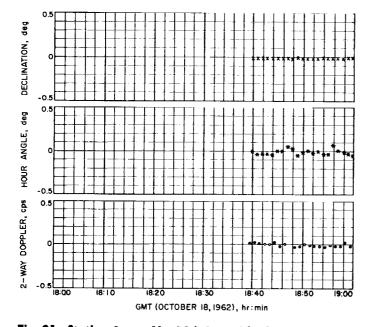


Fig. 21. Station 4 pass No. 10/181 residuals (18:00 GMT)

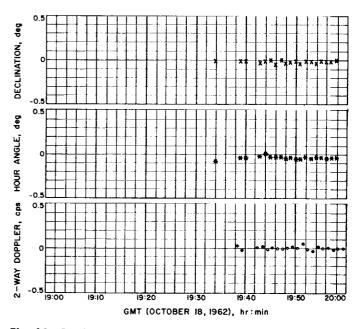


Fig. 22. Station 4 pass No. 10/181 residuals (19:00 GMT)

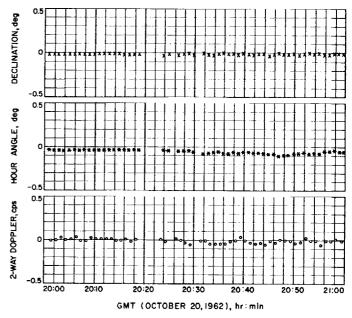


Fig. 23. Station 4 pass No. 10/181 residuals (20:00 GMT)

2. Transponder Tracking

Table 8 and Fig. 31 summarize the transmitter number q versus time, as well as the acquisition times on the first pass. The most critical times are initial acquisition in GM-3 and initial times in GM-1 during pass 1. The time interval where q=0 represents the transmitter "off"

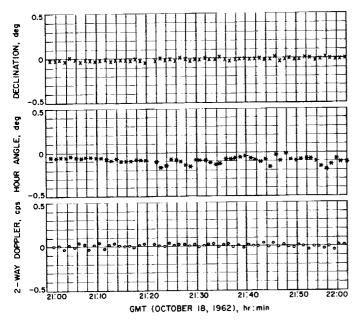


Fig. 24. Station 4 pass No. 10/181 residuals (21:00 GMT)

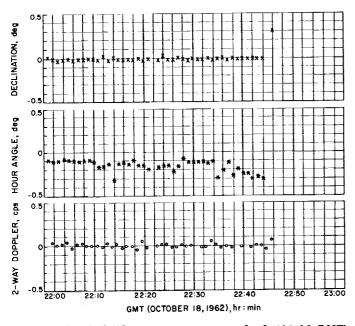


Fig. 25. Station 4 pass No. 10/181 residuals (22:00 GMT)

interval; i.e., no station is transmitting. In this instance no data were lost because the spacecraft was not within the DSIF view period. There was only a 44-sec time lag in shifting the transmitting assignment from DSIF 4 to DSIF 5. This is not evident in Table 8, since the intervals shown in the table represent the periods during which good 2-way doppler data were taken, and omit a few bad points. Station logs confirm the short time lag.

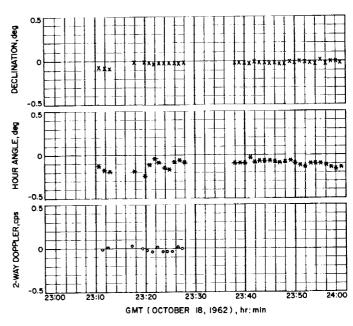


Fig. 26. Station 4 pass No. 10/181 residuals (23:00 GMT)

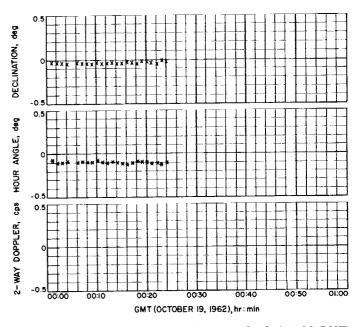


Fig. 27. Station 4 pass No. 10/181 residuals (00:00 GMT)

3. Capsule Beacon Tracking

On October 19, the DSIF stations began alternating between capsule beacon tracking, and searching for the transponder. In most cases, the transponder was not heard. DSIF 2 was tracking the capsule beacon when occultation occurred at 16:38:16.1 GMT on October 21. The spacecraft was reacquired at 17:44:58.4 GMT on the same day. After October 22, DSIF 2 tracked the capsule

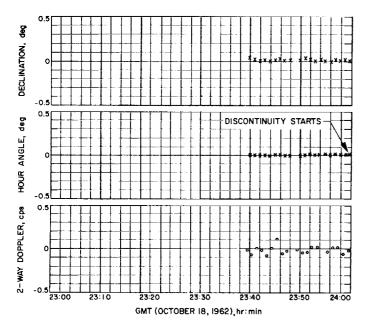


Fig. 28. Station 5 pass No. 10/182 residuals (23:00 GMT)

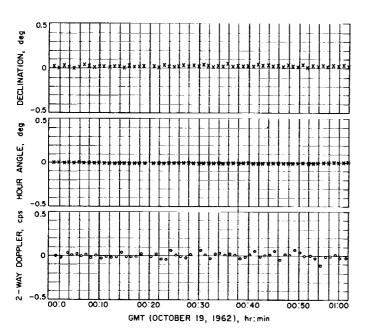


Fig. 29. Station 5 pass No. 10/182 residuals (00:00 GMT)

beacon each day for approximately 2 hr until threshold was reached on October 29. Table 9 summarizes the periods of beacon tracking for the DSIF.

Verification by Time of Signal Loss and Signal Reacquisition

The primary evidence of occultation of the spacecraft by the Moon is the loss of received capsule signal at the

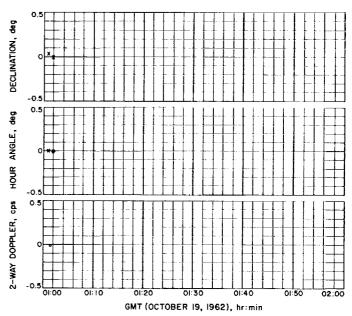


Fig. 30. Station 5 pass No. 10/182 residuals (01:00 GMT)

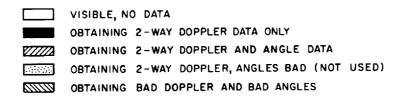
ground station. Various functions related to the received signal are recorded by the DSIF on magnetic tape and independently on direct-write oscillographs.

Figure 32 is a playback of a recording of the receiver functions recorded on magnetic tape at DSIF 2 for the last few seconds prior to occultation. In the figure, the trace labeled SIGNAL STRENGTH is the one of critical interest. At the time noted by the arrow (16:38:16), the signal started to decay. The rate of decay is characteristic of the 10-sec AGC time constant in the receiver.

The time associated with the event is determined from a binary-coded-decimal (BCD) time code which records days, hours, and minutes, from a 1 pulse/sec time code. Both the BCD code and 1 pulse/sec code are derived from the station secondary standard which is synchronized to WWV. The playback mechanism for the magnetic tape precludes the direct display of the BCD code. In order to display time with the playback, a time translator is synchronized with the BCD code. The output of the time translator appears as a 1 pulse/sec trace in Fig. 32. A 6 pulses/min and a 1 pulse/min trace, both from the translator, may also be seen in the figure. The trace labeled ACQUISITION RELAY is an event channel which marks loss of receiver lock; i.e., loss of signal. The change of state of this relay is consistent with the time of signal tail-off seen in the SIGNAL STRENGTH

Transmitter number q	Time interval	Receiving station i	Ground mode GM	Acquisition ^c time GMT 18 and 19 Oct	Acquisition time vs. rise time
1	Injection epoch to to the transfer of the tran	1	1	17:31:25	Set — 3 ^m 53 ^s
0	t ₁ to t ₂ = 17:48:00	-		-	_
4	t ₂ to t ₃ = 23:28:51	4 5 1	1 3 3	18:38:51 23:00:02 22:33:23 ^d	Rise + 52 ^m 20 ^s b Rise + 3 ^m 15 ^s Rise — 23 ^m 24 ^s
5	t ₃ to t ₄ = 01:43:51	4 5 1	3 1 3	23:38:02 23:32:21 23:30:07	11 ^m 2* * 5 ^m 41* * 7 ^m 55* *

Table 8. Transmitter number and acquisition times^a



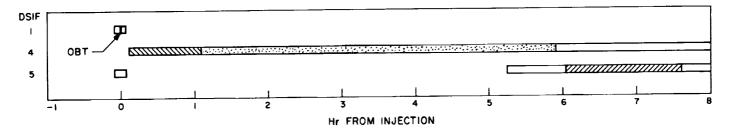


Fig. 31. Tracking station view periods for Ranger 5 mission and their data coverage

trace. Certain areas to the right of occultation time appear to indicate receiver "in-lock". This is most likely caused by noise bursts or transients. It is definitely not a signal since the signal strength trace is wandering about the reference line.

The conclusion is that the spacecraft was occulted by the Moon at 16:38:16.1 GMT on October 21 minus the signal transmission time. The accuracy to which this time can be determined is approximately ± 0.1 sec.

Figure 33 is a playback of the same magnetic tape from DSIF 2 for a few seconds prior to the time that the spacecraft reappeared after being occulted by the Moon. Again the SIGNAL STRENGTH trace is the one of primary interest. At the time noted by the arrow (17:44:58.4), a signal was received from the capsule. Prior to this time, the SIGNAL STRENGTH trace had been wandering about the reference line. Capsule signal acquisition time is also verified by the change in state of the ACQUISITION RELAY.

^{*} Reference 3 plus Section IV of this Report. Times measured from "rise" refer to rise time at the receiving station listed.

b Signal was heard at 17h 45m, but because of trouble in the servo tracking speed, two-way doppler lock was not achieved at first. Another significant factor was a series of false two-way lock indications with the transmitter VCO below the proper lock frequency.

c Acquisition time is the time of the first good data point.

^d Acquired below 5-deg elevation.

e Time shown here is the interval between transmitter "on" time and time of first good data point.

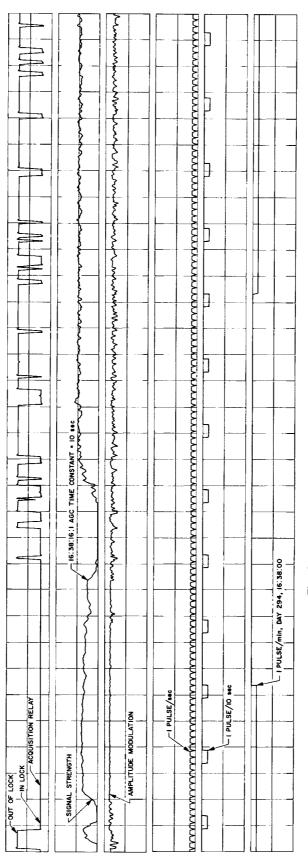


Fig. 32. Station 2 receiver functions prior to occultation

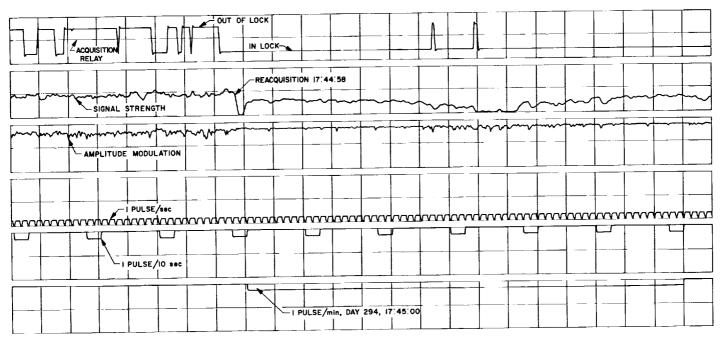


Fig. 33. Station 2 receiver functions after occultation

Table 9. Summary of capsule beacon tracking

Date	DSIF station	Acquisition GMT	End of track GMT
Oct. 19, 1962	4	01:19:00	01:22:14
	4	01:58:32	02:26:50
	4	16:35:00	02:30:00 ^b
	5	02:36:40	04:00:00
	5	04:35:00	04:44:00
	2	08:35:00	14:39:02
	3	08:46:00	08:50:12
	3	15:32:00	17:05:00
	3	18:54:30	19:08:20
Oct. 20	4	17:08:00	02:44:11 ^b
	5	00:37:40	10:00:00
	3	10:19:00	20:30:00
Oct. 21	4	18:25:52	02:40:00 ^b
	5	00:10:00	09:00:10
	2	08:53:00	16:38:16.1°
	2	17:44:58.4 ^d	20:45:00
	3	17:47:50	18:14:00
Oct. 22	5	05:30:00	07:30:00
	2	19:04:00	20:15:00
Oct. 23-29	2*		

Station tracked for approximately 2 hr each day. Threshold was reached on October 29.

It must be noted that reacquisition time is a function of the accuracy of the tracking predictions. Therefore, it cannot be definitely concluded from Fig. 33 that the time noted by the arrow represents the earliest possible moment that a signal could have been received. Hence, it merely indicates the time that the station reacquired the capsule signal.

DSIF 3 did not attempt to observe occultation because it was assigned to searching for the transponder signal.

DSIF 4 could not physically view the occultation. Endof-occultation was available, but the capsule signal was not locked up until 34 min after reappearance.

C. AMR Tracking

Tracking data were received in real time from the AMR radars on Antigua and Ascension Islands. Since injection occurred over South Africa, these stations did not track the Agena C-band transponder in the final transfer orbit. However, data from these stations, together with several minutes of data from the radar at Pretoria, were used to give a good estimate of the parking orbit, which in turn was used, together with an assumed nominal second stage burn of the Agena, to provide acquisition information for subsequent tracking.

b End of track occurred next day.

c Occultation.

d Reacquisition after occultation.

The radar at Pretoria tracked the C-band transponder when it rose 2 deg above the horizon and maintained track until about 20 sec before *Agena* second burnout. At that time the radar lost track, and from then on the data were poor until complete loss of signal, at the horizon, at about 17:37:00.

The TFV ship sent back raw radar data in real time. These data were uncorrected for the pitch and roll of the ship, due to the failure of the computer on board. Later, however, the data were corrected at Cape Canaveral and retransmitted. These data covered the time span from 17:40:18 to 17:45:48 thus including the occurrence of the spring separation. These data have not yet been analyzed. Table 10 summarizes the available AMR raw data.

Table 10. Ranger 5 AMR raw data summary

AMR station	Time points during parking orbit	Remarks
Antigua	28	
Ascension	33	
Pretoria	14	Good data continued on thru 2nd Agena burn, until about 17:34:24.
	After epoch E ₁	
Pretoria	19	Only 4 data time points in- dicated "in lock".
TFV Ship	51	

IV. FLIGHT PATH DETERMINATION USING TRANSPONDER TRACKING

A. Introduction

The real-time determination of the parking orbit is the responsibility of the AMR. Their preinjection tracking of the Agena vehicle C-band transponder is important in establishing the parking orbit and detecting nonstandard flight conditions. The AMR supplies the Jet Propulsion Laboratory (JPL), Pasadena, California, with parking

orbit elements and initial acquisition information for transmittal to the DSIF stations and for preliminary estimation of the spacecraft injection conditions. The only postinjection tracking of the spacecraft is done by the DSIF. Postinjection orbit tracking of the Agena may be incorporated with the DSIF data. In the present case the paucity of postinjection data from land stations and the difficulty of accounting for the additional error sources

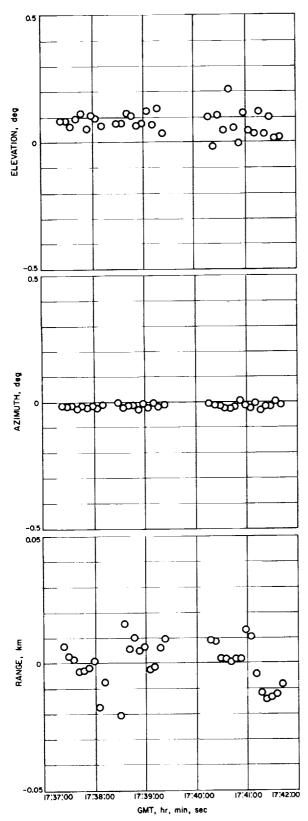


Fig. 34. TFV ship residuals based on orbit determination by ship range and DSIF data

in ship tracking data (ship's velocity) has precluded comparisons between the AMR data and the DSIF orbit. However, an orbit was obtained which fitted the TFV ship range data^b after separation and fitted the DSIF-determined spacecraft orbit (corrected for *Agena* separation velocity). The residuals are shown in Fig. 34. Trial and error methods were used to obtain estimates of the ship speed, heading and range bias. Values obtained were:

Ship speed $3.5 \pm 0.5 \, \mathrm{yd/sec}$ Ship azimuth east of north $110 \pm 10 \, \mathrm{deg}$ Range bias $+700 \pm 100 \, \mathrm{yd}$

B. Flight Path Determination Using DSIF Tracking of the Spacecraft Transponder

1. Summary of Data Taken

The complete sequence of tracking events and ground tracking modes is described in Section III. The estimation method used is discussed in Section VI. Angle tracking data were used whenever the DSIF stations were in GM-1 or -3 and the "data condition" code indicated good data. Doppler was used only when the stations were in GM-1.

Table 11 provides a gross picture of the performance of the data handling system; column 3 gives the total number of data points received at each station during the life of the spacecraft transponder. The editing of the data, described in Section VI, allowed the number of points (and percentage of total) listed in column 4 to be used in the final orbit determination. Of particular interest is the number and percentage of data sets rejected for bad format or as "blunder points". No attempt is made to unscramble data messages containing format errors. "Blunder points" can create significant problems in converging on an orbit when very little data are available. Hence, they are important in influencing the time required to establish our first estimate of the orbit. The number and percentage of the points omitted because of "bad data condition" are listed in column 7. When the tracking station operators or automatic detectors recognized that the data being transmitted would not be usable, the data condition codeword reflected these situations. This situation occurs when retuning the ground

^bAngles from the ship were not used because of systematic orientation errors.

Dele	.	Points received	Points used	Bad format rejection	Blunder points	Bad data condition	Rejection limits on blunder points
DSIF station	Data types	% of received	% of received	% of received	% of received	% of received	
•		12	12 ^s	0	0	0	_
1	2-way doppler	100	100	0	0	0	l cps
4	0	301	214	5	1	81	
4	2-way doppler	100	71	1.7	0.3	27.0	0.15 cps
5	2-way doppler	102	82	4	0	16	0.15 cps
э	and hour angle declination	100	80	4.0	0	16.0	0.15 deg

Table 11. Summary of data used in orbit determination

transmitter to maximize the signal received at the spacecraft when commands are being sent, and during the acquisition phase.

DSIF 1 data actually occurred prior to separation of the spacecraft and the *Agena*. Therefore, its data were used only for *a priori* knowledge of position and velocity and were not used in the final orbit computation of residuals where only DSIF 4 and 5 data were used.

2. Weighting of the Data

The data weights were assigned in accordance with the policy described in Section VI. The weighting assigned to the data depends upon the sampling interval, elevation angle, counting time and range to the spacecraft. During the flight the effective noise due to variation of the transmitter reference frequency was calculated by regular recording of the transmitter frequency. The noise in the doppler due to this variation never became a dominant factor because the oscillator performance exceeded specifications and because transponder tracking ended prematurely. Table 12 summarizes the sampling, counting intervals and weighting used.

3. Discussion of Residuals

Once the data points and weights are fixed, the set of initial conditions which minimizes the weighted sum of

Table 12. Summary of weights, sample and count times

		*E to E + 7 ^h 30 ^m						
DSIF station	Data type	Sample spacing, sec	Count ^e time, sec	Weight, cps ^b or deg				
1	2-way doppler	5	continuous	1.1-2.2 ^d				
4	2-way doppler	60	50	0.184				
5	2-way doppler	60	50	0.188				
5	Hour angle, declination	60	_	0.185				

^{*} E is reference epoch used for orbit determination (Table 5).

the residuals squared is found by an iterative method. The differences between the vector of all observations and the calculated values based on the converged solution is called the vector of residuals. Figures 20 through 30 are the residual plots, by DSIF station, vs. time for the data types used in the final orbit. The detailed analysis of the residuals is published in Ref. 4 and 5. No particular characteristics display themselves as in Ranger 4 (tumbling), except for the bad angles at DSIF 4 which were not used because of a servo tracking problem and because of a discontinuity in hour angle at DSIF 5, due

^{*} There is one more data point here than listed in ODP printout, because two adjacent continuous count data points are differenced to obtain a data point for the ODP.

^b 1 cps = $\frac{c}{2i}$ = 0.156 m/sec where the transmitting frequency, *i*, is 960 × 10⁸ cps.

c Stations time tag the data at the end of the counting interval.

⁴ These weights are high due to the refraction uncertainties, since these data are taken close to the horizon.

to the refraction correction discontinuity $^{\circ}$ (Fig. 28 and 29 at edge of frame).

Table 13. Tracking noise statistics

DSIF station	Data type	No. of points	RMS	Mean
1	2-way doppler, cps	11	0.269	-0.006
4	2-way doppler, cps	213	0.022	-0.0001
5	2-way doppler, cps	81	0.033	-0.002
5	hour angle, deg	81	0.008	-0.003°
5	declination, deg	81	0.023	+0.021*

4. Statistics of Data and Orbital Estimates

a. Tracking data statistics. The root-mean-squared (RMS) noise and mean of the residuals for each station is given in Table 13 for each data type used. Note that the RMS noise and weights of Table 12 differ significantly in most cases. The difference in angle weighting is due to the presence of low-frequency mechanical deflection of the DSIF antennas. As for the doppler data, the point has now been reached where any small unknown correlation or bias could give erroneous results. Therefore, conservative weighting is applied until conclusive results show that there exists no other error source to be considered in the doppler weighting.

b. Statistics of orbit estimate. The accuracy of the orbit obtained depends on the statistics of the tracking noise and on the statistics of all error sources which influence the orbital estimate. The tracking noise statistics

Table 14. Statistics of knowledge of injection conditions including physical constant uncertainties

Stan	dard deviation				Correlation	coefficients			
		х	Y	z	×	Ÿ	ż	GМ⊕	GM∢
					Space fixed cart	esian coordina	tes		
X	0.1215 km	1	0.164	0.224	0.233	0.221	0.372	0.438	-0.0046
Y	0.6160 km	s	1	0.950	0.829	0.960	0.796	0.636	-0.013
Z	1.1887 km	y m		1	0.959	0.995	0.937	0.610	-0.013
z x ÿ	1.2191 m/sec	m			1	0.939	0.977	0.571	-0.011
Ÿ	0.6814 m/sec	•,				1	0,928	0.597	-0.012
ż	1.8096 m/sec	r		į			1	0.520	-0.012
GM⊕	3.8332 km²/sec³	c						1	-0.002
GM €	4.9997 km²/sec²	١				1			1
		R		φ	λ	v		γ	σ
			_ 		Earth fixed sphe	erical coordina	·os		
R	0.6594 km	1	-(0.970	- 0.958	-0.997	_	0.922	0.918
φ	0.008 <i>5</i> 70 deg	s		1	0.944	0.967		0.860	-0.978
λ	0.005574 deg	m			1	0.966		0.946	-0.864
v	0.5603 m/sec	1				1		0.941	-0.907
γ	0.003295 deg	c						1	- 0.753
σ	0.01303 deg								1

^cRefraction correction is handled by two polynomials, one for above 17 deg elevation and one for below. There is a slight discontinuity between the two at 17 deg.

are represented by the "equivalent or worse" white noise method described in Section VI. The Ranger 5 ODP does "solve for" and can directly include the effects of deviations in physical constants such as GM-Earth (GM_{\oplus}) , GM-Moon (GM_{\circlearrowleft}) , and station locations. Table 14 gives the covariance matrix describing the uncertainty in the space-fixed cartesian and also Earth-fixed spherical coordinates at the reference epoch, $E_2{}^d$, where the physical constants are in the estimator. (Note Table 15 for an a priori on physical constants.) The covariance matrix is given in terms of its correlation matrix and standard deviations of the coordinates. Table 16 shows the effect

Table 15. Standard deviations of estimated parameters

Parameters	Without physical constants	With physical constants	A priori 1-sigma
X km	0.096	0.121	
Ykm	0.368	0.616	
Z km	0.657	1.189	None
DX km/sec	0.726 × 10 ⁻¹	1.219 × 10 ⁻³	
DY km/sec	0.372 × 10 ⁻³	0.681 × 10 ⁻³	
DZ km/sec	0.932 × 10 ⁻²	1.810 × 10 ⁻⁸	
GM⊕ km²/sec²		3.833	4.0
GM _€ km³/sec²		5.00	5.0
DSIF 4			
radius, km		0.0599	0.060
latitude, deg		0.00099	0.001
longitude, deg		0.00098	0.001
DSIF 5			
radius, km		0.0598	0.060
latitude, deg		0.00099	0.001
longitude, deg	:	0.00098	0.001
R km	0.4281	0.6594	
φ deg	0.0045	0.0086	
λdeg	0.0032	0.0056	None
V km/sec	0.3515 × 10 ⁻⁸	0.5603 × 10 ⁻⁸	
γ deg	0.0021	0.0033	
σ deg	0.0068	0.0130	

when uncertainties in physical constants are not included in the solution parameters.

Although the estimation of these physical constants from Ranger 5 tracking data has not contributed much to the improvement in their values, the actual estimator of target conditions is statistically better than that which would be obtained by ignoring the fact that these constants are uncertain. At the same time the better estimator has the property of giving a realistic description of its accuracy. It is important that in-flight actions be based on realistic error estimates.

Table 17 summarizes the difference in the solution vectors of estimating with and without physical constants. The differences are well within the standard deviations for either case as can be seen in Table 15. Also from Table 15 the relative strength of the data in furnishing new information on physical constants can be observed.

The covariance of errors in knowledge of the coordinates at E_2 may be "mapped" to the target region using the miss parameters **B** (Appendix A) and T_L , the linearized time-of-flight, as measures of target error. T_L may be considered to represent the flight time to a vertical impact. (The influence of **B** on the parameter T_L is thus removed.) Table 18 represents the standard deviations and correlation matrix in the **B**•**T**, **B**•**R**, T_L system for both estimating the physical constants and for ignoring them.

To present a better feeling for the effect of the various physical constants on the miss parameter statistics, Fig. 35 shows the dispersion ellipses plotted in the B plane (a priori shown in Table 15). $GM_{\mathfrak{Q}}$ is not included since its effect is virtually eliminated due to the coordinate system (i.e. $\frac{\partial \mathbf{B}}{\partial GM_{\mathfrak{Q}}} \approx 0$); however, it does show itself in the third dimension T_L , which is not plotted. Obviously, the major contributor to the error is $GM_{\mathfrak{Q}}$. Therefore, if we are to obtain a higher degree of accuracy for future missions our estimates must be improved.

c. GM_{\oplus} solution: no a priori. In the Ranger 4 Report (Ref. 1) a solution for GM_{\oplus} was obtained by perturbations and a quadratic curve fitting of the weighted sum of the residuals squared. These same Ranger 4 data and

^dBetween E_1 and E_2 the spacecraft was tracked by DSIF 1 and there was an accumulation of these data into a covariance matrix at E_1 . This covariance was then mapped to the separation epoch, corrupted by the separation and used as *a priori* for the final data orbit.

eThe physical constants described in Ref. 6 were used in the trajectory calculation.

Table 16. Statistics of knowledge of injection conditions ignoring physical constant uncertainties

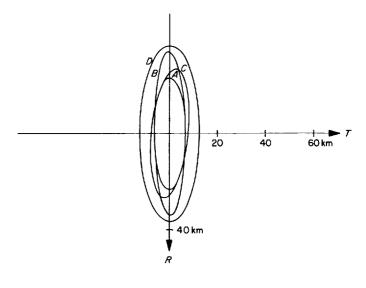
Stand	ard deviation			Correlation	coefficients		
		х	Υ	Z	х	Ý	ż
				Space fixed co	artesian coordinate	18	
x	0.0964 km	1	-0.544	-0.566	-0.510	-0.577	-0.314
Y	0.3678 km	S	1	0.890	0.613	0.928	0.560
Z	0.6571 km	m m		1	0.899	0.989	0.85
x	0.7261 m/sec	t r			1	0.846	0.966
Ÿ	0.3718 m/sec	i c a				1	0.814
ź	0.9322 m/sec	1					1
		R	φ	λ	V	γ	σ
				Earth fixed s	pherical coordinate	98	
R	0.4281 km	1	-0.961	-0.909	-0.999	-0.934	0.852
φ	0.004498 deg	S	1	0.860	0.952	0.846	-0.946
λ	0.003145 deg	m m		1	0.916	0.952	0.67
V	0.3515 m/sec	r l			1	0.942	-0.83
γ	0.002107 deg	c o				1	- 0.664
σ	0.006784 deg	1					1

also the Ranger 3 and Ranger 5 data have been processed through the new Orbit Determination Program (ODP) and the results are displayed in Table 19. Figure 36 summarizes the tracking coverage for each mission. Note that the old results on Ranger 4 compare closely with these and that all the data tell approximately the same story. Although the estimates show a consistently lower value for GM_{\oplus} , they still have large standard deviations and are still within the nominal

1-sigma ($\pm 4.0~{\rm km^3/sec^2}$) uncertainty. These statistics are, however, a direct function of our weighting scheme and since a conservative weight is assumed, possibly these standard deviations are conservative. The results are presented to show the consistency with the nominally accepted value and its uncertainty rather than to show any new value for GM_{\oplus} . When there is more assurance that other sources of error are absent in the data, possibly then a new value for GM_{\oplus} may be derived.

Table 17. Comparison of parameter values

Parameters	Without physical constants		With physical constants	ΔQ (with— without)
X km	5807.2013		5807.2046	+0.0033
Y km	- 1871.2780		— 1871.2696	+0.0084
Z km	-2891.8163		-2891.7849	+0.0314
DX km/sec	3.62544	21	3.6254794	+0.0373 × 10 ⁻²
DY km/sec	9.76461	43	9.7646336	+0.0196 × 10 ⁻³
DZ km/sec	-2.8588116		-2.8587352	+0.0764 × 10 ⁻⁸
GM _⊕ km²/sec²	398603.20	\	398600.05	-3
GM ₍ km³/sec²	4900.7589	N	4900.7604	+0.0015
DSIF 4		m i		
radius, km	6372.6076	n	6372.6097	0.0021
latitude, deg	-31.2123	١ĭ	-31.2123	0.0000
longitude, deg	136.8862	٧	136.8862	0.0000
DSIF 5	_	0 - u	_	
radius, km	6375.4947	•	6375.4971	0.0024
latitude, deg	- 25.7348	3	- 25.7348	0.0000
longitude, deg	27.6848		27.6848	0.0000



ESTIMATED PARAMETERS

A=X,Y,Z,DX,DY,DZ, $B=X,Y,Z,DX,DY,DZ,GM_{\bigoplus}$ C=X,Y,Z,DX,DY,DZ,DSIF STA LOCATION $D=X,Y,Z,DX,DY,DZ,GM_{\bigoplus},DSIF$ STA LOCATION

Fig. 35. Dispersion ellipses in the B plane for uncertainties in various estimated parameters

Table 18. Comparison of covariance matrices at encounter

Danamatan	Win	nout physical constants		With physical constants			
Parameter	Standard deviation	Correlation coefficients		Standard deviation	Correlation coef	ficients	
B∙R	23.64 km	1 -0.030	0.623	36.75 km	1 -0.005	0.563	
в•т	6.34 km	1	0.725	12.63 km	1	0.178	
T _L	4.29 sec	Symmetrical	1	7.19 sec	Symmetrical	1	

Table 19. Estimates of GM_{\oplus} using real tracking data

item No.	Data from	GM⊕ " estimate, km³/sec³	Standard deviation on estimate, km²/sec²	Remarks	Item No.	Data from	GM ⊕ * estimate, km²/sec²	Standard deviation on estimate, km³/sec²	Remarks
1	Ranger 3	398600,49	± 4.09	^b Other parameters estimated in this same run are X, Y,	2	Ranger 4	398601.87 (398600.27)	± 13.3 (± 12.8)	Same as item No. 1 ^c (Results in Ref. 1)
				Z, DX, DY, DZ, J, H, D and station locations	3	Ranger 5	398599.2	±13.2	Same as item No. 1

^{*} Nominal GM $_{\bigoplus}$ = 398603.2 \pm 4.0.

b Although there was no a priori on the parameters X, Y, Z, DX, DY, DZ, GM⊕ in these solutions, there was a priori on J of 0.3 × 10⁻⁴, H of 0.25 × 10⁻⁴, D of 0.875 × 10⁻⁶, station radius of 60 m, latitude of 0.001 deg and longitude of 0.001 deg.

Coccultation was not used as data but will be reported in the next technical report on Ranger.

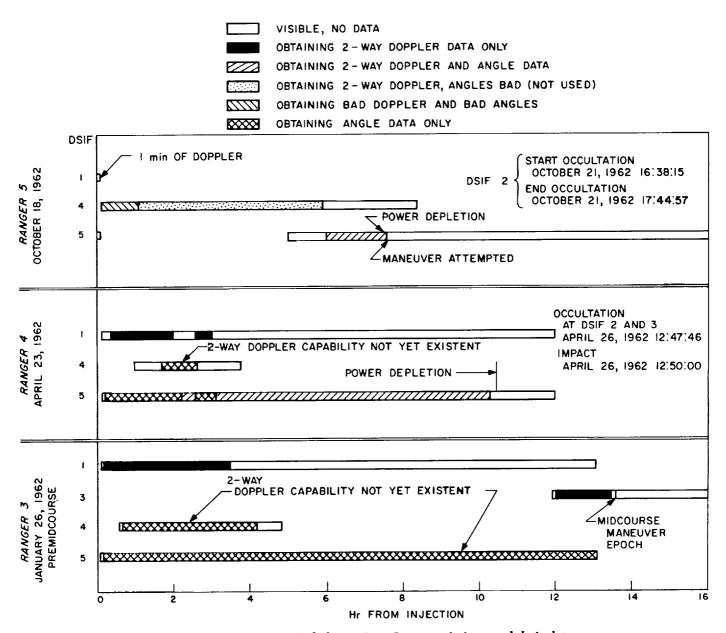


Fig. 36. Tracking station view periods for various Ranger missions and their data coverage

V. MIDCOURSE MANEUVER DETERMINATION USING DSIF CAPSULE BEACON TRACKING

A. Introduction

As soon as it was known that the batteries were being depleted on board the spacecraft and that soon there would be no power available to command a maneuver, commands were sent to place the probe on an impact trajectory. What response the spacecraft made to these commands was partially recorded by intermittent telemetry and the tracking data received from the capsule beacon. The spacecraft operated intermittently after battery failure, when solar power was high. Telemetry received some 10 hr after the command maneuver epoch indicated that the midcourse motor fuel was still intact and that the motor had not burned. It also showed that the attitude control gas, which was almost all available at the commanded maneuver time, was entirely consumed.

In the real time orbit determination it was assumed there was no maneuver and a predicted occultation was made on the early transponder data. There was a discrepancy of 8 min between the calculated and the observed time (calculated was 8 min early). Due to this inconsistency, a systematic study was made to determine whether there actually was some velocity increment added or whether the orbit determination program was wrong and no velocity increment had been added. Since the attitude control gas was completely expended and a known force could have been added, due to pitch jets not being coupled, it is possible that a small velocity increment occurred along the probe–Sun line.

B. Studies of Available Data

Although no single area investigated offered conclusive proof that a maneuver occurred, all study conclusions proposed that it did.

The studies included

- 1. Another independent program to check calculated occultation
- 2. A station compatibility test
- 3. Searches on occultation and beacon doppler peak for velocity increment
- 4. Comparing beacon doppler data (shape fitting) with varying models
- 5. Variation in our Earth-Moon model

- 6. Consistencies with Ranger 3 and Ranger 4 results
- 7. Angular data

In the actual operation when occultation time was first predicted it was computed for a hypothetical DSIF station located at the center of the Earth, but now a more elaborate program with actual station locations has confirmed the previous results. The agreement was within 30 sec and still indicated the same 8 min discrepancy.

A study of the various combinations of tracking data from each DSIF station showed that the 8 hr of transponder data were compatible and gave consistent results. Table 20 displays the effects on the impact parameters. The effects are small. Although DSIF 5 by itself tends to be somewhat different, especially in $\mathbf{B} \cdot \mathbf{R}$, the differences are not significant, considering the error statistics. Occultation time is rather insensitive to changes in $\mathbf{B} \cdot \mathbf{R}$ in this range, and the change in $\mathbf{B} \cdot \mathbf{T}$ is in the wrong direction (i.e., correct occultation $\mathbf{B} \cdot \mathbf{T} = 5195$).

Using the JPL trajectory program (Ref. 7), a search on the three velocity components (keeping position

Table 20. Station compatibilitya

Data from DSIF B•T, B•R, Altitude, closest

Data from DSIF ^b stations	B•T, km	B·R, km	Altitude, km	Time of closest approach	Remarks
1, 4, 5	5051	550	626	15:53:06	
1, 5	5064	574	637	15:53:06	
1, 4	5064	535	634	15:53:06	
4, 5	5061	473	628	15:52:52	
5	5001	72	565	15:53:29	Std. devia- tion ±1341 km on B
Searched in by occultation and doppler peak	5195	542	736	15:53:43	

These results were obtained using a slightly different radius of Earth (REM = 6378.1650) scale factor on the lunar ephemeris than was used in the final orbit displayed in Appendix D.

DSIF 1 and 4 by themselves had insufficient data to converge on an orbit. (Their angular data were not used.)

fixed) to satisfy the three times, start occultation, end occultation, and peak, in the beacon doppler data, was completed. The search was made at both the injection epoch and at the midcourse epoch. Comparing the results obtained with the results of the 8 hr of tracking data it can be seen that large changes are required at the injection epoch whereas at the midcourse epoch only a slight change will satisfy the end times (Table 21).

Table 21. Differences of searches at injection epoch and midcourse epoch with data orbit^a

Parameters	Search at injection minus data orbit	Search at midcourse minus data orbit
DX	-6.63 m/sec	-0.64 m/sec
DY	+0.0047 m/sec	-0.24 m/sec
DZ	-9.33 m/sec	-0.067 m/sec

Table 15, which lists the standard deviations on the velocity components at injection, shows that better than a 5-sigma change would be required. If the errors are mapped to midcourse epoch, the change required at that epoch is still beyond the 1-sigma, as at injection (\approx 3 sigma); however, the magnitude of the departure is smaller and can be physically explained. Another discrediting fact about the injection epoch search is that when these conditions are used to pass a trajectory through the 8 hr of initial tracking, the residuals (actual observation—calculated observation) are greatly biased. This points, then, to an approximate 0.7 m/sec velocity increment at midcourse epoch in a direction 5 deg off the probe—Sun line and directed away from the Sun.

Next, the beacon doppler data near encounter were analyzed. Since there are some unknown constants in the calculations, such as the beacon frequency, the beacon frequency drift, and the reference oscillator bias at the receiving stations, an attempt to just match the shape of the doppler curve was made. Figure 37 shows the various calculated doppler compared to the observed doppler. Notice that the occultation searched in conditions match the observed data best. By putting in various drift rates on the beacon frequency, at times when the probe was in and out of the Moon's shadow, a slightly better fit is obtained (Fig. 37 curve No. 4).

A study of the variations of the physical constants in the model again cannot compensate for the 8 min variation in occultation. Figure 38 shows a cross plot for GM_{\oplus} , $GM_{\mathfrak{q}}$, and $REM^{\mathfrak{l}}$ (the Earth-Moon distance scaling factor holding them in correct proportion) against occultation times and closest approach time. It is impossible to satisfy all the times by perturbing the physical constants. Even if the first occultation time were satisfied the change would be larger than 7- or 8-sigma on REM. The effect of uncertainty in ΔT [Universal Time (GMT) to Ephemeris Time] was negligible.

As further proof of the validity of the model, results from Ranger 3 and Ranger 4 missions were studied. Ranger 3 flew by the Moon with a closest approach of some 35,000 km, and good tracking data were obtained before and after encounter. These data were very useful in the solution for $GM_{\mathfrak{q}}$ and gave approximately the same value as the JPL nominal. Ranger 4, which was occulted by the Moon and definitely had no maneuver, had an orbit based upon about the same amount of tracking data as Ranger 5. The calculated occultation was off only 24 sec from the observed occultation and within 1-sigma uncertainty in the time of flight. This suggests that the model and trajectory program are correct.

There were some good angular data taken by DSIF 5 prior to and after occultation. Residuals shown in Fig. 39 are plotted from injection conditions at midcourse epoch and are based on no maneuver, the commanded maneuver, and the slight maneuver determined by occultation. The residuals definitely show that the commanded maneuver was not executed. They also show that the no-maneuver conditions deviate quite noticeably. However, the conditions determined by the occultation time, the reacquisition time, and the doppler peak time fit the real data fairly well, again supporting the belief that a small maneuver occurred.

C. Conclusions

All results seem to indicate a maneuver. Our best estimate of the magnitude and direction of this maneuver is a 0.69 m/sec velocity increment 5 deg off the probe—Sun line directed away from the Sun. The end conditions are displayed in Section II and in Appendix C of this Report. Although this small maneuver has an assumed epoch about the time of the commanded maneuver, there is a chance that the maneuver might have taken place several hours later when solar power was available. Possibly the maneuver took place slowly by means of the attitude control gas which was almost entirely available after the

Does not affect the Earth's potential function.

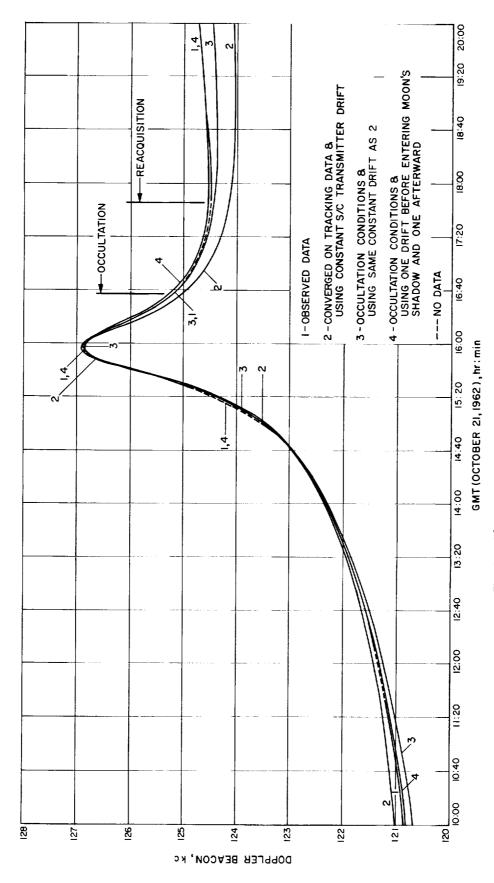


Fig. 37. Observed beacon data vs. calculated data

commanded, midcourse epoch (some 3+ pounds) and was depleted 10 hr later, as verified by telemetry recordings. Since there is not a perfect couple on the pitch jets,

their contribution might have caused a velocity increment along the Sun line, as was indeed the case suggested by the occultation conditions.

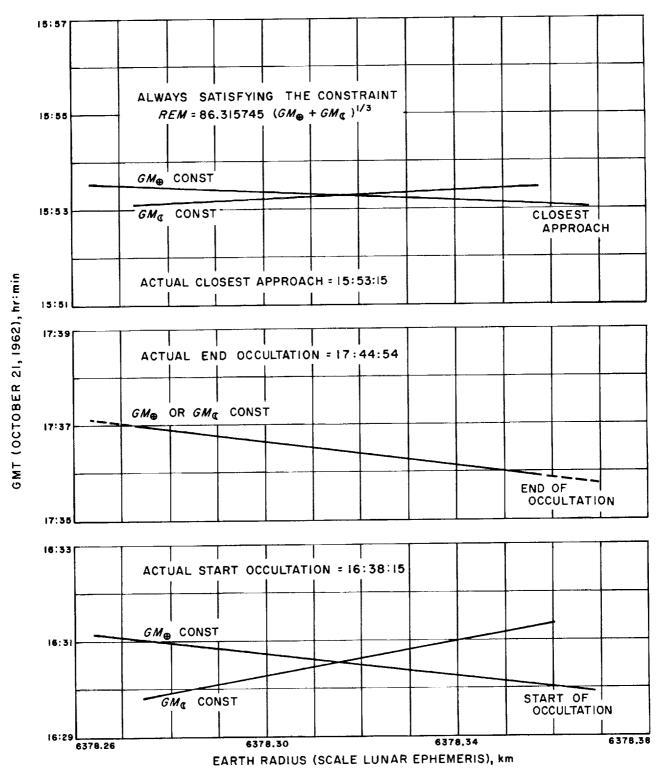
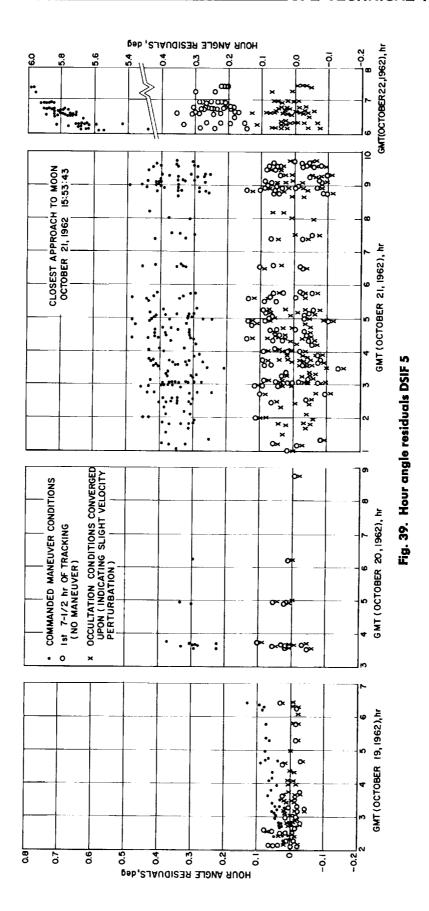


Fig. 38. Physical model study



VI. FLIGHT PATH ANALYSIS OPERATIONS AND POLICIES

A. Introduction

The Flight Path Analysis group is the part of the spaceflight operations team which performs the real-time radio guidance calculations as well as the postflight determination of the spacecraft orbit. Its functions are depicted in Fig. 40.

It should be noted that the functions are sometimes carried on simultaneously in a single digital computer program.

B. Operational Description

1. Data Editing, Analysis, and Evaluation

Editing, analysis, and evaluation of the tracking data are accomplished in several ways.

a. Teletype (TTY) printed display of incoming data is visually scanned in real time to detect any systematic errors.

- b. Station reports, both printed and verbal, are analyzed to detect any abnormalities. In addition, critical information on oscillator drift statistics, frequency changes and changes in transmitter assignment are evaluated.
- c. Newly received TTY data are periodically entered, in batches, into a large digital computer program called the tracking data editing program (TDEP). The TDEP checks the format, data condition code, data range, station and time sequence against the input master format and control cards. All data are listed along with the reason for rejection of any data point. The new data which have not been rejected are added to the TDEP's master data tape, which contains all accepted data.
- d. Once the orbit is reasonably well known, the deviations of the values of new observations from their predicted values (the residuals) are tested to determine whether they are within selected rejection limits. In this way "blunder points" are easily

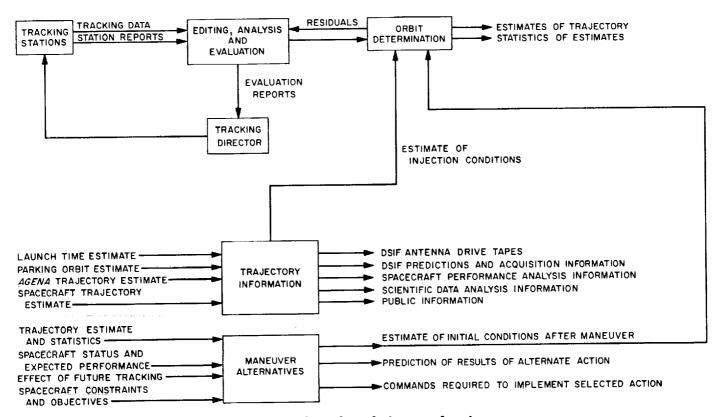


Fig. 40. Flight path analysis group functions

- detected before they influence the estimate of the orbit.
- e. The residuals and rejected data points are analyzed to determine the validity of the noise models and to locate any systematic error source. On the basis of the information gained from the evaluation of the incoming station reports and tracking data corrective action is recommended to the tracking director.

2. Orbit Determination

The tracking data placed on the TDEP's master data tape are the basis for forming an ODP data tape. Control of the information placed on the ODP data tape is exercised through input to the TDEP. The ODP and TDEP are linked in such a way that the ODP can call the TDEP to add new data to the ODP tape. The most important ODP inputs are the edited tracking data, the data weights, and rejection limits.

During the flight, new data points are continually being added to the ODP tape; weights are revised, and residuals from selected converged orbits are plotted and printed. The converged ODP output provides an estimate of the initial conditions and physical constants (parameters, in general) describing the flight path as well as a statistical description of the uncertainties in the parameter vector. The estimated covariance of the parameter vector is then mapped to other regions useful for interpretation of results. Typically, the properties of the "error ellipse" in the target plane (*B*-plane) are computed as well as other quantities useful in considering maneuver alternatives.

3. Trajectory Information

At all times information on a typical mission trajectory is essential to analysis of spacecraft performance and scientific data, supplying tracking station acquisition data, antenna pointing data, and general information. The basis for forming these trajectory estimates varies with the amount of information available and, thus, requires continual updating.

4. Maneuver Alternatives

As suggested in Fig. 40, the trajectory estimate(s), information on expected spacecraft performance and correct status, statistics of current and future knowledge of

the flight path, spacecraft restraints, and mission objectives dependent on the flight path, are input into a digital computer program which is designed to examine the detailed results of following the available alternative maneuvers (trajectory) corrections. Commands necessary to implement any of the various alternate maneuvers are also computed and checked.

C. In-flight Policies

The JPL ODP is designed to find the set of initial conditions at injection epoch which causes the weighted sum of squares of the residuals (observed minus computed) to be minimized. The method is named modified-leastsquares (MLS) to call attention to the method used in obtaining the weights. In the usual least squares method, the individual data points are weighted inversely proportional to their expected (or measured) variances to form the weighted sum of the squared residuals. In MLS, the independent weighting values are determined by the expected (or measured) effective variances.8 In arriving at the effective variance for each data type at each DSIF station (vs. time) consideration is given to the effective correlation width of all recognized error sources, sampling rates, range to the spacecraft, counting time, and elevation angle. The ODP-calculated covariance matrix of injection errors will always give a conservative estimate of the accuracy when effective variances, "equivalentor-worse uncorrelated noise," are used. In editing the data, the policy is that it is better to reject a data set with questionable format than to attempt the real-time correction of the error. An analogous policy is used in weighting the data; there is a maximum weight which can be assigned to any data point independent of whether it appears that the data may be dramatically better in a particular time interval or not. By sacrificing the possibility of extracting the maximum possible information during the flight the sensitivity to "blunder points" or small "hidden" errors, whose effect may be very significant, is reduced.

This approach was first used at JPL by A. R. M. Noton in August, 1959 in an Internal Memorandum Effect of Correlated Data in Orbit Determination from Radio Tracking Data. Further discussion was given by A. R. M. Noton, E. Cutting, F. Barnes (Ref. 8). T. A. Magness and J. B. McGuire of Space Technology Laboratories, Inc., have developed mathematical expressions to contrast the performance of LS, MLS, and minimum covariance estimators (under JPL Contract No. 950045) in terms of the eigenvalues and eigen-vectors of the data noise covariance matrix (Ref. 9).

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APPENDIX A

Definition of the miss parameter B

The miss parameter **B** is used at JPL to measure miss distances for lunar and interplanetary trajectories and is described by W. Kizner in Ref. 10. **B** has the desirable feature of being very nearly a linear function of changes in injection conditions.

The osculating conic at closest approach to the target body is used in defining \mathbf{B} . \mathbf{B} is the vector from the target's center of mass perpendicular to the incoming asymptote. Let \mathbf{S}_I be a unit vector in the direction of the incoming asymptote. The orientation of \mathbf{B} in the plane

normal to S_I is described in terms of two unit vectors R and T, normal to S_I . T is taken parallel to a fixed reference plane and R completes a right-handed orthogonal system. Figure A-1 illustrates the situation.

The Ranger 5 work has used the orbital plane of the Moon as the reference plane. If W is a unit vector normal to the orbital plane (W in direction of $\mathbf{R}_{M} \times \mathbf{V}_{M}$, where \mathbf{R}_{M} is radius vector to Moon from Earth and \mathbf{V}_{M} is the space-fixed velocity of the Moon relative to the Earth's center) then $\mathbf{T} = \mathbf{S}_{I} \times \mathbf{W}$ defines our coordinate system.

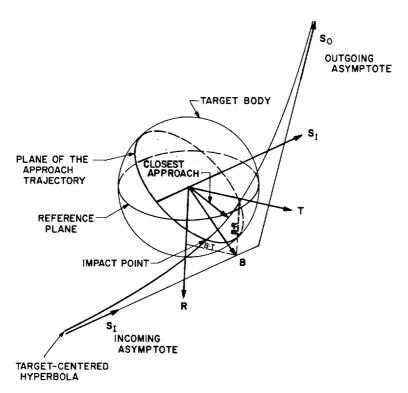


Fig. A-1. Definition of B . T, B . R system

APPENDIX B

Ranger 5 premidcourse orbit

E		EQUATORIAL COORDINATES
SPACE TRAJECTORIES	RA-5 PREMIDCOURSE ORBIT	
CASE 1		GEOCENTR IC

æ		. COORDINATES		.58695064-01 .58695064-01 .10287071 01 .15500289 03			.30333478 02 .37522495 02 .13812271 00 .71517070 02 .37651302 02 .14220968 00	18 47 23.000 . COORDINATES	.83518024 00 .62148529 02 .17447221 02 .56910842-01 .56910842-01 .10283263 01 .14777928 03 .20838614 02	COORDINATES	-10793421 01 -91762923 02 -88977813-03 -74546336-01 -30329368 02 -26503374 02 -26503374 02 -13865293 00 -71275275 02 -71275275 02 -7	19 07 23.000
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CASE 1

RA-5 PREMIDCOURSE ORBIT

EQUATORIAL COORDINATES	.22195830 01 DZ .79120096 00 .31813802 02 AZE .27547114 03 .24658494 02 DZS10691524 02 .93907969-01 DZM .43249133-01 .938213110 06 VT .10254285 01 .98668862 02 LDM .92392306 02 .97053760 01 DEM .20892327 02	ECLIPTIC COORDINATES	29227709 02 D215803671 00 20573204 01 AZ90276080 02 26876569 02 DZE90944766-03 26807618 02 DZI .76129436-01 14912010 09 VST .30297722 02 47293219 01 MEP .19985799 02 72903836 02 ESM .14040588 00 11234211 00 STP .69172694 02 31237824 06 SPN .84286892 02 76348200 02 DI .16691562 00 59656564-01 D3 .35328436-03	OCT. 19,1962 00 37 23.000	EQUATORIAL COORDINATES	18619557 01 DZ .73536865 00 73263950 02 AZ .63066638 02 22360610 02 AZE .27321193 03 24641956 02 DZS10684343 02 11229190 00 DZM .36137603-01 11229190 00 DZT .36137603-01 38265103 06 VT .10239331 01 99849102 02 LOM .63490368 02 97356234 01 DEM .20908413 02	ECLIPTIC COORDINATES	28859360 02 D266994428-01 20575705 01 AZ .90118590 02 26858540 02 DZE91898440-03 26769893 02 DZI .76909065-01 14911034 09 VSI .30281122 02 64787349 01 MEP .20642167 02 73918396 02 ESM .14144723 00 10584016 00 STP .68234360 02 29590945 06 SPN .88498637 02 76660407 02 DI .17620559 00 65696188-01 D3 .41322949-03
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APPENDIX C

Ranger 5 postmidcourse orbit

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		C1 DY .150776 C1 PTH .749C16 C1 PTE .1465139 C2 DYS246139 C1 DYM143070 C1 DYT143070 C1 RT .383523 C3 RAM .101834		C2 DY .284756 C2 DYE .268279 C2 DYE .2670637 C2 BYI .149093 C3 EMP .937510 C3 EMS .756246 C3 TSP .964072 C6 RPI .272333 C2 SIN .770325	33 00	C1 DY .125675 C1 PTH .759721 C2 PTE .102953 C2 DYS245804 G1 DYM179068 C1 DYT179068 G1 RT .384548 O3 RAH .104174		C2 DY .281846 C2 PTH19375 C2 DYE .267915 C2 DYT .266312 C2 RST .149074 O3 EMP .126402 C3 TSP .873564 C6 RPT .248395 C2 SIN .773365 C2 D2 .877210
TRAJECTCRIES		5 DX15263943 2 V .22460394 3 VE .8573311 8 DXS .13201908 6 DXP10110773 6 DXT10110773 6 VP .1C214354 7 RAS .2C343972 1 SHA12249921		5 V14728305 2 V -32C59058 3 DXE13201908 5 DXT14321968 2 LOT25436738 2 EPH -14950143 2 SEM -1C423246 2 TPS -11305175 C RPM -27233362 3 CPT -77398211	ATE 2437956.833333	5 DX13652C79 9 V .19512458 9 VE .10592154 8 DXS .13279794 6 DXM10C26111 6 DXT10C26111 7 NM .1C1852617 8 RAS .2C359617		DX14645CC2 V .31762430 DXE13279794 DXI14282465 LOT .25603909 EPM .14643662 SEM .1C222062 TPS .11444780 C RPM .24839577 C RPM .24839577
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E11		Y .1C339344 06 LAT .1C890383 02 LAT .1C890383 02 YS584C6921 08 YM .35063342 06 YT .35063342 06 VS .25900337 02 ALT .11672199 06 DT .95999999 03		Y .63765859 C8 YE .63661747 C8 YT .64037912 C8 LOE .2529376 C2 ESP .463937C2-01 MSP .964C7213-C1 ETP .93751004 C1 STE .75624679 C2 GCT .28256752 C3 VEP .22460394 C1	IN. 52.000 SEC.	Y .12315872 C6 LAT .122761C5 02 LAT .122761C5 02 YS58761142 08 YM .34831339 C6 YT .34831339 C6 VS .259C2161 02 ALT .14583247 C6 DT .19200CCC C4		Y .641737C0 08 YE .64047832 08 YI .64421966 08 LOE .25459542 02 ESP .58516955-01 MSP .87356456-01 ETP .12640281 02 STE .77634928 02 GCT .282246G3 03 VEP .19512498 01
CASE 1 RA-5 POSTMIDOGURSE OR	CENTRIC	X62628797 C5 R .12309943 C6 XS13471413 C9 XM73473394 C5 XI73473394 C5 XI73473394 C5 DE .10962806 C2 DLT .34CCCCOO C2	HELIQCENTRIC	X .1346515C 09 R .14898695 09 XE .13471413 C9 XT .13464C65 C9 LTE .76042369-04 EPS .95613226 02 MPS .11305179 03 EPT .14950143 03 SET .10423246 03 GCE .11322539 03	C DAYS 6 HRS. 3 H GECCENTRIC	X83380713 C5 R .15220970 G6 XS13452345 09 XM87973703 G5 XT87973703 G5 XT14899222 C9 GED .12357202 02 DUT .34C00C00 02	HEL IOCENTRIC	X .13444006 C9 R .14897112 C9 XE .1345345 C9 XT .13443547 C9 LTE .70662C71-C4 EPS .9753924C 02 MPS .1144478C C3 EPT .14643662 C3 SET .110222062 C3 GCE .11127601 C3 REP .15220970 C6

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EQUATORIAL	DZ AZE DZS DZS VI LOM DEM	LIPTIC	DZ AZ DZE UZT VST WEP ESM STP STP D1	, 1962	EQUATORIAL	DZ AZE DZS DZM DZT VT LOM	LIPTIC	DZE DZE DZT VST WEP ESW STP SPN D1 C3
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	OREIT	Y VE VT VT VT VS PSP PSP ETP STE GCT	CEC LAT LNS SHA SVL	Z Z	CEC LAT VS VY VI VS DT		YELOE ESP MSP ETP STE
CASE 1	RA-5 POSTMIDCOURSE C Heliocentric	X .13155294 09 R .14880283 09 XE .13181432 09 XT .131547193-05 LTE51947193-05 EPS .10437833 03 MPS .939C6377 02 EPT .16C7C738 03 SET .75853598 02 GCE .10333187 03 REP .38883292 06	X .33660844 04 X .33660844 04 R .83552303 G4 R .83552293 04 LTS .15325553 01 ALT .66172303 04 HGE .25562166 C3	2 DAYS 13 HRS. 3 GECCENTRIC	X26273C23 C6 R .391C3371 G6 K .391C3371 G6 XS13178851 09 XM26612311 G6 XT26612311 G6 KS .14889902 09 GED .17723689 02 DUT .3400000C C2	HELIOCENTRIC	X .13152578 09 R .14880175 09 XE .13178851 C9 XT .1315239 C9 LTE61567391-05 EPS .10433059 03 MPS .82272548 C2 EPT .17005297 C3 SET .75619348 02 GCE .10328723 03

14		COORDINATES	.75442349 00 .80835820 02 .97558618 02	.17212052 02	15 30 00.000	CCORDINATES	6198	.24745224 01 .27050704 C3		-16910933 00 -16910933 00		.23292829 03 .17694437 02	CCORDINATES				29758590			1722441 02	4348	COORCINATES	92659552	92669034 02	•	.29862060 02	15 53 43.304
		EQUATORIAL	02 A2 A2R	ASD	1962	EQUATORIAL	70	AZE	•	•		C G M	ECLIPTIC	20	A2	•				N C		EQUATOR 1 AL	20	AZR		ASD	
		EQUAT	01002	-02	21,1962	EQUAT	CI			3 8		03	ECL	02	10	20	60	7 6	-18	4 0	02	EQUAT	15	9 0	1	-01	21,1962
			.14980237 55435359 55869289	- 92849468-0	OCT.	_	.11140224	.36322929	24094468	60730687	.39691130	.13562344		.27585172	13830587	.25637295	.14879852	10446690	.27453512	.34905655	.14668999	_	17213293	36501597		.26288505-0	0CT.
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IES			37C17C64- 16776770 .16690163	13815446 15284091	2437959.145833		11213875	.27401870	.14351403	75804888	.98592586	38130574		15472790	.31629287	15109451	.27906437	75385213	.54057864	.349C5655 .9C622533	6		36333867	.19810168	.66530451	11783986	2437959.16230676
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TRAJ			2000	C1 02	DATE							03			62	3 3	705	03	0	000	03					05	DATE
SPACE			26311443 .31025003 .43521964-	.10969114 .82205188	JULIAN		11950011	.23271453	27589181	.12063745	.39691130	.30367949		.37360000	.27905373	.43332500	.16685443	12594104	.20498850	. 14 / /06 / 1 . 24 19 58 C4	-10197132		-11373390	-28532108	87643877	.72315156 .53491C40	JULIAN
			RA LON LTE	AL P HNG			7 6	LON	25	17	E (DR OR		7	LON 7	77	LTT	SAP	TEP	SIP	CPE		7 8 8			AL P HNG	
			92 63	07			90	05	80	8	05	90		08	200	80	20			9 6	01		40	_	_	07	•
			4C078208 26613790 80876696	.58201207 75148C8C	52.C00 SEC.		.26614925	.17668828	63624671	.26727155	-29927284	.38734473 .59999999		.69640560	.14385497	-69642043	.27758558	-27453512-	-23834629	.27160051	1752809		11222996	10647152	.26220334	.28260365 93950958	35,304 SEC.
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	POSTMIDCOURSE	SELENDCENTRIC		.25566940	DAYS 13 HRS.	GEOCENTRIC	(26438701		513176269 1 - 24749638	- 1		34000000	HELIOCENTRIC		14880057		67339717-		1559601	- 10323279 - 10323279	.39372095	SELENOCENTRIC	31033774				DAYS 13 HRS.
CASE	RA-5	SELEI	X & & ZT	ALT	7	GEOCE	~ 6	c oc	××	X	S S	פבו	HEL IC	×	× 11	×	LTE	, a	EPI	SE	RES	SELE	~ 4	. uz.	Ţ	ALT	8

15		ES	000 000 000 000 000 000 000 000	TES	1 00 6 02 6 02 6 02 7 03 8 03	TES	5 01 6 02 4 02	TES	305 9 04 1 03 2 05 5 05 7 00 1 01	000
		CCORDINATE	2734 C484 3849 4526 2260 2260 7590 1897 5455	COORDINAT	.38789701 .89304496 .80364888- .85387706- .35256216 .14737516 .17168672	COORDINATE	.10284995 .62725377 .82230666	CCORDINATE	15 53 43.3 -24733479 -159493208- -12932261 -11791208 -90437032 -6195805 -6195805	00 00
		CCOR	.85827348 .752C4849 .27038492 10445260 17022601 17022601 .98575903 .22718974	COOR		CCO	.102		224- - 599- - 612- - 613- - 165- - 16	16
		EGUATORIAL	DZ AZE DZS DZM VT VT LCM DEN	ECL IPTIC	DZ D	EQUATORIAL	DZ AZ AZR ASD	QUATOR I AL	1962 RCA TFP MTA PZ RZ TF TZ MZ	1,1962
		ECUA.	000000000000000000000000000000000000000	Ë	000000000000000000000000000000000000000	EQUA	01-04	EQUA	000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000 000	2
			.10C16943 .72600316 .47587901 24C90856 60977963 .39698308 .13523156		-27518275 -27477825 -26557814 -25630645 -14879660 -8082276 -10465205 -98911702 -24733479	101477	.16114740 17C93266- 17C23913- .52356868-	8.R	0CT. 14526641 000C0000 .82813397 .54553661 .42339596 .16762723 -12510121 .71299607	001.
			DY PTH PTE DYS - DYM - NYT - RAM		DYE DYE DYE BEND CONTROL OF TABLE BEND CONTR		PTH PTH OP	T AND	APPE RAPI RAY TAG TAG	
			000 000 000 000 000 000		22222222	•	5515152	At B	26002242	99
			19612389 23635742 26960412 14358671 75558771 75558771 2657903 38405133		0210 6175 8971 8971 871 8971 8971 9173 9179		6512 1448 6800 9481 2137	TICN	2306 1789 4592 11389 5090 6750 4398 8294	. 16666666
					-16320210 -31996175 -14358971 -15114559 -27922728 -7520155 -83131379 -24733479	170	12056512 22601448 225368C0 6369481 67862137-	CNVENTION	959.1623067 -26441789 -63764592 -25261389 -82975690 -53386755- -16699720 -99214398	91.6
RIES							X > X N N N N N N N N N N N N N N N N N	ដ	W K K H X X D X X K	2437959
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ACE			8760 6708 25526 4649 9596 9596 4217		2500 22818 00000 00000 0012 2545 2545 2545 2545 37516		3704 3707 37005 5736 5736	J	JULIAN 3039960 3039960 5263333- 3261995 5505910 7730998 2674851	JULIAN
SP			.12068760 .13486708 .22682526 27604049 .12039596 .12039596 .39698308 .29774839		.41852500 .27922818 19250000 .44545000 .75590712 .17168872 .35256216 .14737516	•	.29163704 .33323707 .87497005 89945736 .1314434		JULIAN -28039960 -55901242 -85263333- -88261995 -45505910 -27730998 -42674851 -19714907	7
			LOS CR		LON 26 26 27 CTT 27 SEP SEP EST SIP	7	RA LON LTE ALP HNG		INC C1 C1 MZ QZ SZU SZU SZI BZ	
			005 005 005 005 005 005 005 005		08 00 00 00 00 00 00 00 00 00 00 00 00 0	5	001 003 003		000000000000000000000000000000000000000	
			5440 5640 5640 5640 560 560 560 560 560		9882 5563 5216 5216 7571 1702 5205 5205	71.0	3010 5065 1483 1256 1043 4390		0680 3997 3997 9718 8472 9580 6748 6558	SEC
			.2677547 .1771656 .1771656 .264054 .2664054 .2992746		.69679882 .16115503 .69386216 .6967852 .27776947 .14770671 .98911707 .10465205		.13493010 .67716665 77901483 .26200256 .35761043		.1578 .1145 .1237 .4404 .7129 .2658 .8972	52.C00
	OREIT		CEC YAN		LAT YE YE KSP ESP ESP ESP ESP ESP	À	LAT CHAT SHA		PASSAGE CCA	Z.
			000 000 000 000 000 000 000 000 000 00		03 03 03 03 03 03	8	4000 4000 6000 6000			•
	COUR		531 612 612 612 612 757 757 682 682		5256 5256 5256 5256 5256 5256 5356 5356	7101	2633 2475 3475 1172 1172 1173		PERICENTER 2786451 04 0702335 01 4150945-04 6418059 00 3343996 00 33846564 00 1313782 00 17992070 00	
	PCSTMIDCOURSE	RIC	26651531 .39659612 .39659612 13174226 26856757 .14889751 .17829682	NTRIC	13147574 14879505 1314726 13147369 17407020 10430140 98824472 75200155	REP .39639 LENDCENTRIC	.20522633 .24733479 .24733475 .15330172 .73534793	ELENOCENTRIC	OF PERICEN42786451 42786451 34150945 53343996 93846564 93846564 131313782 -11313782	DAYS 14 HRS
CASE 1	RA-5 P	GEOCENTR1C	GEN XXXXXX	HEL IOCENTRIC	COETTXXXX	SELENDO	TAL CETSRRX	SELENDO	S C X X Y Y Y Y Y Y Y Y Y Y Y Y Y Y Y Y Y	2 0

16		CCGRDINATES	80153038 C0 82835557 G2 27026204 G3 110444845 G2 117052109 G0 117052109 C0 98571396 C0 22567093 C3	CCCRCINATES	522	527	.15518011 03 .15518011 03 .10341813 03 .27716725 02	COORDINATES	.97205146 00 .66124608 02 .80099692 02 .42734557 02	30 00°000	COORCINATES	.46799388 CO .92695379 C2 .26983885 C3 .10442864 C2 .17192837 CO .98549956 CC .98549956 CC .21841331 C3
			DZ -80 AZ -82 AZE -27 DZS -10 DZM -117 VT -98 LCM -22 DEM -17	ECLIPTIC COO	DZ -41 AZ -89 DZE10		7777		D2 .97 A2 .66 AZR .80 ASD .42	16		DZ -46 AZ -926 DZS -100 DZM -17 DZT -17 VI -98 LCM -217
		EQUATORIAL	910 00 354 02 127 01 253 00 253 00 253 00 662 03 401 02	ECLI	144		512-18 447 04 896 02 627 02	EQUATORIAL	416 C1 186 02 499 02 897-C1	CCT. 21,1962	EQUATCRIAL	2669-C1 0865 02 5469 02 5328 C2 4339 00 4939 00 4953 03
			.80820 .66546 .48315 -24089 -61043 -39700 .13528		32593 26256		.27453 .25611 .59245 .25837		.14186416 C1 .11812186 O2 .11847499 O2 .48828897-C1	ö		. 48458 . 48486 . 38486 61355 . 13556
			PTH PTH DYS DYM DYM DYM DYM DES		PIH	REST	RPT SIN D2		PTH PTH PTH PTH			PTH PTE DYS DYM DYT RTT RAM
			87 C1 C9 C2 75 C2 91 CC 91 CC 94 CC		0000	000	69 02 47 04 45 03 10 02		38 C1 22 C1 01 C1 97 C1 28 C2	220		9 C C C C C C C C C C C C C C C C C C C
ES			2174438 2454350 2673260 1436037 7549349 7549349 7549349		-F 10 D (1	-27527C40 -75789184	.2481946 .2561144 .1019804		14195C38 22298922 22233401 66326797 45646837	2437959.18750000		. 23179320 . 23662309 . 26346951 . 14370545 75180751 75180751 75180751 75180751 75180751
TRAJECTORIES			S C C C C C C C C C C C C C C C C C C C		DXE V	LOT RPR	TPS CPT CPS		UX V V V V V V V V V V V V V V V V V V V	ш		O DO C C C C C C C C C C C C C C C C C C
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SPACE			.121C0142 .13491436 .22529867 .276C7983 .12033178 .12033178 .29617865		.43367500 .27927450 19500000		.36369451 .14770671 17915C84		.66964C41 .51C12538 .1C659354 90554647 .17270497	JUL IAN		.12211290 .13528469 .21814847 27626783 .12002358 .39709234 .28867781
			LON LON 2SS 2M 2TT COS DR		2 LON 2E 2E	SEP	TEP EST SIP CPE		RA LON LTE ALP HNG			RAA LON 25 27 27 10 08
			900000000000000000000000000000000000000				03		20025	•		000000000000000000000000000000000000000
			.26809706 .17724828 .17724830 .63668038 .26617558 .26617558 .29927506 .39107184		.69690213 .16698932- .69396106	.14820265	.99847038 .10470104 .11182300		.19214861 .15156792 48119527 .26194943 .10750844	52.000 SEC		.26878888 .17726166 .17726167 .63711397 .26507400 .26507400 .26507400
	ORBIT		CEC LAT VS VT VT VS OT		LAT YE	ESP MSP	STE STE GCT VEP		CEC LAT LNS SHA SVL	N N		CEC LAT YAS VT VT DT
			9699696		80000		0000		04 03 03 03	. 33		00 00 00 00 00 00 00 00 00 00 00 00 00
1	POSTMIDCOURSE	VTRIC	26729679 39744804 39744803 13173684 26885208 26885208 14889733 .17837989	HEL IOCENTRIC	.13146955 .14879841 .13173684	1		SELENOCENTRIC	.1552925 .25611447 .25611442 .15330522 .82314468	2 DAYS 14 HRS.	TRIC	27147336 -40106957 -40106956 13171C99 27020815 27020815 14889649 -17839333
CASE	RA-5	GEOCENTRIC	DEENAKK	HEL 10(× ~ ~ ×	EPS	SET SET GCE REP	SELENC	X R LTS ALT HGE	2 [GEOCENTRIC	DEGRAXXXXXXX

17		CCORDINATES	7 7 7	CCORDINATES	.63992225 00 .89232405 02 .80251984 02 .23011667 C2	17 00 00-000	CCORDINATES	.33328139 CC .94555171 C2 .26971878 C3 .1044C880 C2 .17333116 C0 .98528665 C0 .21115543 C3	COORDINATES	.35995233 GO .89333665 G2 .10255575-C2 .85269212-01 .29746717 G2 .18036143 GO .1472091C GO .8263G850 G2 .10413071 G3 .7789884C G1
		ECL IPTIC +		EQUATORIAL	02 A2 A2R A5C	362	EGUATORIAL	DZ AZE DZZ - DEM	19710	DZ AZ DZE – DZT VST VST MEP ESM STP STP STP
		ECL	.26515939 02 -42356230 01 .26251788 02 .25620490 02 .14879365 09 .15331122 03 .27453512-18 .27459388 04 .75483077 02	EQUAT	.69852606 00 .46608445 02 .46873893 02 .16204583-01	OCT. 21,1962	EGUAT	13876777 CC -42610605 C2 -32536981 C1 24080752 C2 61663945 C0 61663945 C0 39718221 06 13581218 03	ECLI	.26252078 02 .42891060 01 .26246801 02 .25612104 02 .14879122 09 .16939039 03 .10516900 03 .27453512-18 .69152626 04 .80697377 02
			DY DYE DYE DYT RYT EMP TSP RPT SIN 02		PTH PTR OP			DY - PTE DYS - DYM - DYT - RT RAM		DYE DYE DYE DYE DYE BAST EMS TSP TSP TSP SIN
TRAJECTORIES			9 04 DX16688477 C2 1 02 V -31332985 C2 0 02 DXE14370545 C2 0 04 DXT15122353 C2 2-02 LCT -27947641 C2 0 C2 EPM -26403508 C2 6 0C TPS -74917280 C2 6 0C RPM -74917280 C2 6 0C CPT -98494744 C2 3 C3 CPS -93276045 C2		0 04 CX15661244 C1 2 C3 V .18303510 C1 9 O3 VR .18223885 O1 9 O0 LNE .66122258 C1 2 O3 DR .13300721 C1 4 C3 SIA .33918417 C1	N DATE 2437959.20833333		2 06 DX219C2578 C1 3 C3 V -22198508 C1 8 C3 VE -26478846 C2 9 O8 DXS -14380115 C2 4 C6 DXM7486674 CC 4 C6 DXI7486674 CC 1 O6 VM -98528665 CC 1 O		9 04 DX16570413 G2 15 G2 V -31046412 G2 10 G2 DXE14380115 G2 12-02 LGT15128782 G2 12-02 LGT7968241 G2 19 G2 EPM -10429G30 G2 10 C2 SEM -74683481 G2 10 C2 CPT -95253494 G2 11 G3 CPT -95253494 G2 11 G3 CPT -95253494 G2
SPACE			.5C794999 .27949311 .21250C00 .464C5C00 .17869112- .75197170 .10127469 .28504796 .14770671		.20893210 .1080772 .16022589 93462259 .17886842	JULIAN		.12281552 .13571613 .211C6138 27645579 .11971284 .11971284 .3971821 .28117698		.57552499 .27970855 23500000 .47937500 .18459532- .74814539 .82630850 .18036143 .14720910
			2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2		RA LON LTE ALP HNG			LON LON ZM ZTT LOS DR		LON ZEF ZEF ZEF ZEF ZEF ZEF ZEF ZEF ZEF ZEF
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	OREIT		KAY YEE YES PESP KESP KEP KEP		CEC LAT LNS SHA SVL	NI A		CEC LAT YS YS YN YN YN DI		KATY YE COE COE STEEL ST
CASE 1	RA-5 POSTMIDCOURSE O	HEL IDCENTRIC	X .13143951 09 R .14879452 C9 XE .13171C99 09 XT .13144078 09 LTE81770582-05 EPS .10465351 C3 MPS .78723632 C2 EPT .26403508 02 SET .7491728C 02 GCE .1C304C68 C3 REP .40106557 06	SELENDCENTRIC	X12652C81 C4 R .44459388 O4 R .44459381 O4 LTS .15332178 O1 ALT .27079388 C4 HGE .25534648 C3	2 DAYS 15 HRS. 3	GEOCENTRIC	X27552655 C6 R .40398126 O6 R .40398125 O6 XS13168511 C9 XM27155858 C6 XT27155858 O6 RS .14889565 C9 GED .17811801 O2 DUT .340CCCOC C2	HEL IOCENTRIC	X .13140958 09 R .14879C34 09 XE .13168511 09 XT .13141355 09 LTE90429156-05 EPS .10503533 03 MPS .97366505 02 EPT .10429C30 02 SET .74683481 02 GCE .10294392 03 REP .40398126 06

18		COURDINATES	.50661255 00 .99031571 02 .82356751 02	.14556116 02	17 30 00.000	COURDINATES	_	.95357449 02		17472944 00			_	CCORDINATES	.33999872 00	.89367824 02	1054/230-02 85213186-01	.29742770 02	.18728645 00	.14720910 00	10452155 03	6856559	3630	CCORDINATES		.10325518 03	.83600229 02		70 8689/001•	18 00 00.000
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		EQUA	.47787168 CC .59671726 O2 .60233091 O2	.66983313-02	OCT. 21,	EQUA		.40214366 02		61972063 00	-39727166	3607457	1991010	EC	_	42531647 01		.14878879 09					8818268	EGUA	.39138084 00	.66259187 02			70-64066595•	0CT. 21,
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SPACE TRAJE			.31026752 04 .1299492 03 .17866846 03	7855751 0 6257541 0	JULIAN DATE		12335494	13615243		11939958 06	39727166	27367613			3845000	- 24250000 02	49480000	9053821-	74428753		14720910	95480275	<u>س</u>						.25368337 03	JULIAN DATE
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			.47380871 04 .26658426 02 .86528458 01	.68581449 .7151418C	52.COC SEC.		.26836462 06		63798089	26285409 06					.69833145 08					81-71CCC+17*		~	-21310924 01					901971979	1993886067	52.C00 SEC.
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CASE 1	RA-5 PCSTMIDCOURSE O	SELENOCENTRIC	X39679710 04 R .69152626 04 R .69152614 04		2 DAYS 15 HRS. 33	GEOCENTRIC	27938342	R -40656008 06	13165922	XM - 27290335 06	.14889480		DUT .34c0000c 02	HELIGCENTRIC		13165922	13138631	1	.10542043	CD 11/CIDOT - CAE	77070061-	m	REP .4C656C08 C6	SELENDCENTRIC		.93809357	.93809347	AIT 764203482 UL		2 DAYS 16 HRS. 3

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			LON LON ZS - ZT ZT RH LOS		ZE ZE ZI LTT SEP SMP TEP SIP SIP	א ה	RA LON LTE ALP HNG			RA LON ZS ZH ZH ZT LOS DR														
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			.26790734 .17622528 .17622528 .26173583 .26173583 .26173583 .26173583		.69880141 0 .26894326-0 .69585089 0 .69872597 0 .27862195 0 .15114412 0 .15114412 0 .1563652 0 .1563652 0	01967107•	.61715039 .23642092 .96250930 .26093378 -10985808	52.C00 SEC		.26738318 .17580220 .17580220 .26061207 .26061207 .29928619 .46489534														
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-4	POSTMIDCOURSE	TRIC	28310789 .40896695 .40896694 13163331 27424242 27424242 .14889396 .1735119	ENTRIC	6 6 2 2 3 4 5 6 5 6 5 6 6 5 6 6 6 6 6 6 6 6 6 6 6	REP .40896695 SELENDCENTRIC	88654674 -11791735 -11791733 -15337130 -10053735	DAYS 16 HRS.	ITRIC	28673984 -41127157 -41127156 13160738 27557579 27557579 -14889312 -17692576														
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21	CCORDINATES	.37968200 00 .10794289 03 .85155731 02	19 30 00.000	.18896244 00 .96706126 02 .26961763 03 10430942 02 18027706 00 .98424447 00 .17486211 03	COORDINATES -30770099 00 -89424185 02 -89424185 02 -8999406 00 -14687642 00 -14687642 00 -14687642 00 -14687642 00 -11487686 00 -14397686 00 -14397686 00 -14397686 00 -14397686 00 -14397686 00 -14397686 00 -14397686 00 -14397686 00 -153194221 01	20 00 00-000
	EQUATORIAL	DZ AZ AZR ASD	21,1962	02 AZE AZE DZS DZN	ECLIPTIC 02 DZ 01 AZ 02 DZE 03 MEP 03 ESM 03 ESM 05 DI 01 D3 EQUATORIAL 02 AZ 02 AZ 03 ASC	21,1962
	EQUAT	.30374248 00 .75167124 02 .76925745 02 .11808532-02	OCT. 21,1962	34023787 37691583 25282679 24057824 63189603 63189603 39762514	25984833 0 41248511 0 26221812 0 12877934 0 16428387 0 10633700 0 27453512-1 18747014 0 85508774 0 1465930 0 14659330 0 14659330 0 14659330 0 14659330 0	OCT. 21,
		PTH PTH O		DY	DY EMPTH DYTH DYTH BAND SIN SIN DY DY DY DY DY DY DP TR	
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SPACE TRAJECTORIES		.61508249 .15131234 .19773268 10795379 .17506127	JULIAN	.12494592 .13782115 .17556174 27739506 .11812152 .39762514 .24367281	.86917499 .28077400 .31500000 .55607500 .21414802- .72941543 .61031578 .69994006 .14687642 .1364268 .1097936 .15321841 .19934937	JULIAN
		RA LON LTE ALP HNG		RAA LON ZS ZS ZM ZT RM LDS	LON LON ZT ZT LTT SEP SMP SMP TEP CPE CPE LON LON LTE ALP	
		002 003 003 003 003 003 003 003 003 003		00 00 00 00 00 00 00 00 00 00 00 00 00	08 00 00 00 00 00 00 00 00 00 00 00 00 0	
		.73323870 .21933679 .10116789 .26042594 -14653586	52.000 SEC	.26621578 .17491591 .17491591 .17491397 .25834812 .25834812 .29929063 .4C932607	.70020682 (.33474525-0.69726732 (.27924380 (.27453512-).16428387 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.19758740 (.19758740 (.19758740 (.197540 (.19758740 (.19758740 (.1975840 (.1975840 (.1975840 (.1975840	52.000 SEC
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-	RA-5 POSTMIDCOURSE SELENDCENTRIC	13399730 .16466612 .16466609 .15340442 .14728612	17 HRS	29381304 1570233 -41570231 13155547 2782529 2782529 14889143 .17603454	HELIDCENTRIC X .13126166 R .14877C02 XE .13155547 XT .1312712698 EPS .10690540 MPS .11896216 EPS .10690540 MPS .105016216 SET .73516162 GCE .10252943 REP .41570233 SELENGCENTRIC X15587754 R .18747C114 LTS .15342C73 ALT .17009C14	DAYS 18 HRS
CASE	RA-5 SELEND	X R R ALT HGE	2 [GEDCENTRIC X - 29 R -41 R -41 XS - 14 XY - 27 XT - 27 XT - 27 RS - 14	HELIDO REPS REPS REPS SET SET SELENC SELENC ALTS HGE	7

22		COGRDINATES	.17962242 00 .96912535 02 .26961176 03 10428950 02 18165249 00	.98404051 .16760266 .17228506	COORDINAT	.89430948 02 -810211468-02	.29723127 02	.14604142 00 .59602743 02	.10638801 03 .24916297 01 .12139802 00	COORDINATES	.36127491 00 .10921063 03 .85463539 02	.47477660 01	20 30 00.000	COORDINATES	.17198157 00 .97099167 02 .2696698 03 10426957 02 18362331 00 18302331 00 .98383804 00 .16034296 03
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		EQUA	3521144 .3757874 .2480673 2405322 6349023	.39771243 C .13738262 O 10744755 O	25945212 0	41039020 01 -26216803 02	.14877668	.10657028		EQUA	.28278792 00 .77929262 02 .80366889 02	.72554192-03	OCT. 21,1962	EQUA	36184120 00 -37540929 02 -24422383 C1 24048629 02 63789364 00 63789364 00 63789365 00 13764345 03 10752169 C2
			PTH PTE DYS DYM	RAM	2	PTH	RST	EMS	RPT SIN D2		PTH PTR	90			PTH PTH PTH DYS DYS DYT RAM DES
SPACE TRAJECTORIES			.12527734 06 DX19154212 C1 .13822212 03 V .19557829 G1 .16844218 03 VE .2755708C C2 27758281 08 DXS .14437498 C2 .11779578 06 DXM72954582 C0	03 RAS .20595193 01 SHA39870588	0 81868E91 - XO	28098626 02 V .30687172 0 34000000 C2 DXE14437498 C	04 DXI1516/044 C 02 LOT -28C91785 C 02 FPM .15930725 C	02 SEM .73283026 C	.146C4142 00 RPM .2C998113 05 .11564251 03 CPT .90522128 02 .10086214 03 CPS .93310C79 C2		.74815633 04 DX11858753 01 .15470168 03 V .12715301 01 .20055847 03 VR .12612002 01 11372743 01 NF .64660785 01	03 DR 03 SIA	JULIAN DATE 2437959.35416666		.12559357 06 DX18970548 01 .13861743 03 V .19389367 01 .16.31695 03 VE .27725582 02 .2777752 08 DXS .14447056 02 .11746757 06 DXM72631362 00 .11746757 06 DXM72631362 00 .39779528 06 DXM98383804 00 .22867114 03 RAS .20597163 03 .11814485 01 SMA39996289 06
			LON ZZ ZZ ZZ		7	•			EST SIP CPE		RA LON				LONA 2S Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z
	POSTMIDCOURSE ORBIT		06 CEC 06 LAT 09 YS	02 ALT .41148666 02 DT .24000000	76474007. Y 60	09 LAT .35593688- 09 YE .69773928	03 ESP .15355230	03 MSP .27453512- 02 ETP .16323894	o in m		95 63	05 SHA18111668 03 SVL .85308217	IS. 33 MIN. 52.000 SEC.		10 06 Y .26494944 06 19 06 DEC .17399550 02 18 06 LAT .17399550 02 19 09 YS64057973 08 15 06 YH .25606247 06 15 06 YH .25606247 06 16 02 ALT .41362231 06 10 02 DT .48000000 03
CASE 1	RA-5 POSTMIDCO	GEOCENTRIC	X29727978 R -41786233 R -41786232 XS13152949 XM27954137 XT27954137	.175575 .340000	x •13123221	R .14876605 XE .13152949	1		.7328302 .1024529 .4178623	SELENOCENTRIC	, , , ,	ALT .19260113 HGE .25273742	2 DAYS 18 HRS.	GEOCENTRIC	X30071060 R -41999859 X -41999858 XS13150349 XM28085165 XT28085165 XT28085165 RS -14888974 GED -17510899

CASE 1				SPACE TR	TRAJECTORIES	IES						54
RA-5 POSTPIDCOURSE OR	OREIT											
SELENDCENTRIC									w	ECUATORIAL	IL CCORDINATE	TES
055 055 01		.93789469 04 .20149934 02 .1C471646 02 .25941C24 03	RA LON LTE	.87603C47 04 .15686809 03 .20219561 03 11948758 01	DX DX VR LNE	11582516 .12398576 .12298518 .64234099		PIR	.27C80702 .79816085 .82863227	00 DZ 02 AZ 02 AZR	.34994679 .11001530 .85465207	5 00 0 03 7 02
_23692664 05 _25203784 03	SHA -	21450401 05 .85086346 01	AL P HNG			.12203634 .13214388	C1 02	- - -	.49391629-0	D3 ASD	.39188030	10 0
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8 8	LEC	.17305216 02	LON	.13939229 03 .14705074 03	> w	.19119243 .28C57603	01 02 9	PTH PTE		02 AZ 01 AZE	•	
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.13114396 09 .14875428 09			LON	.10859250 05 .28162233 02	× >	16333672	0.2 0.2	07 PTH -	25919368	02 DZ		
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-15200568-04			רוו	.23767420-02					.25537169 .14876943			3-01
	ESP		SEP	71539971		117517552					11969351	61
.17517952 02			TEP	11969351					-10726949 -27453512-			000
SET .72584262 02 GCE .10223043 03	STE	.1C726949 03	EST	14604142				RPT			107443	160
.42422289 06	VEP	19119243	CPE	957				20	10934307	03	.78285311	- 1
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.25880941 05 .25169495 03	SHA -	23089460 05 .84845147 01	ALP	.17350586 03	DR	.13910070	22	00	-41832172-0	3 ASD	.3607882	6 01
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25		000000000000000000000000000000000000000	S	005 005 001 001 001 001	S	00 03 02 01	00	, 8	000000000000000000000000000000000000000
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	CCORDINATE	15520476 97584222 26959662 10420968 18710782 18710782 98323960	COORDINAT	.29393864 .89448399 .10176897- .84627270- .29707522 .13110285 .14587384 .56049748 .17531190	COORDINATE	34231257 11059241 85169094 33444C63	30 (150913°	97729 26959 10418 18845 18845 98304 13130
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7.2	EQUATORIAL COORDINATES	DY .25833370 00 DZ .33686268 CO PTH .82222180 02 AZ .11104660 03 PTR .86462960 02 AZR .84417527 02 DP .27324191-03 ASD .29214785 01	DCT. 21,1962 23 30 CO.000 EQUATORIAL CCORDINATES	1 DY39902500 00 DZ .14354839 00 1 PTH .38C55305 02 AZ .98001562 02 2 PTE .23C53971 01 AZE .2695894 03 2 DYS24C20962 C2 DZS10414966 02 0 DYH65552288 00 DZM19115006 00 0 PT .39831123 06 VT .98265458 00 3 RAM .13920304 03 LQM .11677936 03 6 DES10796628 02 DEM .16848428 02	ECLIPTIC COORDINATES DY .25872656 02 DZ .26942943 00 PTH40105607 01 AZ .89456312 02 DYT .25504176 02 DZE10153055-02 DYT .25504176 02 DZT .84400653-01 RST .14875980 09 VST .29695883 02 EMP .16014829 03 MEP .16302087 01 EMS .10820033 03 ESM .14553811 00 TSP .27453512-18 STP .54536249 02 SIN .86679351 02 DI .14404581 01 DZ .86679351 02 DI .14404581 01	EQUATORIAL CGGRDINATES 1 DY .25649788 C0 DZ .33469845 C0 1 PTH .82657356 C2 AZ .11124566 C3 1 PTR .87191453 C2 AZR .83718501 C2 1 DP .24149618-C3 ASD .27489598 C1
SPACE TRAJECTORIES		Z .11227549 05 DX11241793 C1 RA .15949737 03 V .12016622 C1 LON .20378271 C3 VR .11928793 01 LTE13096546 01 LNE .63369C48 C1 ALP .17303C89 C3 DR .11906072 C1 HNG .23454969 03 SIA .15216891 C2	JULIAN DATE 2437959.47916666	Z .12727878 06 DX18249315 C1 LON .11846157 03 V .18735533 C1 LON .11846149 C3 VE .2871C3C6 C2 ZM .11544687 06 DXX .14504373 C2 ZT .11544687 06 DXM70665511 C0 RM .35831123 C6 VM .98265458 CC LOS .18366619 03 RAS .20608588 C3 DR .11548992 01 SMA4C701954 C6	Z .129645CC C5 DX16329305 C2 LON .28246930 02 V .3C596148 C2 ZE46500C00 02 DXE14504373 C2 ZT .678C0C0C C4 DXT15211C28 C2 LTT .26113598-C2 LOT .28235813 G2 SEP .7C2C8402 02 EPM .18221490 C2 SMP .54536249 C2 SEM .71654C55 C2 TEP .16302087 C1 TPS .12545238 C3 EST .14553811 CC RPM .36238519 G5 SIP .12270342 03 CPT .89428312 C2 CPE .1C007968 03 CPS .93344C14 C2	Z .118319C7 C5 DX11182764 OJ RA .15995385 G3 V .11951386 C. LON .2C397816 G3 VR .11867632 C LTE13382586 C1 LNE .63150323 CJ ALP .17290979 G3 DR .11853380 CJ
CASE 1 RA-5 PCSTMIDCGURSE ORBIT	SELENDCENTRIC	X30159313 05 Y .11277682 05 R .34100266 05 EEC .19223324 02 R .34100261 05 LAT .1C6C3671 02 LTS .15353548 01 LNS .25839450 03 ALT .32362266 05 SHA27924371 05 HGE .25069165 03 SVL .83851235 01	2 DAYS 21 HRS. 33 MIN. 52.COO SEC. GEOCENTRIC	X32076752 06 Y .26081849 06 R .43257135 06 CEC .17111829 02 R .43257134 06 LAT .17111829 02 XS13134715 09 YS64317556 08 XM28859C11 06 YM .24907751 06 XT28859C11 06 YT .24907751 06 RS .14888469 09 VS .25930836 02 GED .17221566 02 ALT .42619501 06 DUT .34CC0000 02 DT .95999999 03	X	X3217741C 05 Y .11740977 05 R .36238519 05 CEC .19056527 02 R .36238513 05 LAT .1C62358C 02 LTS .15355179 C1 LNS .25814C56 03 ALT .34500519 05 SHA29515649 05

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	COGROINATE		.20121688 .89615704 .73802470 60506641 .30710102 .15769463 .15769463 .15769463 .15769463 .1680374 .16803274 .16803274 .16803274	49742067 07 -18000000 03 -70336431-02 -99997435 00 -0000000 00 00 00 00 00 00	18133283 .11771276 .26996447 22074645 32144642 .96827196
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TRAJECTORIES		2433 SLR DAC TXX TXX PER	CX C	SLR DAI DAI RXX TXX TX PER 243	DXS CX
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	POSTMICCOURSE	ERICE 727146 69648 25723 3030 70921 12282 12282 59838	X .10232351 R .14625629 XE .10460943 XT .10496843 LTE .1317297- EPS .14570511 MPS .145705116 EPT .29334318 SET .12516661 GCE .86861405 REP .23651224 HELIOCENTRIC	806519 171264 127941 127941 546721 976399 546729 503761 22 HR	708818 969821 969823 789115 959250 959250
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30	COORDINATES	.17205024 00 .89671365 02 .74642896-03 .58967866-01 .3947127 02 .30626943 02 .36626943 02 .14376236-01 .4376236-01	00 00 CO.CCO	64097070 C0 11636886 G3 2699598 C3 243C2874 G0 34579959 G0 3457959 G0 3457959 G0 3457959 G0	COORDINATES	.80857459-01 .89846533 02 .18119626-03 .13126328-01 .29918763 02 .13275685 02 .14620880 00 .54137662 02 .12257863 03 .66048595-02	00 00 00	-10293047 C1 -10293047 C1 -10962635 03 -26996241 03 -25980955 00 -25980955 00 -25980955 00 -1291710 03
	ECL IPT IC	DZE DZE DZT VST ESP SPN SPN 03	21,1962 Quatorial	DZ AZE DZS DZM DZT VT LCM	LIPTIC	DZE DZE DZT VST VST NEP PSP SPN D31	, 1963	GUATURIAL 01 DZ 02 AZE 02 DZS 00 DZM 06 VI 06 VI
	EC	02 02 03 03 03 04	ш	000000000000000000000000000000000000000	ECL	01 00 00 00 00 00 00 00 00 00 00 00 00 0		ė.
		.14749597 25495057 .15603896 .14654326 .14646543 .11077637 .85657989 .36268189	DEC.	16818810 -62473676 -17365927 56091862 83091859 40447423 23435408		11867510 84292644 28859982 28859982 14701751 .16605016 .11264039 .25725479 .78941876	JAN.	22886691 44818860 11117072 50265706 50265706 50265706
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TRAJECTORIES		C C D D D C C C D D C C C D D C C C C C	E 2438	SA CO		DX C CPT CCPT CCPT CCPT CCPT CCPT CCPT CC	243	S S S S S S S S S S S S S S S S S S S
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		LON ZE ZE ZE ZT LTT SEP SMP TEP TEP TEP TEP TEP SIP SIP SIP		LON LON ZS ZM ZT LOS DR		LON ZE ZE ZT LTT SEP SMP TEP EST SIP CPE		LON LON ZS ZH ZH CDS DR
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		.12429552 .22083624 .12559652 .12567231 .58194066 .10009288 .85657989 .14646543 .11077637	52.C00 SEC.	46061700 70907403 70907351 13498419 14636600 14636600 14636600		.14249516 .35916549 .14712811 .14698429 .88636659 .27184108 .25725479 .16605016 .1204039 .85567551	52.000 SEC.	99051560 14426043 14426042 11788523 30426800 30426800 30426800
	ORBIT	LAT YE YE LOE ESP ESP ESP ESP ESP ESP ESP ES	N N	CEC LAT VS VT VT VS ALT DT		LAT YE YE YT LOE ESP MSP ETP STE GCT	X X	CEC LAT YS YH YT VS ALT DT
		088 003 005 005 005	•	07 07 07 06 06 09		03 03 03 05 07	Б	07 08 08 06 06 07 02
-	POSTMIDCOURSE Entric	.74182333 .14475050 .77891151 .77895226 -102343412 .13943412 .14251655 .32226244 .69077197 .87412909	22 HRS	68129385 82873064 82873078 35015344 37629950 37629950 37629950	ENTRIC	33114041 -14253643 -35015344 -31252348 -12262272 -12328978 -67412505 -67813645 -85683756	DAYS 22 HRS.	78561860 13054053 13054053 13054052 24142000 24142000
CASE	RA-5 POST HELIOCENTR	P C C C C C C C C C C C C C C C C C C C	62 DAYS Geocentric	Z X X X X X X X X X X X X X X X X X X X	HEL IOC	R G S E T X X X X X X X X X X X X X X X X X X	10 25 DI	X 78 R 13 X S 71 X X 24 XM 24 XT 24 RS 14 GED 14

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RA-- F POSTMIDCOURSE ORBIT

SPACE TRAJECTORIES

COORDINATES	19318378	00 00 00 00.000 . CCGRDINATES	.49972093 00 .82646316 02 .27001555 03 .10391087 02 .3318996 00 .33318996 00 .10630125 01 .13105021 03	. CCCRDINATES 19732344 00 .90381610 02 .80633163-03 65827607-01 .28974972 02 .37491029 02 .37491029 02 .11364091 00 .92057271 02 .81373437 02 .81373437 02	GO GO CO.000	.85741550 00 .76913942 02 .27602475 03 .62780137 01 .40498602 00 .40498602 00 .10825367 01 .13751C27 03
ECLIPTIC	DZE DZE DZT VST WEP ESM SPN SPN SPN	20,1963 EQUATORIAL	DZ AZ AZE DZS DZM OZT VT LCM	ECLIPTIC 2 DZ - 1 DZE 2 DZT - 9 VST 9 WEP 3 MEP 1 STP 1 STP 8 SPN 2 D3	20,1963 Equatorial	DZ AZE DZS DZM DZT VT LOM
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	DYEN DYEN DYEN DYEN DYEN DYEN DYEN DYEN		PTH PTH DYS DYM DYM RAT DES	DY PTH DYE DYT RST RST ERS TSP RPT SIN SIN		DY PTH PTE DYS DYM DYT RAM DES
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	18421193 .11421613 .88178379 .55033899 .17966094 .74873609 .74873609 .15650247 .11945816 .75052807	52.000 SEC.	15641205 21221995 21221997 .67283399 132608C0 132608C0 .29657570 .19401325	90481862 83896065 73396840 73490539 -20920802 -73357782 -72220378 -12597094 -77432042	52.000 SEC.	1C197149 17485859 17485859 .11814162 .9C453999 .29442947 .17280CCC
	LOAT YET YET YET YET YET STEP STEP YET	* I X	CEC LAT YAY YAY VAN OT	LAT YE YT YT KSP ESP ESP ESP ECT VEP	Z X	EEC Y Y Y Y Y Y Y Y Y Y Y Y Y Y Y Y Y Y
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HEL IOCENTRIC	X14690259 R .14805336 XE14900538 XT14811579 LTE10224188-EPS .89068019 MPS .88621512 EPT .44685877 SET .60414711 GCE .75070792	182 DAYS 22 HRS. GEOCENTRIC	X .90916149 R .19407701 R .19407700 XS .13117766 XM .33511200 XT .33511200 RS .15028596 GED21353633	7 7 8 1 1 3 0 8 1 1 0 8 1 1 0 8 1 1 1 3 0 8 1 1 1 3 0 8 1 1 1 3 0 8 1 1 1 3 0 8 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	212 DAYS 22 HRS. GEOCENTRIC	X .12858863 R .17206433 R .17206433 XS .79584970 XM .35127600 XI .35127600 RS .15137917 GED17597690

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.89644731 .17280CCC

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34		ECLIPTIC COORDINATES	C2 DY .13665421 C2 DZ .16938448-C1 C2 PTH60334475 C0 AZ .89969787 C2 DYE .12718507 C2 DZE10859370-C2 DYT .127181595 C2 DZT .90659558-C1 C3 RST .15170226 C9 VST .28365853 C2 EMP .77057363 C2 MEP .10063796 C3 C2 EMS .15558794 C3 ESM .57674939-C1 C2 TSP .26212300 C1 STP .12730182 C3 C7 RPT .9046499C C7 SPN .52339783 C2 SIN .90294090 C2 D3 .84358129-C6 C2 C3	.13364548 07 JULY 23,1963 03 28 23,287 .13364548 07 ANG. 5,1963 00 00 00 000	EQUATORIAL CO	01 DY .23252583 CO DZ .176E0949 CO 01 PTH8232463 C2 AZ .11976829 C3 PTE18948481 CO AZE .26998716 C3 C2 DYS18127615 C2 DXS78662705 C1 CO DYM .63939571 CC DZM .18010169 CC C DYT .63939571 CC DZM .18010169 CC C1 RT .38211805 C6 VT .10336356 C1 C3 RAM .30974051 C3 CM .35691538 C3 C7 DES .17230983 C2 DEM19986186 C2	ECLIPTIC COORDINATE 2 02
SPACE TRAJECTORIES			Z10683825 07 0X .24772199 LON .29827990 03 V .28291445 ZE .10600000 03 DXE .26383382 ZI11090500 05 DXI .25322801 LTT41887234-02 LOT .29568706 SEP .12494675 03 EPH .23046707 SMP .12730182 03 SEM .24354408 TEP .10663796 C3 TPS .50676946 EST .57674939-01 RPM .96464590 SIP .50065940 02 CPT .96305098	JULIAN DATE 2438233.64471 .24396987 07 .29276218 07 JULIAN DATE 2438246.50000		Z11821415 07 DX15404861 RA -35819725 03 V -15679372 LON -45372136 02 VE -46976109 ZS -44956386 08 DXS21683536 ZM13060550 06 DXM -79192114 ZT13C60550 06 DXT -79192114 RM -38211805 06 VM -1C336356 LOS -18151314 03 RAS -13433827 DR15535508 01 SHA -44721719	Z10037865 07 DX .20143050 LON .31347094 63 V .28423505 ZE .11900000 03 DXE .21683536 ZT98495000 04 DXT .22475457 LTT37091880-02 LOT .31186401 SEP .13695596 03 EPM .25656815 SMP .13438353 03 SEM .17484519 TEP .47570357 02 TPS .43968946 EST .13988227-01 RPM .63005606
CASE 1	RA-5 POSTMIDCOURSE ORBIT	HELIOCENTRIC	X .74547100 C8 Y13856536 C9 R .15734920 C9 LAT38903476 C0 XE .65764808 C8 Y13671027 C9 LTE .39945912-C4 LCE .25562957 C3 EPS .52380519 C2 ESP .26787155 C1 MPS .50076946 C2 PSP .267123CC C1 EPT .23046707 C1 ETP .77057363 C2 SET .24354408 C2 STE .15558794 C3 GCE .10179331 C3 CT .1C188449 C3 REP .89708507 C7 VEP .18392528 C1	277 DAYS 1 HRS. 32 MIN. 15.287 SEC. RECTIFICATION93751343-02 289 DAYS 22 HRS. 3 MIN. 52.000 SEC.		X .64413390 07 Y20273600 06 R .65520537 07 DEC10394403 02 R .65520541 07 LAT10394401 02 XS10130672 C9 YS .1C367421 09 XM .22958C00 06 YM27613300 06 XT .22958C00 06 YT27613300 06 RS .15176452 09 VS .29343262 02 GED10463677 02 ALT .65456761 07 BUT .34000C00 02 DT .17280CCC 06	HELIOCENTRIC X

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37		COORDINATES	.24588253 CC .28596216 03 .27014636 03 .11276106 02 .14227442 00 .14227442 00 .16211296 01 .35462543 C3	COGREINATES	07 39 44.191 -15606153 07 -88201469-02 -11145496 03 -40219652 00 -92439199 00 -00000000 00 -25177948 00 -25177948 00	CCORDINATES	.25552714 00 .8951418C 02 .84599493-01 .29594682 02 .14238823 03 .13953216 C0 .78061654 02 .10815095 03 .27743990-C1	000 00 00 0	COORDINATES	29419546 CC 29499272 03 27016760 03 10235587 02 75681849-02 75681849-02 98045491 00 24954635 03
			DZ -2 AZ -2 AZE -2 DZS1 DZM1 LCM -3 DEM -3		963 07 RCA - 1 TFP - 8 MTA - 1 PZ 4 RZ 9 MZ - 0 MZ - 0	ECLIPTIC CO	DZE	00 696	AL	DZ AZE DZS DZM DZM VT UCM CEM
	1	ECUATORIAL	7878036-01 0928119-06 8371003-06 6CC6106 02 4335948 0C 4335948 0C 4335948 0C 8382792 06 8382792 06	EQUATORIAL	0CT. 11,19 .30204701 03 .00CC0000 C0 .16363258 03 .915C8337 00 .10748933 00 .95947435 00 .79745767-01	ECT 1	28371885 02 31841761 01 28345513 02 27790403 02 14922318 09 30451373 02 10848931 03 70779359 00 18793268 07 76539061 02	OCT. 24,1963	EGUATORI	C8518-01 45445 02 26790 C0 01408 02 79961 C0 20068 C6 22252 03
			1 1111 1 	œ	• •					H .27245 E .20426 S -23601 F .23479 T .39926 M .28122
			DY PTH PTE DYS DYM DYT RAM	AND 8	APP RADO PY RADO PY RADO PY RAY RAY PY RAY PY RAY PY RAY PY RAY RAY RAY RAY RAY RAY RAY RAY RAY RA		DY DYE DYT DYT RAT EMS TSP TSP SIN			DY PTH PTH DYS DYT RT RAM DES
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TOR 16			DX V V C C C C C C C C C C C C C C C C C		2438 CAN DAO PXX TXX FEX		DX DX DX CPR SPRENCE CPR SPREN	2438		D D D X X X X X X X X X X X X X X X X X
SPACE TRAJECTORIES			62767404 06 .26816405 03 .13405230 03 17641680 08 .13420267 06 .38382792 06 .61810344 02 .25030191-08	CONIC	JULIAN DATE .15167288 G3 .15240640 G7 .37366633-06 .28025284 G0 .25177947 GC .38144381 GC -28223317 GG		77651250 04 -167C4476 02 5999959 02 -11470250 05 -44041206-02 -71046718 02 -78061654 02 -14238823 03 -14238823 03 -14238823 03	JULIAN DATE		32518465 C6 -23235914 03 20C68298 03 29477281 08 15408527 C6 15408527 C6 1540893 03 -46811473 C0
			LON LON ZS ZT ZT RM LOS		INC C1 MA WZ QZ SZC 8ZC 8Z		2 10N 2E 2T 2T 2T 2T 2F 2F 2F 2F 2F 2F 2F 2F 2F 2F 2F 2F 2F			LON LON 25 27 27 27 27 27 27 27 20 20 20 20 20 20 20 20 20 20 20 20 20
			14280931 07 23715566 02 23715566 02 46683096 08 .28049827 06 .26049827 06 .29839116 02 .15542406 07		GE .27339608 01 .44287869 00 .00000000 00 .39530119 00 .79745768-01 .26048994 0088084161 0069921851 06		.42783543 68 29890593-02 .44343467 08 .44654203 08 .17272642 02 .56816375 00 .7C779359 00 .30451373 02 .10848931 03 .28356232 03	52.000 SEC.		-14168706 07 -10300387 02 -10300387 02 -67976667 08 -36122305 06 -36122305 06 -36122305 06 -18122385 07
	OREIT		CEC LAT VA VY VS VS DT		PASSA(C3)		LATY YESPESSPESSPESSPESSPESSPESSPESSPESSPESSP	FIN.		PEC LAT YS YT VT VS ALT
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	POSTMIDCOURSE	TRIC	45776424 15606153 15606153 14261627 22502052 22562052 23859187 33859187	TRIC	OF PERICEN90002915655402082561320896449348886936412180782383124527	ENTRIC	.14256450 .14884578 .14261627 .14238525 -23018773- .10838511 .10123656 .71603924 .71371033	AYS 22 HR	TRIC	-1092747 -1818616 -18186166 -1290541 -7167163 -7167163 -1036906
CASE	RA-5	GECCENTRI	D G S X X X X X X X X X X X X X X X X X X	GECCENTR	NO SET CENTER OF	HEL IOCENTR	P P P P P P P P P P P P P P P P P P P	369	GEOCENTRI	X X X X X X X X X X X X X X X X X X X

COORDINATES

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.97648737 .14352841 .18065071 00 .89659747 02 .28949976-04 .29342138 02 .9855176-01 .29521050 02

DZ AZ DZT DZT VST WEP ESM STP STP

.91993210-01 .29521050 02 .14911328 03

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ECLIPTIC **EQUATORIAL ECLIPTIC** 25,1963 3,1963 NOV. 13,1963 3,1963 -.47243118-01 .67130975 02 .25495815 00 -.17522492 02 -.87470293 00 -.87470293 00 .40306640 06 .19491748 03 005 005 005 005 005 005 005 005 5 -.25813282 -.25725351 -.25943779 -.14867813 -.12091262 .15809929 .74170639 DEC. -.17522492 -.87470293 -.87470293 -.40306640 -.19491748 -19125885 -14217252 -19C99032 -18145765 -14772843 .54437597 .28018701 .76C71307 .16198G32-00.1 .16207647 .14377763 05 2438337.10815045 .19566645 05 -.13208364 .16745835 116000000 2438327.50815047 2438346.5000000 2438366.5000000 -15329369 -15329369 -14377474 -29716052 .10855179 -11127657 -11275985 -23348327 -23270138 -21168423 .74233626 .21168423 .97648737 .22743623 -.16058012 -.24382844 -.23270138 -.23058454 -.49974528 .14094492 .36130193 .14993452 .76106849 .98033109 .15809929 .16455989 05 .14230191 .19566645 TRAJECTOR LES DXE DXT DXT SEA TPS CPT CPT DATE 05 -. DATE ш DATE DAT .62783850 06 -56466177 02 -3700000 03 -3540600 05 -13732051-01 -30139212 02 -3651105 02 -98451746 01 -91993210-01 -14989897 03 .24160634 .15277899 .48232260 .15294142 -15294142 -45044083 -59949999 -40306640 .26540CG0 .29590266 .50749999 92285977-JULIAN 57331553 16449690 .23947500 JULIAN --44326459 .14452625 JUL I AN JUL IAN .90643606-02 6.200 SEC. SEC. .12091262 .10937268 .32284920 .10333734 .74092761 .73700053 .29861049 SEC. .20376725 .2037674 .2037607 -10374900 -10374900 .3016455 .31917748 .11205903 .24758069 .11322189 .16874535 .14377763 .76100218 SEC. .49883404 .16207647 .11312432 54437597 36.199 36.200 52,000 52.000 LAT YE YT YT COE ESP ESP ESP STE GCT Z Z ORBIT Z Z LAT YE YE YE ESP MSP ETP STE GCT X IN YIN. 1.5 39 .12796144 09 .14715360 09 .12905419 09 .12512586 09 .1555454 03 .16455989 03 .10855179 02 .70482319 02 .30458122 03 m POSTMIDCOURSE 07 07 07 08 08 09 09 00 02 008 003 003 002 002 HRS. HRS. HRS. -31981531 -95397739 -38943900 .14529688 .95397739 .14993452 .14C94492 .36130193 .78508750 -.29122640 .14805379 HRS. --14318740-.14922755 RECTIFICATION RECTIFICATION 12 22 22 HEL IOCENTRIC 22 HEL IOCENTRIC DAYS DAYS DAYS SET GCE REP 403

.23748863 00 .89550535 02 .10397434-02 -2961401 02 .4823220 02 .14452625 00 .15277899 03

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CCORDINATES

CASE	1						SPACE TRAJE	TRAJECTORIES	ıes					39	
R A-5	POSTMIDCOURSE	SE ORBIT	11												
GEOCENTRIC	TRIC											EQU	EQUATORIAL	COORDINATES	
××	48669050		CEC -	16927710	07	Z RA	.23026950 06 .19917832 03	× >		101	ı	.42921603 00 .84285168 02		•	
æ v	.51580291		1	.25587015	10		.128C7617 03 551633C2 08	ox s	.37571805			.18188111 OC 92C94716 C1	AZE DZS	39922215 01	
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×	51690999			.33013500	9	17		DXT	10755208					1	
R S	.14746897			.30199322	70	X (¥ (.1C8481C8			.36272774 06	- 1 > 0	10 80184801.	
GED	.25761151	01 02	ALT DT .	.51516506 .17280000	2 90	08 08	111926771 01	N W I	41733085		DES -			.22892789 02	
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APPENDIX D

Tables related to trajectory printout

Table D-1. Ranger 5 trajectory key

CO RO	LUMN	1	2	3	4	5	6	
	1	GME		Н	D	RE	REM	
GROUP A	2	G	Ă	В	C	OWE	AU	
	3	GMM	GMS	GMV	GMA	GMB	GWI	
	INJ	ECTION CO	ONDITIONS	TARGET	JULIAN E	ATE A	AONTH DAY, YEAR	hr min sec
ROUP B		GEOCENTR		YO	zo	DXO	DYO DZO	
SKOUP B		CARTESIAN		. •	то	GHA	GHO	
	-	#111F D4	ST INJECTION		JULIAN I	DATE	MONTH DAY, YEAR	hr min sec
					302.7		EQUATORIAL COO	RDINATES
		GEOCENTR			= 17	BV.	DZ	
	6	X	Y	Z RA	DX V	DY PTH	AZ	
nolin c	7	R	DEC LAT	LON	VE	PTE	AZE	
ROUP C	8 9	R XS	ŶS	ZS	DXS	DYS	DZS	
	10	XW	YM	ZM	DXM	DYM	DZM	
	11	XT	YT	ZT	DXT	DYT RT	DZT VT	
	12	RS	VS	RM LOS	VM RAS	RAM	LOM	
	13 14	GED DUT	ALT DT	LOS DR	SHA	DES	DEM	
						ITAL BATANDE	R EQUATORIAL COC	PRDINATES
		GEOCENTR	IC					
		EPOCH OF	PERICENTER PASS	AGE	JULIAN DA	TE	MONTH DAY, YEAR	hr min sec
	15	SMA	ECC	INC	LAN	APF	RCA	
	16	VH	C3	C1	SLR	APO	TFP MTA	
GROUP D	17	TA	EA	MA W7	DAO PX	RAO PY	PZ	
	18	WX	WY QY	WZ QZ	RX	RY	RZ	
	19 20	QX SXO	SYO	szo	TX	TY	TZ	
	21	BX	BY	BZ	WX	MY	MZ	
	22	B · T	B•R	В	PER	OWD	NOD	
	23	C31						ND IN LATER
		HELIOCEN1	rric				EQUATORIAL COC	ORDINATES
	24	X	Y	Z	DX	DY	DZ	
	25	R	LAT	LON	_ V	PTH	AZ DZE	
	26	XE	YE	ZE	DXE DXT	DYE DYT	DZT	
GROUP E	27	XT	YT LOE	ZT LTT	LOT	RST	VST	
	28 29	LTE EPS	ESP	SEP	EPM	EMP	MEP	
	30	MPS	MSP	SMP	SEM	EMS	ESM	
	31	EPT	ETP	TEP	TPS	TSP	STP SPN	
	32	SET	STE	EST	RPM CPT	RPT SIN	DI	
	33 34	GCE Rep	GCT VEP	SIP CPE	CPS	D2	D3	
	34			J. J			EQUATORIAL COC	DOINATES
		SELENOCE	NTRIC		_		-	AUDITAILS
	35	X	Υ	Z	DX	DY PTH	DZ AZ	
000112.5	36	R	DEC	RA LON	V VR	PTR	AZR	
GROUP F	37 38	R LTS	LAT LNS	LTE	LNE			
	38	ALT	SHA	ALP	DR	DP	ASD	
	40	HGE	SVL	HNG	SIA			
		SELENOCE	NTRIC		CONIC ORE	ITAL B.T AND B	R EQUATORIAL COC	RDINATES
			PERICENTER PASS	AGE	JULIAN DA		MONTH DAY, YEAR	hr min sec
		SMA	ECC	INC C1	LAN SLR	APF APO	TFP	
GROUP G	42 43	VH TA	C3 EA	MA	DAI	RAI	MTA	
SKOUP G	44	WX	WY	wz	PX	PY	PZ	
	45	QX	QÝ	QZ	RX	RY	RZ	
	46	SXO	SYO	szo	DAO	RAO	TF TZ	
	47	SXI	SYI	SZI BZ	TX MX	YT YM	MZ	
	48 49	BX B•T	BY B• R	B	PER	DEF	C31	
	47				YOCTAL ZOCTA			
GROUP H	50	XOCTAL	YOCTAL ZO					

Table D-2. Ranger 5 trajectory key definitions

G	Group	Trajectory constant	G	roup	Trajectory constant
Group A Row 1	GME J H	Universal gravitational constant times the mass of Earth, km³/sec² Coefficient of the second harmonic in the Earth's potential function Coefficient of the third harmonic in the	Group C		Inertial position and velocity of the probe, Sun, Moon and target body in a geocentric equatorial system. The principal direction X is the vernal equinox direction of date and the principal plane XY is the equatorial plane of date. Z is along the direction of the Earth's spin axis of date. Miscellaneous parameters are also included.
	D RE REM	Earth's potential function Coefficient of the fourth harmonic in the Earth's potential function Earth radius used in the potential function, km Conversion factor for converting lunar ephemerides into km	Row 6	X Y Z DX DY DZ	Cartesian components of the probe radius vector, km Cartesian components of the probe space-fixed velocity vector, km/sec
Row 2	G A B C OME AU	Universal constant of gravitation, km³/kg sec² Moments of inertia about principal axis for the Moon, kg km² Sidereal rotation rate of the Earth, deg/sec Astronomical unit, km	Row 7	R DEC RA V PTH	Probe radius distance, km Probe declination angle, deg Probe right Ascension angle, deg Probe space-fixed velocity, km/sec Pitch angle of the probe space fixed velocity vector with respect to the local horizontal, deg Azimuth angle of the probe space-fixed velocity vector measured East of true North, deg
Row 3	GMM GMS GMV GMA	Universal gravitational constant times the mass of Moon, km²/sec² Universal gravitational constant times the mass of Sun, km³/sec² Universal gravitational constant times the mass of Venus, km³/sec² Universal gravitational constant times the mass of Mars, km³/sec²	Row 8ª	R LAT LON VE PTE	Probe radius distance, km Probe geocentric latitude, deg Probe East longitude, deg Probe Earth-fixed velocity, km/sec Pitch angle of the probe Earth-fixed velocity vector with respect to the local horizontal, deg Azimuth angle of the probe Earth-fixed velocity vector measured East of true North, deg
	GMB GMJ	Universal gravitational constant times the mass of Earth—Moon, km³/sec² Universal gravitational constant times the mass of Jupiter, km³/sec²	Row 9	XS YS ZS DXS DXS DYS	Cartesian components of the Sun radius vector, km Cartesian components of the Sun space-fixed
Group B		Injection conditions are vernal equinox cartesian coordinates in a geocentric equatorial system. The principal direction X is the vernal equinox direction of date and the principal plane XY is the equatorial plane of date. Z is along the direction of the Earth's spin axis of date.	Row 10	XM YM ZM DXM DXM DYM DZM	Cartesian components of the Moon radius vector, km Cartesian components of the Moon space-fixed velocity vector, km/sec
Row 4	XO YO YO DXO DYO DZO	Cartesian components of the probe radius vector, km Cartesian components of the probe space-fixed velocity vector, km/sec	Row 11	XT YT ZT DXT DYT DZT	Cartesian components of the target radius vector, km Cartesian components of the target space-fixed velocity vector, km/sec
Row 5	TO GHA GHO	Time of injection in seconds past midnight of day before launch, sec HA of Greenwich at injection epoch, deg HA of Greenwich at midnight of day before launch, deg	Row 12	RS VS RM VM RT VT	Sun radius distance, km Sun space-fixed velocity, km/sec Moon radius distance, km Moon space-fixed velocity, km/sec Target radius distance, km Target space-fixed velocity, km/sec

These are Earth-fixed spherical coordinates in a geocentric equatorial system. The principal direction X is directed towards Greenwich and is the intersection of the meridian plane of Greenwich with the equatorial plane. The principal plane is the Earth's geometrical equatorial plane X, Y, Z is along the direction of the Earth's geometrical north direction.

Table D-2. (Cont'd)

Gr	oup	Trajectory constant	Grou	P	Trajectory constant
Row 13	GED ALT LOS RAS RAM LOM	Geodetic latitude of the probe, deg Altitude of the probe above the Earth's surface, km East longitude of the Sun in coordinate system defined in Row 8, deg Right ascension of the Sun, deg Right ascension of the Moon, deg East longitude of the Moon in coordinate system defined in Row 8, deg	E A A	X X X X X X X X X X	Components of the impact parameter \mathbf{B} , \mathbf{b} km Components of a unit vector which lies in the orbit plane and is normal to the radius vector \mathbf{R} . $\mathbf{M} = \mathbf{W} \times \frac{\mathbf{R}}{ \mathbf{R} }$
Row 14 Group D	DUT DT DR SHA DES DEM	Ephemeris time minus Universal Time, sec Adams-Moulton step size, sec Radial velocity of probe, km/sec Sun shadow parameter, km Declination of the Sun, deg Declination of the Moon, deg Characteristics of the Earth conic in the geo-	B P	• R	Projection of the impact parameter B ^b upon the vector T, km Projection of the impact parameter B ^b upon the vector R, km The magnitude of the impact parameter, ^b km Period, min Rate of change of argument of perigee, deg/day Rate of change of RA of the ascending node, deg/day
		centric equatorial system described under Group B	Row 23 C	:31	Earth—Moon Jacobi constant, km²/sec²
Row 15	SMA ECC INC LAN APF RCA	Semimajor axis, km Eccentricity Inclination of the orbit plane to the equatorial plane, deg Longitude of the ascending node, deg Argument of pericenter, deg Magnitude of the closest approach vector, km Hyperbolic excess speed, km/sec	Group E		Inertial position and velocity of the probe Sun, Moon, and target body in a heliocentric equatorial system. The principal direction X is the vernal equinox direction of date and the principal plane XY is the equatorial plane of date. Z is along the direction of the Earth's spin axis of date. Miscellaneous parameters are also included.
Row 17	C3 C1 SLR APO TFP	Twice the energy (vis viva energy integral, km²/sec²) Angular momentum, km²/sec Semi-latus rectum, km Apogee distance, km Time from pericenter passage, sec True anomaly, deg		} (X) (Y) (X)	Cartesian components of the probe radius vector, km Cartesian components of the probe space-fixed velocity vector, km/sec
Row 18	MA DAO RAO MTA WX WY PX PY PZ PZ	Eccentric anomaly, deg Mean anomaly, deg Declination of the outgoing asymptote, deg Right ascension of the outgoing asymptote, deg Maximum true anomaly, deg Components of a unit vector normal to the conic $\mathbf{W} = \frac{\mathbf{R} \times \mathbf{V}}{ \mathbf{R} \times \mathbf{V} }$ Components of a unit vector in the direction of perigee	L V P	AT ON	Sun probe radius distance, km Probe celestial declination, deg Probe celestial right ascension, deg Probe space-fixed velocity, km/sec Pitch angle of the probe space-fixed velocity vector with respect to the local horizontal, deg Azimuth angle of the probe space-fixed velocity vector measured East of true North, deg
Row 19	QX QY QZ RX RY RZ	Components of a unit vector perpendicular to the perigee direction, vector P , and being in the orbit plane Q = W × P Components of the unit vector R ^b	Z D	E }	Cartesian components of the Earth radius vector, km Cartesian components of the Earth-space-fixed velocity vector, km/sec
Row 20	SXO SYO SZO TX TY TZ	Components of the unit vector S _o ^b along the direction of the outgoing asymptote Components of the unit vector T ^b	Z D	T	Cartesian components of the target radius vector, km Cartesian components of the target space-fixed velocity vector, km/sec

Table D-2. (Cont'd)

roup	Trajectory constant	G	roup	Trajectory constant
LTE LOE LTT LOT RST VST	Celestial latitude of the Earth, deg Celestial longitude of the Earth, deg Celestial latitude of the target, deg Celestial longitude of the target, deg Sun-target range, km Sun-target velocity, km/sec	38	,	Selenocentric-fixed spherical coordinates of the probe, Sun and Earth in a selenocentric equatorial system. The principal direction X is in the direction of the mean Moon-Earth line. The principal plane XY is the mean selenocentric equatorial plane. Z is along the direction.
EPS ESP	Earth-probe-Sun angle, deg Earth-Sun-probe angle, dea			tion of the Moon's mean spin axis. Miscella neous parameters are also included.
SEP			_	
EPM		Row 35	χЪ	Cartesian components of the probe radius
EMP			Y }	vector, km
MEP	Moon–Earth–probe angle, deg		Z	Yellof, Kill
			DY }	Cartesian components of the probe velocity
MPS	Moon-probe-Sun anale, dea		DZ J	vector, km/sec
	Sun-Moon-probe angle dea	1	_	
	Sun-Farth-Moon angle dea	1 Kow 36		Probe radius distance, km
				Probe declination angle, deg
				Probe right ascension angle, deg
LOM	Larin-Sun-moon angle, deg	1		Probe space-fixed velocity, km/sec
		İ	PTH	Pitch angle of the probe space-fixed velocity
EDT	Forth proba toront apple dos	İ		vector with respect to the local horizontal, deg
			AZ	Azimuth angle of the probe space-fixed velocity
				vector measured East of true North, deg
		Pov. 37	D	Probe radius distance, km
		KOW 3/		Probe selenocentric latitude, deg
SIF	Sun-target-probe angle, deg			Probe selenocentric tattique, deg
CET	Sun Fauth toward and land	1		Probe selenocentric-fixed velocity, km/sec
		- 1	KIK	Pitch angle of the probe selenocentric-fixed
				velocity vector with respect to the local
			A7D	horizontal, deg
			72K	Azimuth angle of the probe selenocentric-
				fixed velocity vector measured East of the Moon's mean spin axis, deg
	Total properties and the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of the control of			
GCE	Clark angle of Earth dog	Row 38	LTS	Selenocentric latitude of the Sun, deg
			LNS	Selenocentric longitude of the Sun, deg
			LTE	Selenocentric latitude of the Earth, deg
			LNE	Selenocentric longitude of the Earth, deg
DI	Radius of a circle (target) used in construction	Row 39	ALT	Altitude of the probe above the Moon's
	ot visible planet, cm	İ		surface, km
				Sun shadow parameter, km
DED	Fruit to distance to		ALP	Illuminated crescent orientation viewing
			D.D.	angle, deg
,			UK	First time derivative of the probe radius distance, km/sec
CPE			DP	First time derivative of the probe radius
CPS		1	٥.	direction, deg/sec
D2			ASD	Angular semidiameter of Moon as seen from
	of visible planet, cm			the probe, deg
D3	Distance from intersecton of ellipse with			
		Row 40	HGE	Right ascension of Earth in probe coordinate
		1	-	system," deg
	Visible planet, cm		SVL	Declination of the Moon in probe coordinate
			unic	system, ^c deg
		1	HNG	Right ascension of the Moon in probe coordi-
	Inertial position of probe in a selenocentric	1	CI A	nate system, deg
	equatorial system. The principal direction	1	SIA	Earth-probe-Moon angle minus ASD, deg
	X is the vernal equinox direction of date and			
	the principal plane XY is the geocentric equa-	Group G		Characteristics of the selenocentric conic in the
	torial plane of date. I is along the direction	1		geocentric equatorial system described un-
	of the Earth's spin axis of date.			der Group B except centered at the Moon.
	LTE LOE LTT LOT RST VST EPS ESP EEMP MEP MEP MEP SEMS ESM EETP TEP TPS TSP STP STP STP STP STP STP CPT STP STP CPT STP STP CPT STP STP CPT STP STP CPT STP STP CPT STP STP STP STP STP STP STP STP STP S	LTE Celestial latitude of the Earth, deg Celestial longitude of the Earth, deg Celestial longitude of the target, deg SST Sun-target range, km VST Sun-target range, km VST Sun-target velocity, km/sec EPS Earth-probe-Sun angle, deg ESP Sun-Earth-probe angle, deg ESP Sun-Earth-probe angle, deg EMP Earth-Moon-probe angle, deg EMP Earth-Moon-probe angle, deg EMP Earth-Moon-probe angle, deg MSP Moon-Earth-probe angle, deg MSP Moon-Sun-probe angle, deg SSM SUN-Earth-Moon angle, deg EMS SUN-Earth-Moon angle, deg EMS Earth-Moon-Sun angle, deg EMS Earth-Moon-Sun angle, deg EMS Earth-Moon-Sun angle, deg ETT Earth-probe-target angle, deg ETT Target-Earth-probe angle, deg ETT Target-Sun-probe angle, deg TSP Target-Sun-probe angle, deg STP Sun-target-probe angle, deg STP Sun-target-probe angle, deg STP Sun-target-probe angle, deg STP Sun-target-target angle, deg STP Sun-target-target angle, deg STP Sun-target-probe angle, deg STP Sun-target-probe angle, deg STP Sun-target-target angle, deg STP Sun-target-target angle, deg STP Sun-target-target angle, deg STP Sun-target-target angle, deg STP Sun-target-target angle, deg STP Sun-target-target angle, deg STP Sun-target-target angle, deg STP Sun-target-target angle, deg STP Sun-target-target angle, deg STP Sun-target-target angle, deg STP Sun-probe-near limb of target angle, deg Canopus-probe-near limb of target an	LTE Celestial latitude of the Earth, deg Celestial longitude of the Earth, deg LTT Celestial longitude of the target, deg Celestial longitude of the target, deg RST Sun-target renge, km Sun-target velocity, km/sec EPS Earth-probe-Sun angle, deg ESP Earth-bro-probe angle, deg EPM Earth-morpe hoon angle, deg EMP Earth-Moon-probe angle, deg EMP Earth-Moon-probe angle, deg EMP Earth-Moon-probe angle, deg EMP Earth-Moon-probe angle, deg EMS SEM Sun-Earth-moon angle, deg EMS Sun-Earth-Moon angle, deg EMS Sun-Earth-Sun-moon angle, deg EMS Earth-Sun-moon angle, deg ETP Earth-probe-target angle, deg ETP Earth-probe-target angle, deg ETP Earth-probe-angle, deg ETP Target-probe-Sun angle, deg ETP Target-probe angle, deg ETP Target-probe angle, deg ETP Target-probe angle, deg ETP Sun-target-probe near limb of target angle, deg ETP Clack angle of Earth, deg Clack angle of target, deg ETP Clack angle of target angle, deg CTP Canopus-probe-near limb of target angle, deg CTP Canopus-probe-near limb of target angle, deg CTP Canopus-probe-near limb of target angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle, deg CTP Canopus-probe-Earth angle,	ITE Celestial latitude of the Earth, deg Celestial longitude of the Earth, deg Celestial longitude of the target, deg Colestial latitude of the target, deg Sun-target range, km Sun-target renge, km Sun-target renge, km Sun-target renge, deg Earth-Sun-probe angle, deg Earth-Sun-probe angle, deg Earth-Probe—Sun angle, deg Earth-Sun-brobe angle, deg EMP Earth-probe—Sun angle, deg EMP Earth-probe—Sun angle, deg EMP Earth-probe—Sun angle, deg EMP Sun-Earth-probe angle, deg EMP Sun-Earth-probe angle, deg EMS Sun-Earth-brobe angle, deg EMS Sun-Earth-Moon—probe angle, deg EMS Sun-Earth-Moon—probe angle, deg EMS Earth-Sun-Moon angle, deg EMS Earth-Sun-Moon angle, deg EMS Earth-sun-Moon angle, deg EMS Earth-sun-Moon angle, deg EMS ITS Target-probe-sun angle, deg ETP Target-probe-sun angle, deg Target-probe-angle, deg Target-probe-angle, deg Target-probe-angle, deg Target-probe-angle, deg Sun-target-probe angle, deg Sun-target-probe angle, deg EMS Sun-Earth-target angle, deg EMS Sun-Earth-target angle, deg EMS Sun-Earth-probe angle, deg EMS Sun-target-Earth angle, deg EMS Sun-target-probe angle, deg EMS Sun-target-Earth angle, deg EMS Sun-target-probe angle, deg EMS Sun-target-Earth angle, deg EMS Sun-target-probe angle, deg EMS Sun-target-Earth angle, deg EMS Sun-target-Earth angle, deg Clock angle of target, deg SIN Canopus-probe-near limb of target angle, deg Corposus-probe-near limb of target angle, deg Corposus-probe-near limb of target angle, deg Corposus-probe-sone angle, deg Corposus-probe-sone angle, deg Corposus-probe-sone angle, deg Corposus-probe-sone angle, deg Corposus-probe-sone angle, deg Corposus-probe-sone and the principal direction of visible planet, cm DAS Direction intersection of ellipse with circle to the diameter (of the circle) that is perpendicular to D1, in construction of visible planet, cm DAS Direction intersection of ellipse with circle to the diameter (of the circle) that is perpendicular to D1, in construction of visible planet, cm DAS Direction intersection of dota and the principal p

Table D-2. (Cont'd)

Gr	oup	Trajectory constant	Gı	onb	Trajectory constant
Row 41	SMA ECC INC LAN APF RCA	Semimajor axis, km Eccentricity Inclination of the orbit plane to the equatorial plane, deg Longitude of the ascending node, deg Argument of pericenter, deg Magnitude of the closest approach vector, km	Row 47	SXI SYI SZI TX TY TZ	Components of the unit vector \$1 ^b along the direction of the incoming asymptote Components of the unit vector T ^b
Row 42	VH C3 C1 SLR APO TFP	Hyperbolic excess speed, km/sec Twice the energy (vis viva energy integral, km²/sec²) Angular momentum, km²/sec Semi-latus rectum, km Apogee distance, km Time from pericenter passage, sec	Row 48	BX BY BZ MX MY MZ	Components of the impact parameter B, b km Components of a unit vector which lies in the orbit plane and is normal to the radius vector R. $\mathbf{M} = \mathbf{W} \times \frac{\mathbf{R}}{\mid \mathbf{R} \mid}$
Row 43	TA EA MA DAI RAI MTA	True anomaly, deg Eccentric anomaly, deg Mean anomaly, deg Declination of the outgoing asymptote, ^b deg Right ascension of the incoming asymptote, ^b deg Maximum true anomaly, deg	Row 49	B·T B·R B PER DFF C3J	Projection of the impact parameter B ^b upon the vector T , km Projection of the impact parameter B ^b upon the vector R , km The magnitude of the impact parameter, ^b km Period, min Angle between the incoming and outgoing asymptotes, deg Earth—Moon Jacobi constant, km²/sec²
Row 44	wy wz	Components of a unit vector normal to the conic $\mathbf{W} = \frac{\mathbf{R} \times \mathbf{V}}{ \mathbf{R} \times \mathbf{V} }$	Group H	·-	Cartesian coordinates and epoch of injectic conditions in the geocentric equatorial sy tem described under Group B.
	PX PY }	Components of a unit vector in the direction of perigee	Row 50	XOCTAL YOCTAL ZOCTAL XOCTAL	Cartesian components of the probe radius vector at injection in octal representation, kn
Row 45	QY }	Components of a unit vector perpendicular to the perigee direction, vector P , and being in the orbit plane Q = W × P	Row 51	YOCTAL }	velocity vector at injection in octal represen- tation, km/sec Epoch of injection
	RX RY RZ	Components of the unit vector R ^b		YY MM DDD HH	Years past 1900 Month Day of month Hours
Row 46	sxo syo szo	Components of the unit vector 5 ₀ ^b along the direction of the outgoing asymptote		TT SSSSS SOCTAL	Min Msec Sec in octal representation The time past midnight Greenwich Meridia
	DAO	Declination of the outgoing asymptote, ^b deg			Time on (DD), month (MM) and year (YY
	RAO	Right ascension of the outgoing asymptote, ^b deg			1900) at which the injection epoch occurs
	TF	Time from injection to epoch of pericenter passage, hr			the time determined by the sum of HH, SSSSS, and SOCTAL.

Table D-3. Ranger 5 trajectory constants and conversion factors

Constants	Conversion factors	Constants	Conversion factors		
GM _{Sun}	1.32715445 × 10 ¹¹ km ³ /sec ²	Moon moments of inertia about	$A = 0.88746 \times 10^{29} \text{ kg km}^2$		
GMvenus	$3.247695 \times 10^5 \mathrm{km}^3/\mathrm{sec}^2$	principal axis	$B = 0.88764 \times 10^{29} \text{kg km}^2$		
GM⊕"	$3.986032 \times 10^5 \mathrm{km}^3/\mathrm{sec}^2$		$C = 0.88801 \times 10^{29} \text{kg km}^2$		
GM⊕_€	4.03503 × 10 ⁵ km ³ /sec ²	Lunar and solar ephemerides	The Moon and Sun positions are		
GM (b	$4.900759 \times 10^3 \mathrm{km}^3/\mathrm{sec}^3$		obtained from the joint JPL—S ephemerides. For purposes of		
GMMars	$4.297780 \times 10^4 \mathrm{km}^3/\mathrm{sec}^2$		converting into kilometers, the		
GM _{Jupiter}	1.267106 × 10 ⁸ km ⁸ /sec ²		conversion factors are: 1 AU = 1.495990 × 10 ⁸ km		
M _{Sun} /M _{Venus}	408645		1 e.r. = 6378.3149		
Msun/MEarth	332951.3	Geometrical Earth model, used in	Clarke spheroid of 1866		
Mearth/MMoon	81.335	locating tracking and launch- ing facilities upon the Earth	a = 6378.2064 km		
MSun/M Earth-Moon	328908		$b = 6356.5838 \text{ km}$ $e^2 = 0.006768657997291$		
Msun/Mmars	3,088,000	Earth potential function:	0.000,000.,,,,		
Msun/MJupiter	1047.39	'	n 3		
Equatorial radius of Earth	6378.3149 km	$\Phi (R, \phi) = \frac{GM \oplus}{R} \left[1 + \frac{JR_B^2}{3R^2} (1 - 3 + \frac{JR_B^2}{3R^2}) \right]$	$\sin^2\phi) + \frac{H}{5} \frac{\kappa E}{R^3} (3-5\sin^2\phi)$ (sin ϕ		
1 AU	1.495990 × 10 ⁸ km				
Ellipticity of Earth	1/298.3	$+\frac{D}{R_B}\frac{R_B^4}{(3-30)}$	$\sin^2\phi + 35 \sin^4\phi$		
Conversion from feet to meters	0.3048	1 "	*** *		
Atmospheric model	1959 ARDC	where			
Sidereal rotation rate of Earth	4.1780742 × 10 ⁻³ deg/sec	R = geocentric distance φ = geocentric latitude			
Universal constant of gravitation	6.671 × 10 ⁻²⁰ km ³ /kg sec ²	$J = 1.62345 \times 10^{-3}$			
Speed of light	2.997925 × 10 ⁵ km/sec	$H = -0.575 \times 10^{-6}$			
Mean Moon radius	1738.09 km	$D = 0.7875 \times 10^{-8}$			

 $^{3.9860005 \}times 10^{5} \text{ km}^{3}/\text{sec}^{2}$ was used for the premidcourse orbit.

 $^{^{\}rm b}$ 4.9007604 imes $10^{\rm a}$ km $^{\rm 8}/{\rm sec}^{\rm 2}$ was used for the premidcourse orbit.

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APPENDIX E

Ranger 5 orbit determination program printout

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		DYE DYE DYTE BRST EMP TSP TSP RPT	AND B.R	APPF APO RAI PY TY -		DY PTH - PTR -
TORIES		DX14840290 02 V .32397895 02 DXE13144366 02 DXT14160779 02 LOT .25313240 02 EPM .15240507 03 SEM .10572508 03 TPS .11184784 03 RPM .29270319 06	ORBITAL B.T	2437968.94619231 LAN .20074083 03 SLR .17535632 09 DAI29399695-01 PX .78124532 00 RX38197126-03 TX .62422358 00 MX42591886 00 PER .48752473 03		DX67951095 00 V .21514998 01 VR .24729806 01 LNE .80824179 01 DR20710000 01 SIA .15206486 03
SPACE TRAJECTORIES		Z19394000 05 ZE .20750000 03 ZI14497500 03 LTT55707544-02 SMP .68047762 02 TEP .20746221 02 EST .14179264 00	CONIC	JULIAN DATE JULIAN DATE C1 .48241571 10 MA91583388 01 WZ .99999862 00 Q215900818-02 S2I51310489-03 B2 .15900825-02 B .17832994 09		Z12072290 06 RA .27368568 03 LON .19310836 01 LTE .28183512 01 ALP .39241978 00 HNG .11232868 03
		Y .63461581 08 LAT74577254-02 YE .63376436 08 YI .63753740 08 LOE .25171779 02 ESP .3830838-01 MSP .1044440 00 ETP .6848699 01 STE .74133345 02 GCT .28289590 03		PASSAGE ECC .18185834 00 C373180260 03 EA11178339 02 WY .15625373-02 QY .78124393 00 SYI .6242232 00 SYI .6242232 00 BY78124426 00 B.R27816277 06		Y26609648 06 DEC24358288 02 LAT19733697 00 U LNS29389710 03 U SHA27148098 06 SVL11612395 02 H
CAŠE 1	HELIOCENTRIC	X .13480859 09 R .14899909 09 XE .13485415 09 XT .13479145 09 LTE .7978917-04 EPS .9788917-04 EPT .1524050 03 SET .10572508 03 SAC .00000000 00 GGE .11542093 03	HELIOCENTRIC	EPOCH OF PERICENTER PASSMA .18135415 09 VH .22507565 02 TA13416383 02 MX59170647-03 QX62422316 00 SXI .78124532 00 BX .62422343 09	SELENDCENTRIC	X .17140902 05 R .29270319 06 R .29276315 06 LTS .15113509 01 ALT .29096519 06 HGE .26712013 03 SAC .00000000 00

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	116	0202	1836581	8402 8405	184126	8422	184251	184351	184451	184526	184626	184726	184826	184851	849	185051	185126 185151	185226	185326	185351	185451	185551	185626 185651	185726 185751	185826	185926	193326	193321	193851	193951	194226 194251	194326 194351	194426

		.0068	•0039	0000	0.00	•0127	-0244	•0088	.0557	0078	-,0283		• 0 2 1 2	*000	*0029	0176		0010	0059	0137	-,0059		•050•	0050	.0107	-0303	0117	10101	1010	•020•	•0103	•0039	8600-
10/181 PAGE 2	EDD CC3	.12492133 06 .183 00	.12485984 06 .183 00	00 881 90 0000291	691.	.12473887 06 .183 00	.12467937 06 .183 00	.12462053 06 .183 00	.12456232 06 .183 00	.12450475 06 .183 00	.12444779 06 .183 00	201 70	5	.12433569 06 .183 00	.12428053 06 .183 00	.12422596 06 .183 00	;	.12417196 06 .183 00 .12411852 06 .183 00	90	.12401331 06 .183 00	.12396152 06 .183 00	60.	00 -103	.12385954 06 .183 00	.12380933 06 .183 00	.12375963 06 .183 00	.12371043 06 .183 00	12346172 04 163 00	G1. 00		.12356579 06 .183 00	.12351853 06 .183 00	.12347175 06 .183 00
PASS NUMBER		- 0220	*0660	0312*	0400	0354*	0475+	-0527	0373	7100	0388*	-*0396	0375*	0425*	7070	* 90.*0	0377*		0314#	• • • • • • • • • • • • • • • • • • •	0365*	0427*	0417*	0358*	0327*	1 0365		0332+	0408*	0411*	0425*	0425*	0353*
3	НА	00 101	•	03 .101 00	03 .101 00	03 .101 00	03 .101 00	101		101	03 .101 00	03 .101 00	03 .101 00	03 .101 00	161	161.	03 .101 00		00 101 00		00 161. 00	00 101 00	00 101. 00	00 101.00	00 101. 10	101		01 .101 00	00 101 10	00 101. 10	01 -102 00	01 -102 00	01 -192 00
NUMBER		34794042	V	.35809663	.35824401	.35839267	.35854259	35869372	35884606		.35899955	.35915421	.35930998	.35946685	35042401	10220100	.35978383		26703244		.43008431	.59410468	.75906304	.92494795	.10917442	1259432		.14280018	.15974252	.17676950	.19387998	.21107129	.22834344
ITERATION		0526#		0077*	0323*	0304+	0101*	0454#	-0262*	1	0245*	0365*	0241*	0313*	- 0300		0144*		0232*		0141*	0200+	0247*	0225*	0280*	0251+		+6670*-	0284	0245*	0164*	0139*	0211*
4 62/10/18	DEC	00 [01- [0 898088.		.62739139 01 .101 00	.63165575 01 .101 00	.63587463 01 .101 00	.64004871 01 .101 00	.64417873 01 .101 00	826565 01 . 101		98 31	.65631265 01 .101 00	.66027419 01 .101 00	.66419552 01 .101 00	101 10 21		.67191988 01 .101 00		.68322055 01 .101 00		00 101. 10 40616080.	.69057078 01 .101 00	.69419266 01 .101 00	00 101. 10 27977799.	.70133263 01 .101 00	.70485191 01 .101 00		00 101. 10 86/8880/-	.71179144 01 .101 00	-71521272 01 .101 00	.71860231 01 .101 00	.72196075 01 .101 00	.72528855 01 .101 00
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	TIME	194526	194626	194651	194751		4 4 4	195026 195051	195126	195226	195251	195351	151	195551	562	195726	195751	195926	200026 200051	200126	200226	200251	200351	200451	200526 200551	200626	200726	200826	200851	0095	U IO	55	201226 201251

		.0117	0020	0049	8400		9) 10*-	6500.	8900-	0070	1110.	.0010	0273	0488	0107		6200*-	-•0469	0430	0450	0450	0195	-,0088	-0342		6000*-	0273	0439	0332	0498	- 0107	•
10/181 PAGE 3	600	.12342543 06 .183 00	.12337956 06 .183 00	12333414 06 .183 00	183		06 .183	06 .183	00.183	00 .183	.12286257 06 .183 00	.12282210 06 .183 00	.12278201 06 .183 00	.12274229 06 .183 00	12266395 06 183 00		06 .183	.12258705 06 .183 00	.12254912 06 .183 00	.12251154 06 .183 00	.12247430 06 .183 00	90	06.183	06 . 183		06 .183	.12229307 06 .183 00	.12225778 06 .183 00	.12222281 06 .183 00	.12218815 06 .183 00	701 70	601.
PASS NUMBER		0430*		0333*	0386*	0405+	0371	+5550*-	0378	0453+	0541+	47870	1	-*0414*	-0490	0828+	0713*	*0790			0754*	0801+	0652*	0767*	0608	-*0654*	0725*	- 10 70	* 1890 * L	0801-	0867	0816*
PA		00 201		102 00	.102 00	.102 00	.132 00	.102 00	.102 00	102 00	192 00		701.	.102 00	102 00	.102 00	.102 00	162 00		.102 00	102 00	102 00	.102 00	102 00	.102 00	. 102 00		5	00 201.	. 102 00	132 00	.102 00
NUMBER 3	НА	24569375 01		.26312108 01	.28062505 01	.29820337 01	.31585527 01	.33357924 01	.42323995 01	.44137264 01	47782904 01			.51453364 01	.53297705 01	.57004002 01	.58865806 01	.60733290 01		.62606263 01	.64484765 01	.66368642 01	.68257818 01	.70152256 01	.72051802 01	.73956455 01	.75866103 01		*///80831 UI	. 19700039 01	.81624211 01	.83553110 01
ITERATION		40220	•0770•-	0246*	0289*	0289*	0167*	0281*	0313*	0292*	0321*		*7170	0100*	0286*	0111*	0290*	0366*		0141#	0153*	0223*	0110*	0336*	-*0039*	0301*	0280*		021/*	0233*	0365*	0217*
4 62/10/18	DEC	10 0000	101 • 10:06	.73185363 01 .101 00	.73509195 01 .101 00	.73830125 01 .101 00	.74148202 01 .101 00	.74463479 01 .101 00	00 101. 10 66686657.	.76298206 01 .101 00	76889005 01 . 101 00		00 101. 10 2/8081//-	.77469889 CI .101 00	.77756698 C1 .101 00	.78323184 01 .101 00	.78632940 Cl .101 00	78880405 01		.79155608 01 .101 00	.79428583 61 .101 00	.79699352 01 .101 00	.79967945 01 .101 00	.80234393 01 .101 00	.80498725 01 .101 00	.80760953 C1 .101 00	1021117 01		.81279247 01 .101 00	.81535352 01 .101 00	.81789467 01 .101 00	.32041617 01 .101 00
NUMBER	a G	76.	8276.5	76.	9 9	~ ~	76. 76.	8276.5	76. 76.	76. 76.	76.	76.	76.	76	76.	76.	76	76.	76.	76.	. 42.	76.	9/1	92	9,7	276. 276.	276.	276	276. 276.	276.	27.0	276. 276.
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	TIME	132	135 142	145	152 155	162 165	172	201826 201851	232	242	262	272	27	282	29	31.	322	33.	34	34.	35	396	37	38	39	940	147	245	042	0435	3 3	0452 0455

			7660	.0088	0039	0068	0352		*970"-	.0215	0068	-,0068	į	0266	0127	0127		8900.	0107	0225	0000		•0039	0322	.0078	0107	6450	7.00	•0244	0381	.0117		2050.	0254	.0127
10/181 PAGE 4	600	00 701 70 62011621	* DO	.12208597 06 .184 00	.12205250 06 .184 00	.12201933 06 .184 00	.12198643 06 .184 00	70.		.12192150 06 .184 00	.12188945 06 .184 00	.12185767 06 .184 00			.12179491 06 .184 00	.12176393 06 .184 00	è		.12170275 06 .184 00	.12167254 06 .184 00	.12164258 06 .184 00		. 12161288 U6 .184 U0	.12158341 06 .184 00	.12155419 06 .184 00	.12152521 06 .184 00	12149647 06 -184 00		.12146795 06 .184 00	.12143968 06 .184 00	.12141163 06 .184 00	00 701 70 002021C1	61.	12135620 06 .184 00	.12132883 06 .184 00
PASS NUMBER			1072*	1072*	0955*	70.00	****	0737*	0693+	0835*	- 0720*		0830*	0564	0542*	!	-*0464	*6990*-	0558	- 12.30	-1750-1	-*0908	+0670*-	0574*	- 0422	1 4 L	-,0553*	0608+	0487*	0508#		-*0574*	0602*	0713*	0850+
NUMBER 3 P	Ħ		.85486623 01 .102 00	.87424746 01 .102 00	.89367331 01 .102 00		761- 10	.93265804 01 .102 00	.95221502 01 .102 00	.97181506 01 .102 00	99145665 01 102 00		.10111394 02 .152 00	.10308633 02 .102 00	.10506273 02 .102 00		.10704309 02 .102 00	.10902733 02 .102 00	.11101547 02 .102 00		707 - 70	11500302 02 102 00	.11700240 02 .102 00	.11900539 02 .102 00				.12503581 02 .102 00	.12705287 02 .102 00	.12907336 02 .102 00		13109724 02 .102 00	.13312440 02 .102 00	.13515487 02 .102 00	.13718855 02 .102 00
ITERATION			0106*	0253*	0239*	0242#		0364*	0284*	0162*	0198*		*2610*-	0245*	0355*		0103+	0233*	0218*	0262*		0265*	0126*	0345*	.0037*	- 0179	-	0393*	0206+	0198*		*0500-	0217*	0124*	0330*
4 62/10/18	DEC		.82291812 01 .101 00	.82540092 01 .101 00	.82786477 01 .101 00	00 101 10 066088.		.632/3642 Ul .101 UU	.83514465 01 .101 00	.83753467 01 .101 00	.83990676 01 .101 00	00 101 10 7613228.	101. 10 101011	.84459823 01 .101 00	.84691803 01 .101 00		430776	.85150644 01 .101 00	.85377540 01 .101 00	.85602779 01 .101 00		.85826394 01 .101 00	.86048394 01 .101 00	.86268795 01 .101 00	.86487606 01 .101 00	.86704864 01 .101 00		.86920564 01 .101 00	.87134743 01 .101 00	.87347402 01 .101 00	87558578 01 101 00	101. 10 (100//10	.87768249 01 .101 00	.87976464 01 .101 00	.88183232 01 .101 00
NUMBER	F.R.	27	276.		276.	276. 276.	.975	7.6	8276.5	76.	9.	8276.5	76.	8276.5	76	76	9	8276.5	76.	8276.5	76.	76.	76.	76.	8276.5 8276.5	76. 76.	76.	76.	76.	~ ~	76.	76.	8276.5	76.	8276.5
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		.0313	•0146	-0049	-0020	0088	.0137	.0313	.0244	0000	5400		3050.	.0146	•0225	•0254	.0049	0.195	8400		000	*070*	.0303	.0010	•0205	•0303	0068	.0098	0166	7510	
10/181 PAGE 5	623	06 .184	06 .184	.12124799 06 .184 00	.12122148 06 .184 00	.12119517 06 .184 00	.12116906 06 .184 00	.12114317 06 .184 00	90	781 70	781		100	06 .184	.12096742 06 .184 00	.12094307 06 .184 00	.12091891 06 .184 00	184	781	101.	00 -184	06 . 184	.12080083 06 .184 00	.12077774 06 .184 00	.12075482 06 .184 00	.12073207 06 .184 00	.12070949 06 .184 00	.12068707 06 .184 00	90	06.184	
PASS NUMBER		0708+	1008*	- 0013-		*0260*-	0811*	0825*	0881	1062*	1665*	1475*	0767	-0840*	1035*		1372*	1558*	*6110*-	0825*	0854	1167*	1343*	-1299*	0715#		0/18+	0621*	0568*	0515*	0725*
	_	.102 00		201	701.	102	2 -102 00	2 .102 00	2 .102 00	2 .102 00	2 .102 00	2 .102 00	2 .102 00	2 .102 00	2 102 00	701	2 .102 00	2 -102 00	02 -102 00	00 201. 20	02 -102 00	02 -102 00	02 -102 00	107	102		02 -132 00	00 201. 20	00 201. 20	00 201. 20	02 .102 00
NUMBER 3	НА	.13922547 02	14126546 02			.14535476 02	.14740395 02	.14945618 02	.15151131 02	.15563032 02	.15769406 02	.15976066 02	.16182995 02	.16390204 02	14597478 02		.16805423 02	.17013424 02	.17221683 0	.17430209 0	.17638978	.17848003 0	-18057272 0				.18686529 0	.18896761	.19107217	.19317910	.19528824 0
ITERATION		0173*	0216*		* p / T n * -	0278*	*9600*-	0293*	0488*	0276+	0049*	0239*	0088*	0276*	7 100	- 700.	0148*	0332*	0135*	-*0520*	0017*	0197*	0175*	\$1.20.	-0367		-*0342#	0235*	0187*	0139*	0249#
4 62/10/18	DEC	.88388565 01 .101 00	8592662 01 - 101	372402 01 - 101	101. 10	.88996077 01 .101 00	.89195809 01 .101 00	.89394175 01 .101 00	00 101. 10 98119368.	.89981234 01 .101 00	.90174280 01 .101 00	.90366036 01 .101 00	.90556509 01 .101 00	45699 01		101 • 10 14	.91120333 61 .101 00	.91305795 01 .101 00	.91490033 01 .101 00	.91673055 01 .101 00	.91854890 01 .101 00	.92035527 01 .101 00	00 101- 10 26671226	02303392 01 101	101 10 00702500	101-10	.92746447 01 .101 00	.92921317 C1 .101 00	.93695057 01 .101 00	.93267693 01 .101 00	.93439244 01 .101 00
NUMBER	F.R.C.	8276.5	76.	8276.5	76. 76.	76.	76.	1.6	76.	6.5	76. 76.	76. 76.	76.	76.	92	9	76.	9.	76.	76.	76.	76.	76		9	276	276	276	276	276	276 276
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		• 0049	0039	.0313	8000	1050.	47.10	87.00	0186	9500	0000	6600	0400	0029	00068	0273	0049	0371	7710	20.	-0186	.0303	.0322	.0068	.0146	0371		**>0	.0127	.0273
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10/181		.12062079	.12059902	.1205//41	75765061	**************************************	1204021	12047163	12045092	12043035	1204009		. 12038966	.12036952	.12034952	.12032967	.12030994	.12029036	12027090		86162021•	.12023239	.12021333	.12019439	.12017558	12015690		+cocrost.	16611021*	•12010159
SS NUMBER		+6180*-	1152*	*6980*-	1631+	0331*	*******	0220-	0828*	+1160	+6980*-	0784*	-6110	1096*	1636+		*0661*-	1401*	0784*	1211+	1221•	-0950-	1042*		* / / 60 *	0872*	-*0830*	*6860*-	1069*	0991
PASS		.102 00	.102 00	.102 00	.102 00	.102 00	.102 00	.102 00	.102 00	102 00	.102 00	.102 00	.102 00	.102 00	102 00				.102 00	.102 00	.102 00	102 00	102 00		00 201.	102 00	.192 00	.103 00	.103 00	.103 00
ю	Ą	02	02	02	02	05	02	02	02	05.	05	02	02	02	02				02.	05	02	02	02		70	02	02	02	02	05
NUMBER		.19739958	.19951321	.20162897	-20374694	.20586700	.20798919	.21011344	.21223971	.21436810	.21649844	.21863079	.22076505	.22290132	.22503946	7702 (266	0+617177*	.22932141	.23146514	.23361073	.23575811	.23790728	.24005824		18017747.	.24436529	.24652137	.24867918	.25083861	.25299968
ITERATION		0138*	-*0066*	0233*	0139*	*0017*	0387*	0189*	0111+	0272*	0231*	.0010*	0047*	0344*	0160*	***	5 200 .	*8800°-	0240*	0173*	0123*	0073*	0222*	1 6	*0000*I	0177*	0164*	0249*	0153*	0177*
62/10/18	CEC	93609681 C1 .101 00	3779048 Cl .101 00	3947345 Cl .101 CO	4114573 01 .101 00	4280756 01 .101 00	4445895 01 .101 00	4610006 01 .101 00	4773081 01 .101 00	4935153 01 .101 00	5096217 01 .101 00	5256288 01 .101 00	5415355 01 .101 00	5573455 01 .101 00	5730579 01	5886750 01 101	1011	101. 10 4581408	6196225 01 •101 00	00 101 10 956669	6501949 01 .101 00	6653426 01 .101 00	6803987 01 .101 00	101 10 0676307	70 6585568	97102400 01 .101 00	97250269 01 .101 00	97397247 01 .101 00	97543359 01 .101 00	97688596 01 .101 00
4		6,	6.	6.	6.	6.	6	6.	6.	6.	6	6.	6	σ.	•	O	•	•	σ,	6.	6.	6	6.	C	•	6	6.	6.	6	6
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	TIME	41	1422	1435 1435 1442	1445	1455	1465	1475	1485	1495	1505	1515	1525	1532 1535	1542 1545	1552	1562	1565	1575	1585	1595	2002	2012	2022	2032	2035	2045	2055	2065 2065 2033	2012

		•0029	0000	0200-	60.00	0156	.0264	0215	0000	0088	0439	.0527	0146		0100	•0225	-0244	0127	8600	-0166	•0029	0010	0225	74.10	0.10	A	00000	6120.	0117
10/181 PAGE 7	ဧဘ	06 -185	06 .185	12004/36 06 185 00	601. 00	06 -185	06 .185	.11995928 06 .185 00	.11994200 06 .185 00	.11992483 06 .185 00	11990776 06 185 00	06 .185	Š		06 .185	06 .185	06 .185	.11979123 06 .185 00	.11977499 06 .185 00	.11975884 06 .185 00	.11974279 06 .185 00	.11972684 06 .185 00	.11971098 06 .185 00	00 901 70 00907011	105	00	681. 90	C81. 40	.11963307 06 .185 00
PASS NUMBER		+1160	1140*	1729*	1616*	1387*	3223*	1294*	1167*	1382+	0981*	1519*	1523	1943+	1733+	1577*	1548+	2212*	13707	*9091*-	-*0776*	1089*	1204*	1218*	1116*	1255*	1152*	2910*	2114*
PΑ		.103 00	.103 00	.103 00	.103 00	.103 00	.103 00	.103 00	.103 00	.103 00	•103 00	•103 00	.103 00	.103 00	.103 00	.103 00	.103 00				.103 00		.103 00	.103 00	.103 00	.103 00	.103 00	.103 00	.133 00
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	AND B.R	APP APO RAO PY TY TY OMD		DY PTH - DYE DYT RST EMP EMP TSP	SIN AND B.R	APPF APD RAI PY RY TY -	PTH PTH PTR
S	CRBITAL B.T A	.85036772 02 .85036772 02 .22279253 06 .21528134 02 .77941747 00 .15937591 00 .50521149 00 63293868 00		16379770 02 -32030639 02 14358929 02 15114530 02 .27922636 02 .96900760 02 .75201185 02 .75201185 02	.10109439 03 GRBITAL B.T A	978.72300189 .27795756 02 .1774400 09 .26360944 00 .65611123 00 .30427929-02 .75465786 00 46824320 00	-12652398 01 -22986936 01 -22925111 01 -66370633 01
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-		PASSAGE CC C3 C3 C4 C7 C7 SYC BYC BYC		LAT YE YT LOE ESP MSP ETP STE	GCT VEP	PASSAGE ECC C3 - EA - EA - QY - SYI B.R -	DEC LNS
	TRIC	OF PERICENTER -10378810 06 -19597239 01 -10430764 03 -38416634 00 49489864 00 84815130 00 -36476786 00	ENTRIC	.13147573 09 .14879894 09 .13174237 09 .13147381 09 .1347381 09 .10431366 03 .97049648 01 .96900760 02	0316712 9669124 RIC	CH OF PERICENTER MA .17511857 09 VH .23473234 02 TA21074296 02 WX .59324982-02 XX .75464039 00 XI .65611123 00 BX .75464068 00 • T .17290515 09	.19199375 04 .23683385 04 .23683367 04 .15330163 01
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SMA	42785396 04	ECC	15534725 01		.28179120 02	LAN	.21110008 32	APF	.14716034 02	RCA	.23680540 04	
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ITERATION	λQ	.31217070-0150999581-0110339658-0217416160-0421368627-0217011449-0133065676-0419376253-0433254107-0649376253-0433254107-064551738-0315258661-024551738-0315258661-024551738-0315258661-024551738-0315258661-024551738-03167354-024551738-03167354-024551738-03167354-024551738-03167366709-075666709-07
	ΝO	.16856791 0011788990 0019179592 00 .11702219-03 .14507317-04 .24801543-01 .37780545-01 .37780545-01 .14336361-04 .55198239-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06 .41193827-06
IMPACT	2	24373983 03 17465137 03 19179592 00 101339658-02 101339658-02 46202296 02 51212636 02 51212646 02 513249-02 54399211-01 93716736-03 9408259-01 9408259-01 9468526-02 54399211-01 9468526-02 54399211-01 93716736-04 19376253-04 19376253-04 19376253-04 19376253-04 3468526-02 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 3626958-05 36297933-06 3626958-05 3626958-05 3626958-05 3626958-05
CE MATRIX AF	>	17965148 03 20389116 03 17465137 03 17465137 03 17465137 03 50995811-01 516115518 02 52753750 02 74031388-01 1740799-02 10007366-01 16740799-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03 45517338-03
CCVARIANCE	×	.25760664 03 -17865148 03 -184573983 03 -1845791 00 -31217070-01 -32747848 02 -4045273-01 -59793069 02 -4045273-01 -35586000-03 -98403259-01 -17107875-02 -17107875-02 -175703987-03 -175703987-03 -17947118-05 -17947118-05 -17947118-05 -17947118-05 -18502896-06 -1850289596-06 -1850289596-06 -185028596-06 -1850289596-06 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -18604473-03 -186044070-08
		C C C C C C C C C C C C C C C C C C C

IMPACT PARAMETERS 62/10/21 155353

N MATRIX (TARGET ORBITAL PLANE)

	8.RO	B. TO	7.	C3	5.15	S.RS
8.RO	.13498505 04	22402019 01	.41325050-01	15793145-01	18645745 00	59219263-02
B.10	22402077 01	.16038554 03	45120993-02	37801365-01	69219189-01	15618060-02
7	.41325051-01	-,45121056-02	.39904870-05	30254727-05	78761252-05	50680620-06
63	15793137-01	37801369-01	30254775-05	.17535673-04	.22095362-04	.11603023-05
S. TS	18645745 00	69219182-01	-,78761249-05	.22095361-04	.63648742-04	.19505274-05
S.RS	-,59219254-02	15618038-02	50680666-06	.11603017-05	.19505277-05	.10676296-06
NORMAL	NORMALIZED N MATRIX					
	B.RO	8.10	7.	C3	5.15	S.RS
8.RO	00 66666666	48146168-02	.56306379 00	10265133 00	63612429 00	49329858 00
B.T0	48146290-02	.100000000 01	17835410 00	71279338 00	68509214 00	37742805 00
7	.56306382 00	17835435 00	.10000000001.	36167548 00	49420229 00	77645924 00
C3	10265127 00	71279345 00	36167606 00	00 16666666	.66137077 00	.84800732 00
\$1.8	63612430 00	68509209 00	49420226 00	.66137072 00	1000000001.	.74825018 00
S.RS	-,49329849 00	37742751 00	77645994 00	.84800692 00	.74825029 00	.100000001
DM/000	DM/DQO MATRIX					
	B.RO	8. T0	T.	C3	\$1.8	S.RS
×	.37860242 03	62915043 03	76171701 03	.11090058-01	.88300154 00	.92214173-02
>	73392525 02	.23021360 03	.22165992 03	34133501-02	28989519 00	26028199-02
7	14238562 03	.28774196 03	.41061202 03	57004266-02	44743651 00	52061935-02
č	.19943532 06	26777504 06	40179114 06	.55533881 01	.42461314 03	.50394530 01
ργ	.43053032 06	83235111 06	97959106 06	.14335638 02	.11545693 04	.11866120 02
70	16147645 06	.22848712 06	.29892461 06	43002419 01	33374950 03	36611142 01

													7.	.40779151-01	80733845-02	.39904870-05
. 04	. 03	- 04	. 03	. 04	1 02	1 02	. 02	1 02	101	. 02	101	RIAL PLANE)	8.11	10481256 03	.16968991 03	80733910-02
.51182302	. 55679747	. 50878539	.99525261	.50205437	. 69771308	.36740368	.12664175	.89892090	.71914333	.38861755	.76975188	IX (TARGET EQUATORIAL	B.RT	.13405460 04	10481257 03	.40779152-01
80	8.RO	8.TO	B.RT	8.11	1.	SMAA	SMIA	THETA	DEL T	DEL 8	DEL S	N MATRIX		B.RT	8.11	7.F

APPENDIX F

ODP format description

Sheet No. references are to Appendix E. All units are in kilometers and seconds unless otherwise specified.

is run out to lunar encounter (impact or closest approach). (See Appendix D.)

Sheet No. 1 Control card input.

Sheet No. 2 Inverse of the *a priori* covariance matrix of estimated parameters.

Sheet No. 3-5 Trajectory based on initial injection conditions before any convergence on data is started. Its format is explained in Appendix D.

Sheet No. 6 The normal equation coefficients combined with the *a priori* matrix.

Sheet No. 7 See the next page of this Appendix for an explanation of the format.

Sheet No. 8 Covariance matrix of estimated parameters or inverse of Sheet No. 6.

Sheet No. 9 Correlation matrix of estimated parameters.

Sheet No. 10 Residual page for a particular station.

Following the trajectory printout is the U matrix which maps the covariance matrix at injection to encounter. Immediately below the U matrix is the covariance matrix on the estimated parameters at impact or closest approach epoch. This is formed by mapping the covariance matrix at injection to impact in double precision.

The sheet following the covariance matrix contains three blocks. The first block is a covariance matrix N formed by mapping the upper 6×6 matrix of the covariance of impact into a new coordinate system (explained in Appendix A of this Report) (σ_{TL}^2 is in hr²). The second block is simply the correlation matrix of the first block covariance matrix. The third block is a mapping matrix which maps injection components into the $\mathbf{B} \cdot \mathbf{T}$, $\mathbf{B} \cdot \mathbf{R}$, etc. system.

B = The vector measured from the center of the Moon perpendicular to the incoming asymptote (in kilometers).

GMT	TC	Q	FRQ
XX XX XX hr min sec	X Doppler count time in sec.	X Trans- mitting station	XXXX.X Last digits in transmitter frequency 2966 XXXX.X in cps.

CC3 (HA or DEC)

.XXXXXXX XX Two-way doppler (CC3) value in cps, or hour angle in deg or declination in deg (floating point number)	.XXX XX Associated weight in floating point	.XXXX ^h Residual (observed minus calculated) in cps.
--------------------------------------------------------------------------------------------------------------------	------------------------------------------------------------	-----------------------------------------------------------------

The sheet following the residuals has statistics on the previous residuals (self explanatory). The sheets following the statistics will have more residuals and statistics from other tracking stations. The sequence is then repeated for a few more iterations. On the last iteration a trajectory based on the converged estimated parameters

B·RO = The **B** vector dotted on the R axis in km (T axis in the Moon's orbital plane).

 $[\]mathbf{B} \cdot \mathbf{TO}$ = The **B** vector dotted on the **T** axis in km (**T** axis in the Moon's orbital plane).

 $[\]mathbf{B} \cdot \mathbf{RT}$ = The **B** vector dotted on the R axis in km (T axis in the equatorial plane of the Moon).

Data have been deleted from the fit.

1 Weighted sum of the squares of the residuals plus the product $\delta x^T \Gamma^{-1} \delta x$ where δx is the difference in the a priori Q and the value of Q on the particular iteration and Γ is an a priori covariance on Q.

JOB TITLE

Heration number		Epoch year	year/month/day	XX XXX XX	Clock XXXXX	SOS ^h XXXXX QSOS ¹ XXXXX
		GMT	•	hr min sec	(PC time now) hr min sec	Floating pointing numbers
O	DØ	STDEVDQ	OID Q	NEW Q	NOMINAL Q	DQ (NOM)
X, Y, Z = Position space fixed cartesian component in km	Difference in estimated parameters from previous iteration and this iteration	Standard deviations on estimated parameters	Value of estimated parameters from previous iteration (Initial estimate on 1st iteration)	Value of estimated parameters on this iteration	Initial estimate of parameters	Total difference in new Q and nominal Q
DX, DY, DZ = Velocity S.F.C. in km/sec						
$RI = Radius$ in $KE = GM_{\oplus}$ in km $^3/sec^2$						
LA = Latitude RE = Radius of in deg Earth to scale ephemeris in km						
LO = Longitude KM = GM (in in deg km3/sec2						
h Weighted sum of the squares of the residuals.	e residuals.					

 $\mathbf{B \cdot TT}$ = The **B** vector dotted on the *T* axis in km (*T* axis in the equatorial plane of the Moon).

 T_L = Linearized time of flight in hours.

SMAA = The largest eigenvalue of the upper 2 \times 2 of the N matrix (commonly called the semimajor axis of a 40% dispersion ellipse in the B plane).

SMIA = The semiminor axis of the dispersion ellipse or the other eigenvalue of the upper 2×2 .

THETA = The orientation angle of the semimajor axis of the dispersion ellipse measured counterclockwise from the T axis.

DEL T = Uncertainty in the time of flight in sec.

DEL B = $(N_{11} + N_{22})^{1/2}$ where N's are from the first block of this sheet.

 $DELS = V_{\infty}$ (DEL T) The position uncertainty in the direction of the incoming asymptote. Where $V_{\infty} =$ Hyperbolic excess velocity in km/sec.

ACKNOWLEDGMENT

The analysis presented in this Report represents the work of many people. Section VI illustrates the nearly complete dependence of the flight path analysis upon several complex digital computer programs. The steps in the development of such computing programs include the formulation of the physical and mathematical models of the processes, input and output requirements, programming and coding, checkout, continual modification and verification, and development and execution of in-flight operational procedures.

The development of the digital computer programs is a joint responsibility of the Systems Analysis Section (312) and the Computer Applications and Data Systems Section (372) at JPL. While these responsibilities often overlap, Section 372's responsibility includes programming and numerical analysis aspects, while Section 312's responsibility includes the physical models, specification of operational output, in-flight control, and overall coordination.

JPL's basic trajectory program has been developed almost completely by D. B. Holdridge of Section 372. His work includes the physical model as well as the programming. A. L. Laxdal has been very helpful in coordinating the trajectory with the ODP, in addition to developing the occultation package. Additional contributors are acknowledged in Ref. 7.

The ODP represents a new effort primarily by J. D. Anderson, Section 312, and R. H. Hudson, Section 372 (Ref. 11). This program

ACKNOWLEDGMENT (Cont'd)

allows for estimation of DSIF station locations and the major physical constants [i.e. GM_{\oplus} , $GM_{\mathfrak{q}}$, radius of the Earth, J, D, H (Earth's harmonics), astronomical unit (AU), speed of light (C)]. T. W. Hamilton, of Section 312, provided continual guidance in its development.

The tracking data editing program represents the work of D. W. Trask (312) and R. E. Holzman (372).

The very broad interface with the DSIF has involved the Communications Engineering and Operations Section 332 and Section 312 in joint efforts, including the noise models, calibration of antennas, physical and mathematical models of the systems used, accuracy requirements, data format and condition coding, and prediction and acquisition information. Primary contributions in these areas have been made by J. P. Fearey, C. W. Johnson, W. Wollenhaupt, D. D. Meyer of Section 332 and D. L. Cain, M. S. Johnson, O. Asderian, J. Reuyl, and T. W. Hamilton of Section 312.

Additional contributions to the analysis and programming were made by various members of Sections 372, 312, and 332, D. L. Cain, C. T. Thornton, V. C. Clarke, W. L. Sjogren, D. W. Trask, H. Lass, C. B. Solloway, F. G. Curl, M. R. Warner, M. W. Nead, K. C. Oslund and C. D. Coltharp. The authors regret that the above list is not complete and extend their appreciation to all other contributors.

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